



CLIENT/OWNER: APMM SUBANG AIR STATION	SERIAL NO.	HOURS	LDG/CYCLE	WORKSHEET NO: M72-02-5063
AIRCRAFT TYPE: AW139	AIRCRAFT: 31316	REFER WORKPACK		WORK/INSP/DESC: FLEET CHECK - ONE TIME INSPECTION
REGISTRATION: M72-03	#1 ENGINE: NIL	REFER WORKPACK		WORKPACK REF: 2024-5063
BASE/FACILITY: APMM,WMSA	#2 ENGINE: NIL	REFER WORKPACK		AJL REF NO.:
DATE IN: REFER WORKPACK	DATE OUT: REFER WORKPACK	NG / N1	NF / N2	SHEET: 1 OF 6

Reason for raising: FLEET CHECK FOR ONE TIME INSPECTION ON TAIL SECTION TO BE CARRIED OUT. REFER TO DGTA LETTER REF.: DGTA 213/3 DTD 21 MAC 2024	Raised by and date: CAMO; Nur Atira Najihah binti Anuar 11/03/2024	Other requirements/information: N/A
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Item	Description	Technician	* Eng. CRS	Date
1	03-01 TAIL ROTOR DRIVE COMPONENTS REFERENCE (DMC): N/A DO A GVI FOR CONDITION, SECURITY AND DAMAGE (TAIL ROTOR SHAFTS COWLINGS OPENING REQUIRED). PAY PARTICULAR ATTENTION TO NUMBER 2 TRDS IN THE SECTION ADJACENT TO THE AREA WHERE INTAKES ARE INSTALLED ON LEADING EDGE FAIRING. INCLUDES A GVI OF BALANCE PATCHES FOR CONDITION AND SECURITY OF ATTACHMENT. INCLUDES A GVI OF TR SHAFT COWLING AND ATTACHMENT FOR DAMAGE, CORROSION AND WEAR. (TAIL ROTOR SHAFTS COWLINGS OPENING REQUIRED). REMARKS:			
2	04-01 INTERMEDIATE AND TAIL ROTOR GEARBOX REFERENCE (DMC): N/A DO A GVI FOR LEAKS AND CORRECT OIL LEVEL. IF LEAKS ARE DETECTED, DETERMINE AMOUNT OF LEAKAGE (39-A-65-21-00-00A-364A-A AND 39-A-65-22-00-00A-364A-A), INCLUDING A GVI FOR GENERAL CONDITION AND SECURITY OF ATTACHMENT. REMARKS:			
3	04-02 TAIL ROTOR SERVOACTUATOR REFERENCE (DMC): N/A DO A GVI FOR DAMAGE, CONDITION, SECURITY AND LEAKS. IF LEAKS ARE DETECTED, DETERMINE AMOUNT OF LEAKAGE (39-A-67-32-01-00A-364A-A) REMARKS:			
4	04-03 TAIL ROTOR COMPONENTS, BLADES AND ROTATING CONTROLS REFERENCE (DMC): N/A DO A GVI FOR CONDITION, SECURITY AND DAMAGE. PITCH CHANGE MECHANISM AND ROTOR DAMPERS FOR CONDITION AND SECURITY. SPIDER AND SLIDER BOOT FOR DAMAGE AND CONDITION REMARKS:			
5	04-04 TAIL ROTOR PITCH CHANGE LINK ASSEMBLY SPHERICAL BEARINGS REFERENCE (DMC): N/A DO A GVI AND AN OC FOR PLAY. NO REMOVAL NECESSARY. NO QUANTITATIVE MEASUREMENT NECESSARY. IF UNUSUAL PLAY IS FELT, REMOVE PITCH LINK AND PERFORM A DI FOR CONDITION, DAMAGE AND PLAY (39-A-64-31-00-00A-31AA-A) REMARKS:			
6	04-05 TAILPLANE ATTACHMENTS REFERENCE (DMC): N/A DO A GVI FOR SIGNS OF RUBBER EXTRUSION AND TO CHECK TAILPLANE FREE PLAY REMARKS:			

- The work recorded above has been carried out in accordance with the AW139 Interactive Electronic Technical Publication (IETP), Air-vehicle Maintenance Planning Information (AMPI) - EASA and PT6C-67C Engine Maintenance Manual.
- The work recorded above has been carried out in accordance with the Airbus AS555SN Maintenance Manual, Airbus AS555SN Airworthiness Limitations Section (ALS) and Master Servicing Manual (MSM) and Arrius 1 A Maintenance Manual.
- The work recorded above has been carried out in accordance with Super Lynx Compound Interactive Electronic Technical Publication (CIETP) and CTS800-40N Engine Maintenance Manual.
- The work recorded above has been carried out in accordance with the Airbus AS365N3 Maintenance Manual, Airbus AS365N3 Airworthiness Limitations Section (ALS) and Master Servicing Manual (MSM) and Arriel 2C Maintenance Manual

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Item	Description	Technician	* Eng. CRS	Date
7	04-06 TAIL ROTOR BOOT REFERENCE (DMC): N/A DO A GVI FOR DAMAGE AND CONDITION REMARKS:			
8	04-07 TAIL ACCESS PANELS REFERENCE (DMC): N/A DO A GVI FOR CONDITION AND SECURITY REMARKS:			
9	29-02 PCM 1 AND ASSOCIATED HELICOPTER INSTALLATION REFERENCE (DMC): 39-A-29-10-02-00A-320A-A DO AN OC TO DETECT DORMANT FAILURES OF PCM 1 AND ASSOCIATED AIRCRAFT INSTALLATION REQUIREMENTS AFTER JOB COMPLETION 1 REMOVE ALL THE TOOLS AND THE OTHER ITEMS FROM THE WORK AREA. MAKE SURE THAT THE WORK AREA IS CLEAN. 2 CLOSE THE ACCESS DOOR 473AL. REFER TO 39-A-06-41-00-00A-010A-A 3 REMOVE THE PLATFORMS FROM THE LEFT AND RIGHT SIDES OF THE FUSELAGE. REMARKS :			
10	29-05 PCM 2 AND ASSOCIATED HELICOPTER INSTALLATION REFERENCE (DMC): 39-A-29-10-02-00A-320A-A DO AN OC TO DETECT DORMANT FAILURES OF PCM 1 AND ASSOCIATED AIRCRAFT INSTALLATION REQUIREMENTS AFTER JOB COMPLETION 1 REMOVE ALL THE TOOLS AND THE OTHER ITEMS FROM THE WORK AREA. MAKE SURE THAT THE WORK AREA IS CLEAN. 2 CLOSE THE ACCESS DOOR 473AL. REFER TO 39-A-06-41-00-00A-010A-A 3 REMOVE THE PLATFORMS FROM THE LEFT AND RIGHT SIDES OF THE FUSELAGE. REMARKS :			

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REGISTRATION: M72-03	#1 ENGINE: NIL	REFER WORKPACK		WORKPACK REF: 2024-5063
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DATE IN: REFER WORKPACK OUT: REFER WORKPACK			NG / N1 NF / N2	SHEET: 3 OF 6

Reason for raising: FLEET CHECK FOR ONE TIME INSPECTION ON TAIL SECTION TO BE CARRIED OUT. REFER TO DGTA LETTER REF.: DGTA 213/3 DTD 21 MAC 2024	Raised by and date: CAMO; Nur Atira Najihah binti Anuar 11/03/2024	Other requirements/information: N/A
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Item	Description	Technician	* Eng. CRS	Date
11	<p>29-07 TAIL ROTOR SHUTOFF VALVE REFERENCE (DMC): 39-A-29-12-04-00A-320A-A DO AN OC TO DETECT DORMANT FAILURES OF TRSOV AND PCM 2 OIL LEVEL SWITCHES</p> <p>REQUIREMENTS AFTER JOB COMPLETION 1 REMOVE ALL THE TOOLS AND THE OTHER ITEMS FROM THE WORK AREA. MAKE SURE THAT THE WORK AREA IS CLEAN. 2 CLOSE THE ACCESS DOOR 473AL. REFER TO 39-A-06-41-00-00A-010A-A 3 REMOVE THE PLATFORMS FROM THE LEFT AND RIGHT SIDES OF THE FUSELAGE.</p> <p>REMARKS :</p>			
12	<p>64-38 TAIL ROTOR INSTALLATION REFERENCE(DMC): 39-A-64-21-00-00A-310A-A DO A GVI OF TAIL ROTOR INSTALLATION COMPONENTS</p> <p>REQUIREMENTS AFTER JOB COMPLETION: 1 REMOVE ALL THE TOOLS AND THE OTHER ITEMS FROM THE WORK AREA. MAKE SURE THAT THE WORK AREA IS CLEAN. 2 REMOVE THE PLATFORM FROM THE RIGHT SIDE OF THE VERTICAL FIN.</p> <p>REMARKS:</p>			
13	<p>65-27 TAIL ROTOR DRIVE LINE COMPONENTS [27][28] REFERENCE (DMC): 39-A-65-00-00-00A-310A-A</p> <p>DO A GVI DUE TO MAINTENANCE OPERATIONS TO EXCLUDE ACCIDENTAL DAMAGE</p> <p>CONDITION LIMIT: OPPORTUNE AT EACH TAIL ROTOR DRIVE LINE COWLINGS REMOVAL (REF. ITEM 03-01)</p> <p>NOTE 27: HELICOPTERS AW139 THAT HAVE THE TAIL ROTOR DRIVE SHAFT INSTALLATION P/N 3G6500A00112 NOTE 28: HELICOPTERS AW139 THAT HAVE THE TAIL ROTOR DRIVE SHAFT INSTALLATION P/N 4G6500A00212</p> <p>REQUIREMENTS AFTER JOB COMPLETION: 1 REMOVE ALL THE TOOLS AND THE OTHER ITEMS FROM THE WORK AREA. MAKE SURE THAT THE WORK AREA IS CLEAN. 2 INSTALL/CLOSE THE APPLICABLE ACCESS PANELS/DOORS. REFER TO 39-A-06-41-00-00A-010A-A 3 REMOVE THE PLATFORM FROM THE RIGHT SIDE OF THE HELICOPTER.</p> <p>REMARKS :</p>			

- The work recorded above has been carried out in accordance with the AW139 Interactive Electronic Technical Publication (IETP), Air-vehicle Maintenance Planning Information (AMPI) - EASA and PT6C-67C Engine Maintenance Manual.
- The work recorded above has been carried out in accordance with the Airbus AS555SN Maintenance Manual, Airbus AS555SN Airworthiness Limitations Section (ALS) and Master Servicing Manual (MSM) and Arrius 1 A Maintenance Manual.
- The work recorded above has been carried out in accordance with Super Lynx Compound Interactive Electronic Technical Publication (CIETP) and CTS800-40N Engine Maintenance Manual.
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14	<p>67-03 ELECTRICAL CABLES IN CLOSE PROXIMITY TO FIXED FLIGHT CONTROLS REFERENCE (DMC): 39-A-67-31-00-00B-310A-A DO A GVI TO VERIFY CONDITION AND CLEARANCE (FAULT FINDING TASK)</p> <p>NOTE 3: USE THE LIMIT THAT OCCURS FIRST.</p> <p>NOTE 17: INSPECTION INTERVAL MUST BE REDUCED FROM 1200 FH/2 YEAR TO 1200 FH/1 YEAR IN CASE OF SALINE AND/OR HIGH HUMIDITY ENVIRONMENT OPERATION CONDITIONS.</p> <p>REQUIREMENTS AFTER JOB COMPLETION 1 CLOSE THE ACCESS DOOR 473AL. REFER TO 39-A-06-41-00-00A-010A-A 2 REMOVE THE PLATFORMS FROM THE LEFT AND RIGHT SIDE OF THE FUSELAGE.</p> <p>REMARKS:</p>			
15	<p>67-05 TAIL ROTOR CONTROL - CONTROL LINKAGES REFERENCE (DMC): 39-A-67-20-00-00A-310A-A DO A GVI</p> <p>NOTE 3: USE THE LIMIT THAT OCCURS FIRST.</p> <p>NOTE 17: INSPECTION INTERVAL MUST BE REDUCED FROM 1200 FH/2 YEAR TO 1200 FH/1 YEAR IN CASE OF SALINE AND/OR HIGH HUMIDITY ENVIRONMENT OPERATION CONDITIONS.</p> <p>REQUIREMENTS AFTER JOB COMPLETION 1 REMOVE ALL THE TOOLS AND THE OTHER ITEMS FROM THE WORK AREA. MAKE SURE THAT THE WORK AREA IS CLEAN. 2 INSTALL THE ACCESS PANELS 130AL, 131AL, 132AR, 140BL, 150AL, 160AL, 160BL, 180AL, 180BL, 310BL, 360CL AND 360CT. REFER TO 39-A-06-41-00-00A-010A-A 3 REMOVE THE PLATFORM FROM THE LEFT SIDE OF THE FUSELAGE.</p> <p>REMARKS:</p>			

- The work recorded above has been carried out in accordance with the AW139 Interactive Electronic Technical Publication (IETP), Air-vehicle Maintenance Planning Information (AMPI) - EASA and PT6C-67C Engine Maintenance Manual.
- The work recorded above has been carried out in accordance with the Airbus AS555SN Maintenance Manual, Airbus AS555SN Airworthiness Limitations Section (ALS) and Master Servicing Manual (MSM) and Arrius 1 A Maintenance Manual.
- The work recorded above has been carried out in accordance with Super Lynx Compound Interactive Electronic Technical Publication (CIETP) and CTS800-40N Engine Maintenance Manual.
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Item	Description	Technician	* Eng. CRS	Date
16	67-09 TAIL ROTOR SERVOACTUATOR INSTALLATION REFERENCE (DMC): 39-A-67-32-00-00A-310A-A DO A GVI NOTE 3: USE THE LIMIT THAT OCCURS FIRST. NOTE 17: INSPECTION INTERVAL MUST BE REDUCED FROM 1200 FH/2 YEAR TO 1200 FH/1 YEAR IN CASE OF SALINE AND/OR HIGH HUMIDITY ENVIRONMENT OPERATION CONDITIONS. REMARKS:			
17	67-10 TAIL ROTOR CONTROL BELLCRANKS Y8/Y9 AND Y9/Y10 REFERENCE (DMC): 39-A-67-20-00-00A-31AA-A DO A DI FOR EVIDENCE OF PLAY BETWEEN LEVER AND SUPPORT REQUIREMENTS AFTER JOB COMPLETION 1 INSTALL THE ACCESS PANELS 360DL, 360CL, AND 360AT. REFER TO 39-A-67-11-04-00A-720A-A 2 REMOVE THE PLATFORM FROM THE LEFT SIDE OF THE FUSELAGE. REMARKS :			
18	67-11 TAIL ROTOR CONTROL ROD YAW Y6 AND Y9 REFERENCE (DMC): 39-A-67-20-00-00A-31AB-A DO A DI FOR CORROSION DAMAGE AND CRACKS LOCATED AT THE END OF THE ALUMINIUM ALLOY ROD BODY IN THE CONICAL SECTION REQUIREMENTS AFTER JOB COMPLETION 1 REMOVE ALL THE TOOLS AND THE OTHER ITEMS FROM THE WORK AREA. MAKE SURE THAT THE WORK AREA IS CLEAN. 2 INSTALL THE ACCESS PANELS 180BL AND 360CL AND CLOSE THE ACCESS DOOR 360BL. REFER TO 39-A-06-41-00-00A-010A-A 3 REMOVE THE PLATFORM FROM THE LEFT SIDE OF THE FUSELAGE. REMARKS:			

<input checked="" type="checkbox"/>	The work recorded above has been carried out in accordance with the AW139 Interactive Electronic Technical Publication (IETP), Air-vehicle Maintenance Planning Information (AMPI) - EASA and PT6C-67C Engine Maintenance Manual.
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REGISTRATION: M72-03	#1 ENGINE: NIL	REFER WORKPACK		WORKPACK REF: 2024-5063
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Item	Description	Technician	* Eng. CRS	Date						
19	<p>MI64-02 TAIL ROTOR ELASTOMERIC SPHERICAL BEARING (PART NUMBER 3G6420V00153 AND 3G6420V00154)</p> <p>REFERENCE (DMC): 39-A-64-11-00-00A-31AB-A</p> <p>DETAILED INSPECTION FOR CRACKS IN THE ELASTOMER</p> <p>SPECIAL TOOLS:</p> <table border="1"> <thead> <tr> <th>DESCRIPTION</th> <th>SERIAL NO OR TC NO</th> <th>CALIBRATION DUE DATE</th> </tr> </thead> <tbody> <tr> <td>_____</td> <td>_____</td> <td>_____</td> </tr> </tbody> </table> <p>REQUIREMENTS AFTER JOB COMPLETION 1 REMOVE ALL THE TOOLS AND THE OTHER ITEMS FROM THE WORK AREA. MAKE SURE THAT THE WORK AREA IS CLEAN. 2 REMOVE THE PLATFORM FROM THE RIGHT SIDE OF THE FUSELAGE.</p> <p>REMARKS:</p>	DESCRIPTION	SERIAL NO OR TC NO	CALIBRATION DUE DATE	_____	_____	_____			
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_____	_____	_____								
20	<p>MAIN HYDRAULIC SYSTEM - OPERATION TEST</p> <p>REFERENCE (DMC): 39-A-29-10-00-00A-320A-A</p> <p>REQUIREMENTS AFTER JOB COMPLETION 1 CLOSE THE ACCESS DOOR 473AL. REFER TO 39-A-06-41-00-00A-010A-A 2 REMOVE THE PLATFORMS FROM THE LEFT AND RIGHT SIDES OF THE FUSELAGE</p> <p>REMARKS :</p>									
21	<p>ROTOR FLIGHT CONTROLS - OPERATION TEST</p> <p>REFERENCE (DMC): 39-A-67-00-00-00A-320A-A</p> <p>REQUIREMENTS AFTER JOB COMPLETION 1 REMOVE ALL THE TOOLS AND THE OTHER ITEMS FROM THE WORK AREA. MAKE SURE THAT THE WORK AREA IS CLEAN. 2 INSTALL/CLOSE THE ACCESS PANELS/DOOR 131AL, 180BL AND 213AL. REFER TO 39-A-06-41-00-00A-010A-A</p> <p>REMARKS :</p>									
22	<p>POWER PLANT - GROUND RUN</p> <p>REFERENCE (DMC): 39-A-71-00-00-00A-13BA-A</p> <p>REMARKS :</p>									

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