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AgustaWestland Products

SERVICE BULLETIN

OPTIONAL

N° 189-291

DATE: January 18, 2024

REV. : /

TITLE

ATA 25 - LOWER WIRE STRIKE PROTECTION SYSTEM KIT INSTALLATION

REVISION LOG

First Issue

1. PLANNING INFORMATION

A. EFFECTIVITY

S/N 49054, 49064, 49065, 49066, 49067

B. COMPLIANCE

At Customer's option.

C. CONCURRENT REQUIREMENTS

N.A.

D. REASON

This Service Bulletin is issued in order to provide the necessary instructions on how to install the “kit lower wirestrike protection system” P/N 8G9540F00211.

LH issued this SB for the following reason:

Helicopter Reliability/Maintainability	
Product Improvement	
Obsolescence	
Customization	✓
Product/Capability Enhancement	

E. DESCRIPTION

Part I of this Service Bulletin provides the necessary instructions on how to install the “Wirestrike Protection Structural Provision” P/N 8G5310A41211, which strengthen the lower nose area performing the “Fitting outboard assy” P/N 8G9540A02051 and “Fitting inboard assy” P/N 8G9540A02251 to allow the installation of the protection system.

Part II of this Service Bulletin gives the necessary instructions on how to perform the installation of the lower cutter P/N 8G9540V00131.

F. APPROVAL

The technical content of this Service Bulletin is approved under the authority of DOA nr. EASA.21.J.005. For helicopters registered under other Aviation Authorities, before applying the Service Bulletin, applicable Aviation Authority approval must be checked within Leonardo Helicopters customer portal. EASA states mandatory compliance with inspections, modifications or technical directives and related time of compliance by means of relevant Airworthiness Directives.

If an aircraft listed in the effectivity embodies a modification or repair not LH certified and affecting the content of this Service Bulletin, it is responsibility of the Owner/Operator to obtain a formal approval by Aviation Authority having jurisdiction on the aircraft, for any adaptation necessary before incorporation of the present Service Bulletin.

G. MANPOWER

To comply with this Service Bulletin, the following MMH are deemed necessary.

Part I: approximately ninety (90);

Part II: approximately six (6);

MMH are based on hands-on time and can change with helicopter configuration, personnel and facilities available. MMH are not comprehensive of the overall hours necessary to get access to work areas and to remove all the equipment that interferes with the application of the prescribed instructions.

H. WEIGHT AND BALANCE

PART I

WEIGHT (Kg)	1	
	ARM (mm)	**MOMENT (Kg mm)**
LONGITUDINAL BALANCE	396.6	396.6
LATERAL BALANCE	-276.9	-276.9

PART II

WEIGHT (Kg)	2,7	
	ARM (mm)	**MOMENT (Kg mm)**
LONGITUDINAL BALANCE	366.7	990.09
LATERAL BALANCE	-277.7	-749.79

I. REFERENCES

I.1 PUBLICATIONS

Following Data Modules refer to AMP:

DATA MODULE		**DESCRIPTION**	**PART**
DM01	89-A-00-20-00-00A-120A-A	Helicopter on ground for a safe maintenance.	I, II
DM02	89-A-06-41-00-00A-010A-A	Access doors and panels general data.	I, II
DM03	89-A-52-41-01-00A-520A-A	Nose door remove procedure.	I
DM04	89-A-11-00-01-00A-720A-A	Decal install procedure.	II

I.2 ACRONYMS & ABBREVIATIONS

AMDI	Aircraft Material Data Information
AMP	Aircraft Maintenance Publication
DM	Data Module
DOA	Design Organization Approval
EASA	European Aviation Safety Agency
IPD	Illustrated Parts Data
LHD	Leonardo Helicopter Division
MMH	Maintenance Man Hours
P/N	Part Number
S/N	Serial Number

I.3 ANNEX

N.A.

J. PUBLICATIONS AFFECTED

AW189 Aircraft Maintenance Publication (AMP)

AW189 Illustrated Parts Data Publication (IPD)

K. SOFTWARE ACCOMPLISHMENT SUMMARY

N.A.

2. MATERIAL INFORMATION

A. REQUIRED MATERIALS

A.1 PARTS

PART I

#	P/N	ALTERNATIVE P/N	DESCRIPTION	Q.TY	LVL	NOTE	LOG P/N
1	**8G9540F00211**		**KIT LOWER WIRE STRIKE PROTECTION**	**REF**	**.**		**-**
2	**8G5310A41211**		**WIRESTRIKE PROTECTION STRUCTURAL PROVISION**	**REF**	**..**		**-**
3	HL18PB-5-6		Pin-rivet	31	...		189-291L1
4	MS20426E3-7		Rivet	6	...		189-291L1
5	A298A05TW06		Rivet blind titanium	3	...		189-291L1
6	A298A05TW07		Rivet blind titanium	7	...		189-291L1
7	A875A3V06		Bolt	1	...		189-291L1
8	A875A3V07		Bolt	2	...		189-291L1
9	A428A3C11		Screw	1	...		189-291L1
10	A297A05TW05		Rivet blind	1	...		189-291L1
11	A297A05TW06		Rivet blind	3	...		189-291L1
12	A297A05TW07		Rivet blind	1	...		189-291L1
13	NAS1719H5L4A		Rivet	4	...		189-291L1
14	NAS1720H5L4A		Rivet	10	...		189-291L1
15	NAS1720H5L6A		Rivet	4	...		189-291L1
16	HL82-5AW		Collar	31	...		189-291L1
17	AN3C7A		Bolt	3	...		189-291L1
18	MS21069L3		Nut self-locking plate	3	...		189-291L1
19	NAS1149D0363K		Washer	3	...		189-291L1
20	**8G9540A01431**		**Fitting outboard assy**	**REF**	**...**		**-**
21	8G9540A02051	8G9540A01431A	Fitting outboard	1	....		189-291L1
22	NAS1097AD3-6		Rivet	0,1 kg	....		189-291L1
23	NAS1097AD3-7		Rivet	0,1 kg	....		189-291L1
24	NAS1791A4-2		Anchor nut	3	....		189-291L1
25	**8G9540A02131**		**Fitting inboard assy**	**REF**	**...**		**-**
26	8G9540A02251	8G9540A02131A	Fitting inboard	1	....		189-291L1
27	A900A3E2-03		Anchor nut	4	....		189-291L1
28	NAS1097AD3-6		Rivet	0,1 kg	....		189-291L1
29	NAS1097AD3-7		Rivet	0,1 kg	....		189-291L1
30	NAS1791A4-2		Anchor nut	3	....		189-291L1
31	8G9540A02551		Lower panel LH	1	...	(1)	-
32	8G9540A02631		FWD lower panel assy	1	...	(1)	-
33	A259A04-12		Setscrew, Nylon	6	...	(2)	-

PART II

#	P/N	ALTERNATIVE P/N	DESCRIPTION	Q.TY	LVL	NOTE	LOG P/N
34	**8G9540F00211**		**KIT LOWER WIRE STRIKE PROTECTION**	**REF**	**.**		**-**
35	**8G9540A02911**		**LOWER WIRESTRIKE PROTECTION INSTALLATION**	**REF**	**..**		**-**
36	NAS624H8		Bolt	3	...		189-291L2
37	NAS624H12		Bolt	3	...		189-291L2

#	P/N	ALTERNATIVE P/N	DESCRIPTION	Q.TY	LVL	NOTE	LOG P/N
38	MS20002C4		Washer	6	...		189-291L2
39	MS20995C32		Lock wire	0,45 Kg	...		189-291L2
40	8G9540A02751		Packer lower cable cutter	1	...		189-291L2
41	8G9540V00131		Lower cutter assy	1	...		189-291L2
42	**8G1120A18311**		**MARKINGS LOWER WIRE STRIKE**	**REF**	**..**		**-**
43	AW003DBHC023E08I		Decal	2	...		189-291L2

Refer also to IPD for the spares materials required to comply with the AMP DMs referenced in the accomplishment instructions.

A.2 CONSUMABLES

The following consumable materials, or equivalent, are necessary to accomplish this Service Bulletin:

#	SPEC./LH CODE NUMBER	DESCRIPTION	Q.TY	NOTE	PART
44	AWMS05-001 Type I, Class C, Grade 1	Sealant MC-780 C2 (C354)	AR	(3)	I,II
45	AWMS05-001 Type I, Class B, Grade 2	Sealant MC-780B-1/2 (C355)	AR	(3)	II
46	Commercial	Corrosion preventive compound CORBAN 27L spec BMS3-38	AR	(3)(4)	I,II
47	MIL-PRF-16173 Class I, Grade 1	Corrosion preventive compound Tectyl 891D	AR	(3)	I,II
48	MIL-PRF-16173 Class II, Grade 1	Corrosion preventive compound Tectyl 502C	AR	(3)	I,II
49	DTD900AA/4488A	Corrosion preventive compound JC5A (C001)	AR	(3)	I,II

Refer also to AMDI for the consumable materials required to comply with the AMP DMs referenced in the accomplishment instructions.

A.3 LOGISTIC MATRIX

In order to apply this Service Bulletin, the following Logistic P/N can be ordered in accordance with the applicable notes:

LOGISTIC P/N	Q.TY (PER HELO)	NOTE	PART
189-291L1	1		
8G9540A02551	1	(1)	Part I
8G9540A02631	1	(1)	
A259A04-12	6	(2)	
189-291L2	1		Part II

NOTES

(1) Component to be ordered only if kit “FLIR star SAFIRE 380 HLD” P/N 8G9350F00511 is installed on the helicopter.

(2) Item required only if Part II is NOT intended to be embodied immediately after Part I.

(3) Item to be procured as local supply.

(4) Item that may be used as alternative to jointing compound DTD900AA/4488A.

B. SPECIAL TOOLS

Refer also to ITEP for the special tools required to comply with the AMP DM referenced in the accomplishment instructions.

C. INDUSTRY SUPPORT INFORMATION

Customization.

3. ACCOMPLISHMENT INSTRUCTIONS

GENERAL NOTES

a) Place an identification tag on all components that are re-usable, including the attaching hardware that has been removed to gain access to the modification area and adequately protect them until their later re-use.

b) Where appropriate, parts are to be painted to match aircraft exterior final paint finish.

c) Exercise extreme care during drilling operations to prevent instruments, cables and hoses damage.

d) After drilling, remove all swarf and sharp edges. Apply on bare metal a light film of primer unless the hole is used for ground connection.

e) During the installation of bonding braids or components requiring grounding, clean the surface structure in order to obtain a good ground contact.

f) Let adhesive cure at room temperature for at least 24 hours unless otherwise specified.

g) Exposed thread surface and nut must be protected using a layer of tectyl according to MIL-C-16173 grade I.

h) All lengths are in mm.

PART I

1. In accordance with DM 89-A-00-20-00-00A-120A-A, prepare the helicopter on ground for a safe maintenance. Disconnect the battery, all electrical power sources and/or the external power supply.
2. In accordance with DM 89-A-06-41-00-00A-010A-A and with reference to Figures 1 thru 5, remove all external panels, internal panels and internal liners as required to gain access to the area affected by the installation and perform the “wirestrike protection structural provision” P/N 8G5310A41211 as described in the following procedure:

NOTE

All sharp edges shall be removed with chamfering 0.150÷0.400 x 45 degrees or radius 0.15÷0.400mm.

NOTE

Unless otherwise specified and except for electrical bonding areas, in low/medium indirect/direct exposure zones, perform the installation of riveted structural parts and riveted vendor components as follows:

- Apply a layer of sealant MC-780 C2 (C354) on all faying surfaces.
- Wet assemble fixing fasteners using MC-780 C2 (C354).

NOTE

Unless otherwise specified and except for electrical bonding areas, in low/medium indirect/direct exposure zones, perform the installation of bolted structural parts and bolted vendor components as follows:

- Apply a layer of MC-780 C2 (C354) on all faying surfaces.
- Wet assemble fixing fasteners using jointing compound DTD900AA/4488A (C001) applied under the head and on the shank of fasteners.

2.1 With reference to Figure 4 View C1 (WAS), remove and discard n°4 existing anchor nuts P/N A900A3 from “BL 275 LH keel beam longeron assy” P/N 4F5334A02331.

2.2 In accordance with applicable steps of DM 89-A-52-41-01-00A-520A-A, remove the radome hinge P/N 4F5331A58452 from the “hinge rear reinforcement LH” P/N 4F5331A21753.

2.3 With reference to Figure 4 View C1 (WAS), remove and discard the hinge rear reinforcement LH P/N 4F5331A21753 from “STA 280 bottom profile assy” P/N 4F5331A21432 and the panel assy P/N 4F5331A65634.

2.4 With reference to Figure 4 View C1 (WAS), remove and discard n°1 anchor nut P/N A900A3 from “STA 280 bottom profile assy” P/N 4F5331A21432.

NOTE

SAFIRE 380 HLD P/N 8G9350F00511 is installed on the helicopter.

2.5 With reference to Figure 4 View C1 (WAS), remove and discard the stand off P/N AW001CL005A01-X1 from the “BL 275 LH keel beam longeron assy” P/N 4F5334A02331.

NOTE

Restore protective treatment of the cut edge with Bonderite M-CR 1132 AERO (C598).

2.6 With reference to Figure 2 View E trim the “FLIR lower angle assy” P/N 8G9350A03631 in accordance with dimensions shown.

2.7 With reference to Figure 5 View C2 (BECOMES), replace the “FWD lower panel assy” P/N 8G5331P06031 with the “FWD lower panel assy” P/N 8G9540A02631. Fix it by means the existing hardware.

2.8 With reference to Figure 5 View C2 (BECOMES), replace the “lower panel LH” P/N 8G9350P00531 with the lower panel LH P/N 8G9540A02551. Fix it by means the existing hardware. Discard the 5 existing panel screws P/N A428AC09 in excess.

2.9 With reference to Figure 2 View B1 (WAS), remove and discard n°3 existing anchor nuts P/N A407A3C2P from canopy shell assy P/N 8G5320A00432.

2.10 With reference to Figure 2 View B1 (WAS), increase the diameter of n°3 existing holes on the canopy shell assy P/N 8G5320A00432 to Ø 6.71÷6.86.

2.11 With reference to Figure 2 View B1 (WAS), remove n°15 existing rivets from the “BL 275 LH keel beam longeron assy” P/N 4F5334A02331 and “STA 280 bottom profile assy” P/N 4F5331A21432.

2.12 With reference to Figure 1 View A1 and View A2, countersink n°10 existing holes on the canopy shell assy P/N 8G5320A00432 to Ø 8.46 100°.

2.13 With reference to Figure 1 View A1 and View A2, countersink n°2 existing holes on the canopy shell assy P/N 8G5320A00432 to Ø 6.26 100°.

NOTE

Perform the following step only if kit P/N 8G9900F00111 is NOT installed on the helicopter.

2.14 With reference to Figure 1 View A1, increase the diameter of n°3 existing holes on the “access panel assy” LH P/N 8G5320A12331 to Ø 6.71÷6.86.

NOTE

Perform steps from 2.15 thru 2.17 only if kit P/N 8G9900F00111 is installed on the helicopter.

2.15 With reference to Figure 1 View A2 remove and discard n°1 screw P/N A428AC10 and n°2 screws P/N A428AC12 from the “mounting flange assy MWR/RWR FWD LH” P/N 8G9930A01131 and packer P/N 8G9930A01951.

2.16 With reference to Figure 1 View A2, increase the diameter of n°2 existing hole on the mounting flange assy MWR/RWR FWD LH P/N 8G9930A01131 and on the “EMC filter bonded assy” LH P/N 8G9930A04731 to Ø 6.71÷6.86.

2.17 With reference to Figure 1 View A2, increase the diameter of n°3 existing holes on the packer P/N 8G9930A01951 to Ø 6.71÷6.86.

2.18 With reference to Figure 5 View C2 (BECOMES), temporarily locate the fitting inboard P/N 8G9540A02251 on the panel assy P/N 4F5331A65634 and on the BL 275 LH keel beam longeron assy P/N 4F5334A02331, and countermark n°50 holes positions.

2.19 With reference to Figure 5 View C2 (BECOMES), drill n°3 holes Ø 4.90÷5.03, n° 12 holes Ø 4.06÷4.17 and n°4 holes Ø 6.33÷6.43 thru the fitting inboard P/N 8G9540A02251.

2.20 With reference to Figure 5 View C2 (BECOMES), drill n°3 holes Ø 6.35÷6.50 thru the fitting inboard P/N 8G9540A02251.

2.21 With reference to Figure 5 View C2 (BECOMES), install on the fitting inboard P/N 8G9540A02251 n°4 anchor nuts P/N A900A3E2-03 by means of sealant MC-780 C2 (C354).

2.22 With reference to Figure 5 View C2 (BECOMES), install on the fitting inboard P/N 8G9540A02251 n°3 anchor nuts P/N NAS1791A4-2 by means of n°4 rivets P/N NAS1097AD3-7 and n°2 rivets P/N NAS1097AD3-6.

2.23 With reference to Figure 3 View B2 (BECOMES), temporarily locate the fitting outboard P/N 8G9540A02051 on the panel assy P/N 4F5331A65634 and on the BL 275 LH keel beam longeron assy P/N 4F5334A02331, and countermark n°52 holes positions.

2.24 With reference to Figure 3 View B2 (BECOMES), drill n°6 holes Ø 4.06÷4.17 and n°15 holes Ø 4.17÷4.24 thru the fitting outboard P/N 8G9540A02051.

2.25 With reference to Figure 3 View B2 (BECOMES) and Figure 5 View C2 (BECOMES), drill n°31 holes Ø 4.15÷4.18 thru the fitting outboard P/N 8G9540A02051, the BL 275 LH keel beam longeron assy P/N 4F5334A02331 and the fitting inboard P/N 8G9540A02251.

2.26 With reference to Figure 3 View B2 (BECOMES), drill n°3 holes Ø 6.71÷6.86 thru

the fitting outboard P/N 8G9540A02051.

2.27 With reference to Figure 3 View B2 (BECOMES), install on the fitting outboard P/N 8G9540A02051 n°3 anchor nuts P/N NAS1791A4-2 by means of n°4 rivets P/N NAS1097AD3-7 and n°2 rivets P/N NAS1097AD3-6.

NOTE

During the following step, to grant proper installation of bolt P/N A875A3V06 and bolts P/N A875A3V07, it is allowed to countersink panel P/N 8G9540A02551 and P/N 8G9540A02631 holes.

2.28 With reference to Figure 5 View C2 (BECOMES), install the fitting inboard P/N 8G9540A02251 on the "BL 275 LH keel beam longeron assy" P/N 4F5334A02331 by means of n°1 screw P/N A428A3C11, n°1 bolt P/N A875A3V06, n°2 bolts P/N A875A3V07 and n°2 rivets P/N NAS1719H5L4A.

2.29 With reference to Figure 5 View C2 (BECOMES), fix the fitting inboard P/N 8G9540A02251 on the STA 280 bottom profile assy P/N 4F5331A21432 and on the panel assy P/N 4F5331A65634 by means of n°8 rivets P/N NAS1720H5L4A and n°2 rivets P/N NAS1719H5L4A.

2.30 With reference to Figure 5 View C2 (BECOMES), install on the fitting inboard P/N 8G9540A02251 n°3 nut self-locking plates P/N MS21069L3 by means of n°6 rivets P/N MS20426E3-7.

2.31 With reference to Figure 4 View D, fix the radome hinge P/N 4F5331A58452 to the panel assy P/N 4F5331A65634 by means of n°3 washers P/N NAS1149D0363K and n°3 bolts P/N AN3C7A.

2.32 With reference to Figure 3 View B2 (BECOMES), install fitting outboard P/N 8G9540A02051 on the BL 275 LH keel beam longeron assy P/N 4F5334A02331 and on the bottom profile assy P/N 4F5331A21432, by means of n°7 rivets blind titanium P/N A298A05TW07, n°1 rivet blind P/N A297A05TW07, n°1 rivet blind P/N A297A05TW05, n°3 rivet blind titanium P/N A298A05TW06 and n°3 rivet blind P/N A297A05TW06.

2.33 With reference to Figure 3 View B2 (BECOMES), fix the fitting outboard P/N 8G9540A02051 on the STA 280 bottom profile assy P/N 4F5331A21432 and on the panel assy P/N 4F5331A65634 by means of n°4 rivets P/N NAS1720H5L6A and n°2 rivets P/N NAS1720H5L4A.

2.34 With reference to Figure 3 View B2 (BECOMES) and Figure 5 View C2 (BECOMES), fix the fitting inboard P/N 8G9540A02251 and the fitting outboard P/N 8G9540A02051 to the BL 275 LH keel beam longeron assy

P/N 4F5334A02331 by means of n°31 pin-rivet P/N HL18PB-5-6 and n°31 collar P/N HL82-5AW.

NOTE

Perform the following step only if Part II is NOT intended to be embodied immediately after Part I of this SB.

2.35 Close n°3 holes drilled at step 2.20 and n°3 holes drilled at step 2.26 by means of n°6 setscrews P/N A259A04-12.

3. In accordance with weight and balance changes, update the Chart A (see Rotorcraft Flight Manual, Part II, section 6).
4. Return the helicopter to flight configuration and record for compliance with Part I of this Service Bulletin on the helicopter logbook.
5. Gain access to My Communications section on Leonardo WebPortal and compile the "Service Bulletin Application Communication".

   As an alternative, send the attached compliance form to the following mail box:

   engineering.support.lhd@leonardo.com

   and (for North, Central and South America) also to:

   AWPC.Engineering.Support@leonardocompany.us

PART II

1. In accordance with DM 89-A-00-20-00-00A-120A-A, prepare the helicopter on ground for a safe maintenance. Disconnect the battery, all electrical power sources and/or the external power supply.
2. In accordance with DM 89-A-06-41-00-00A-010A-A and with reference to Figure 6, remove all external panels, internal panels and internal liners as required to gain access to the area affected by the installation and perform “lower wirestrike protection installation” P/N 8G9540A02911 as described in the following procedure:

NOTE

Perform the following step only if Part I of this SB has NOT been performed immediately before to Part II.

2.1 Remove and discard n°6 setscrews P/N A259A04-12 installed with step 2.35 during the application of Part I of this Service Bulletin. Clean the area and make sure that no damage has occurred to the aircraft or to the wirestrike protection structural provision P/N 8G5310A41211.

NOTE

Wet assemble bolts using sealant MC-780 C2 (C354).

NOTE

Perform the following step only if kit P/N 8G9900F00111 is NOT installed on the helicopter.

2.2 With reference to Figure 6 View A and View B, install the “lower cutter assy” P/N 8G9540V00131 and the “packer lower cable cutter” P/N 8G9540A02751 on the fuselage by means of n°6 washers P/N MS20002C4, n°3 bolts P/N NAS624H12 and n°3 bolts P/N NAS624H8. Safety the six bolts by means of lockwire
P/N MS20995C32. Torque the fixing hardware to 6.32 thru 8.92 Nm.

NOTE

Perform the following step only if kit P/N 8G9900F00111 is installed on the helicopter.

2.3 With reference to Figure 6 View A, install the lower cutter assy P/N 8G9540V00131 on the fuselage by means of n°6 washers P/N MS20002C4, n°3 bolts P/N NAS624H12 and n°3 bolts P/N NAS624H8. Safety the six bolts by means of lockwire
P/N MS20995C32. Torque the fixing hardware to 6.32 thru 8.92 Nm.

2.4 With reference to Figure 6 View A, apply sealant MC-780 C2 (C354) all around the

edge of the lower cutter assy P/N 8G9540V00131 at the interface with the fuselage only.

NOTE

When red colour 11136 is not suitable, according to the paint customisation of the aircraft exterior, colour yellow 13538 is a suitable alternative that shall be applied.

2.5 With reference to Figure 6 View B, paint the indicated surface with 11136 red colour.

3. With reference to Figure 7, perform “Markings Lower Wire Strike” P/N 8G1120A18311 as described in the following procedure:

3.1 In accordance with DM 89-A-11-00-01-00A-720A-A and with reference to Figure 7, apply n°2 decals P/N AW003DBHC023E08I in indicated positions.

4. In accordance with weight and balance changes, update the Chart A (see Rotorcraft Flight Manual, Part II, section 6).

5. Return the helicopter to flight configuration and record for compliance with Part II of this Service Bulletin on the helicopter logbook.

6. Gain access to My Communications section on Leonardo WebPortal and compile the “Service Bulletin Application Communication”.

As an alternative, send the attached compliance form to the following mail box:

engineering.support.lhd@leonardo.com

and (for North, Central and South America) also to:

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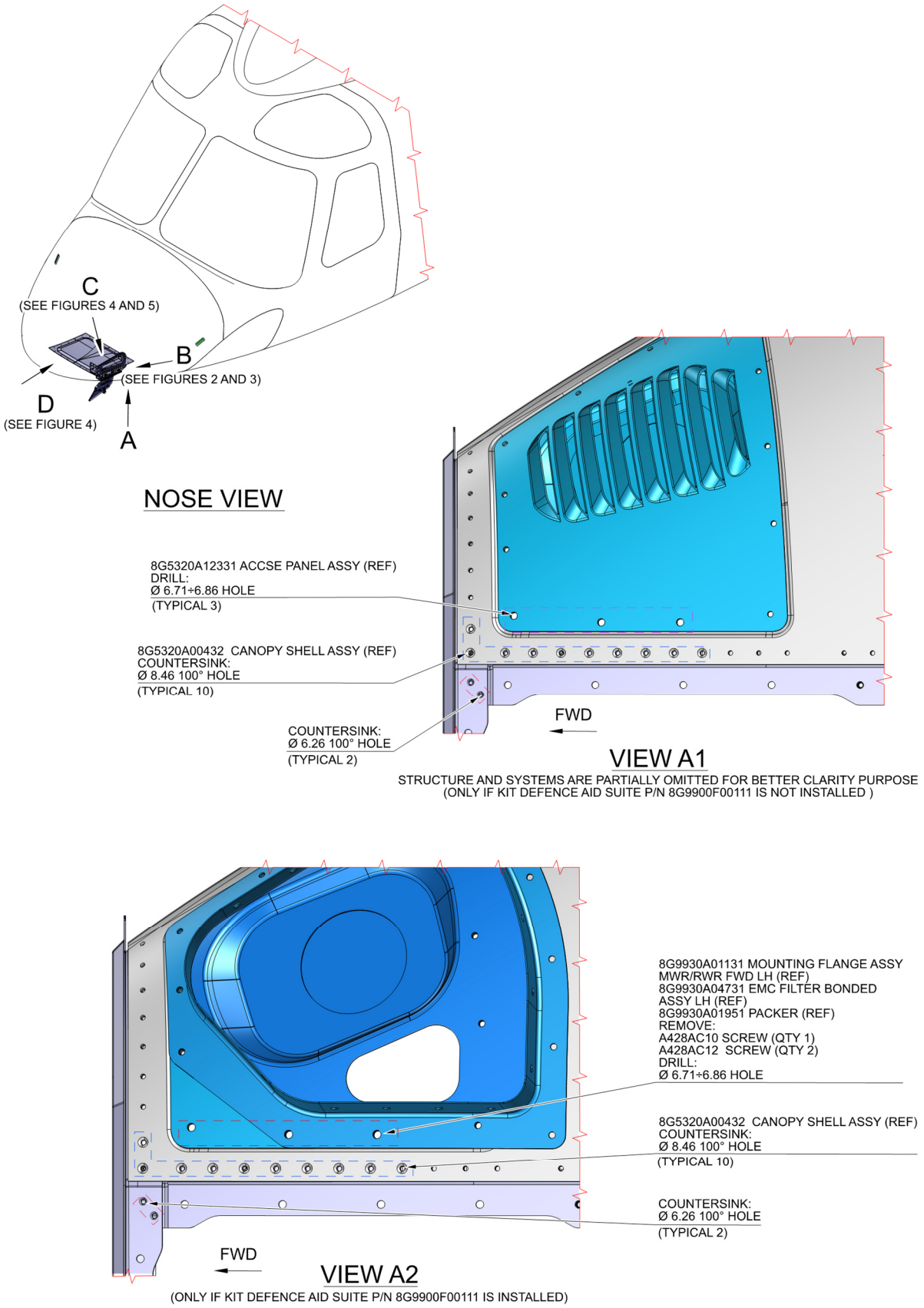
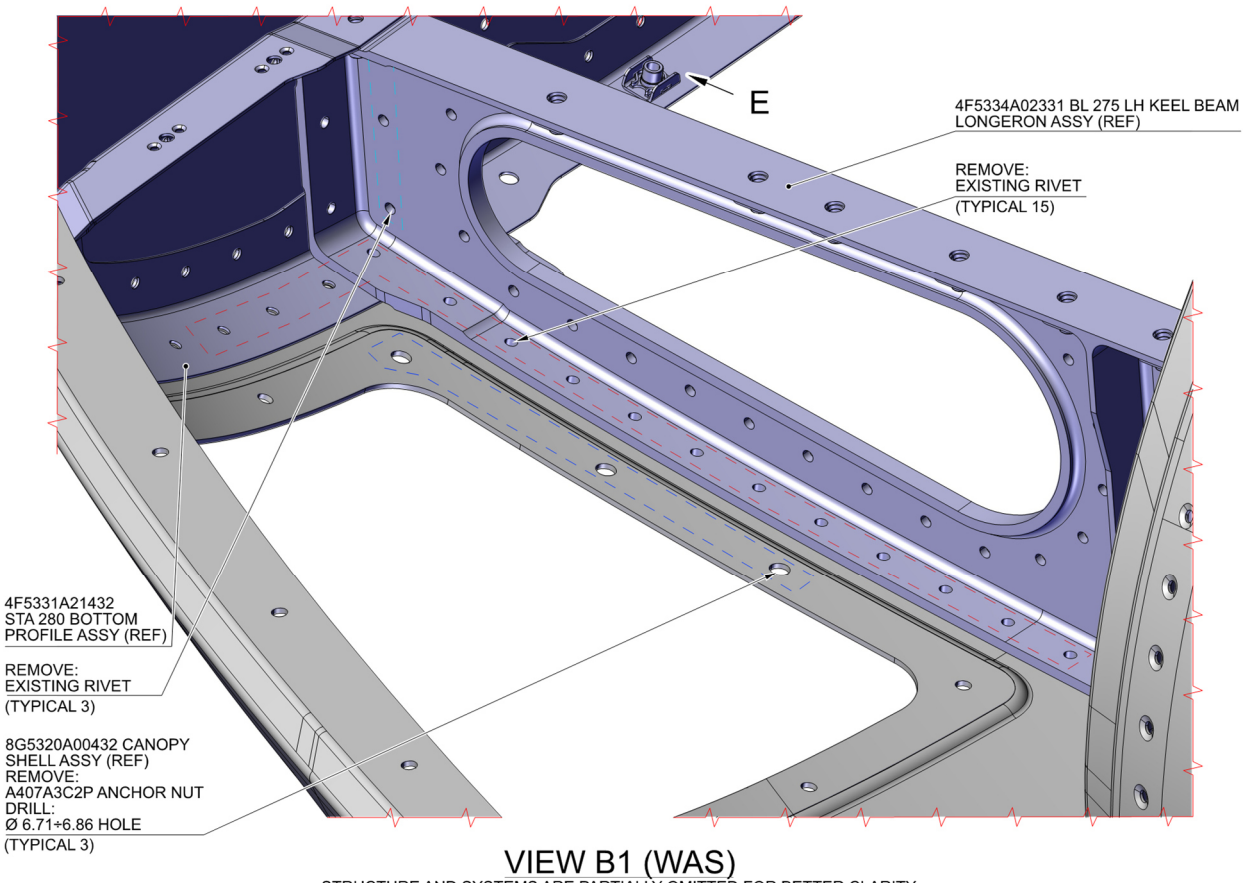
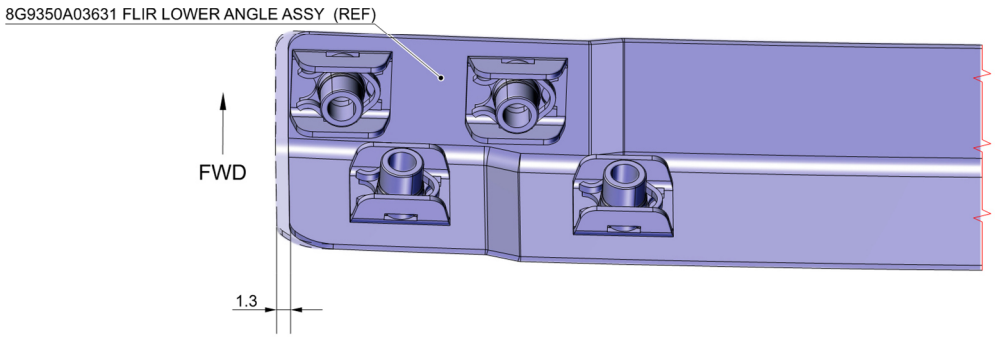

C
(SEE FIGURES 4 AND 5)
B
(SEE FIGURES 2 AND 3)
D
(SEE FIGURE 4)
A
NOSE VIEW
8G5320A12331 ACCSE PANEL ASSY (REF)
DRILL:
Ø 6.71÷6.86 HOLE
(TYPICAL 3)
8G5320A00432 CANOPY SHELL ASSY (REF)
COUNTERSINK:
Ø 8.46 100° HOLE
(TYPICAL 10)
COUNTERSINK:
Ø 6.26 100° HOLE
(TYPICAL 2)
FWD
VIEW A1
STRUCTURE AND SYSTEMS ARE PARTIALLY OMITTED FOR BETTER CLARITY PURPOSE
(ONLY IF KIT DEFENCE AID SUITE P/N 8G9900F00111 IS NOT INSTALLED)
8G9930A01131 MOUNTING FLANGE ASSY
MWR/RWR FWD LH (REF)
8G9930A04731 EMC FILTER BONDED
ASSY LH (REF)
8G9930A01951 PACKER (REF)
REMOVE:
A428AC10 SCREW (QTY 1)
A428AC12 SCREW (QTY 2)
DRILL:
Ø 6.71÷6.86 HOLE
8G5320A00432 CANOPY SHELL ASSY (REF)
COUNTERSINK:
Ø 8.46 100° HOLE
(TYPICAL 10)
COUNTERSINK:
Ø 6.26 100° HOLE
(TYPICAL 2)
FWD
VIEW A2
(ONLY IF KIT DEFENCE AID SUITE P/N 8G9900F00111 IS INSTALLED)
8G5310A41211
WIRESTRIKE PROTECTION STRUCTURAL PROVISION

Figure 1



VIEW B1 (WAS)

STRUCTURE AND SYSTEMS ARE PARTIALLY OMITTED FOR BETTER CLARITY
(REFER TO FIGURE 1)



VIEW E

ONLY IF KIT FLIR STAR SAFIRE 380 HLD P/N 8G9350F00511 IS INSTALLED

Figure 2

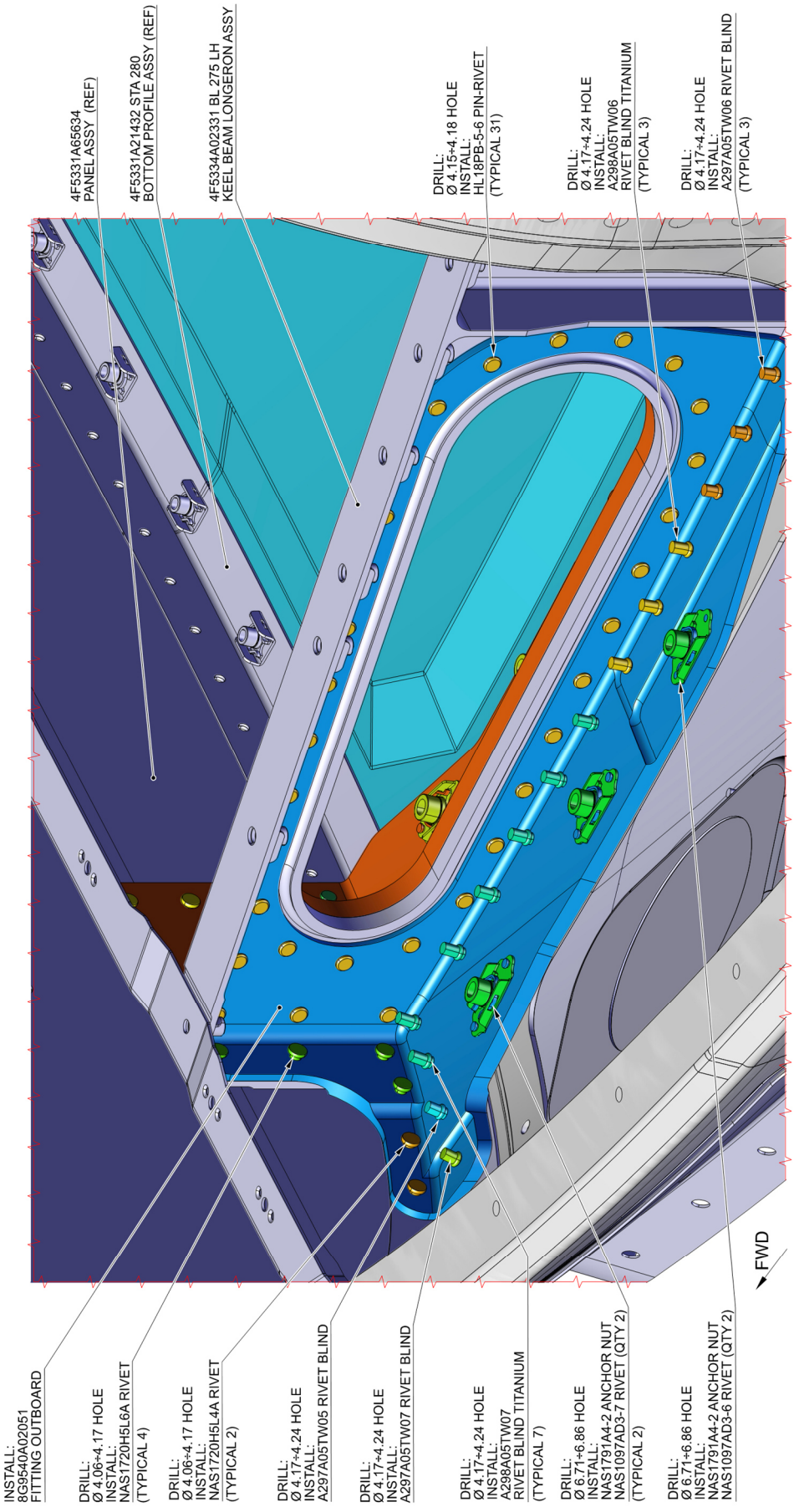
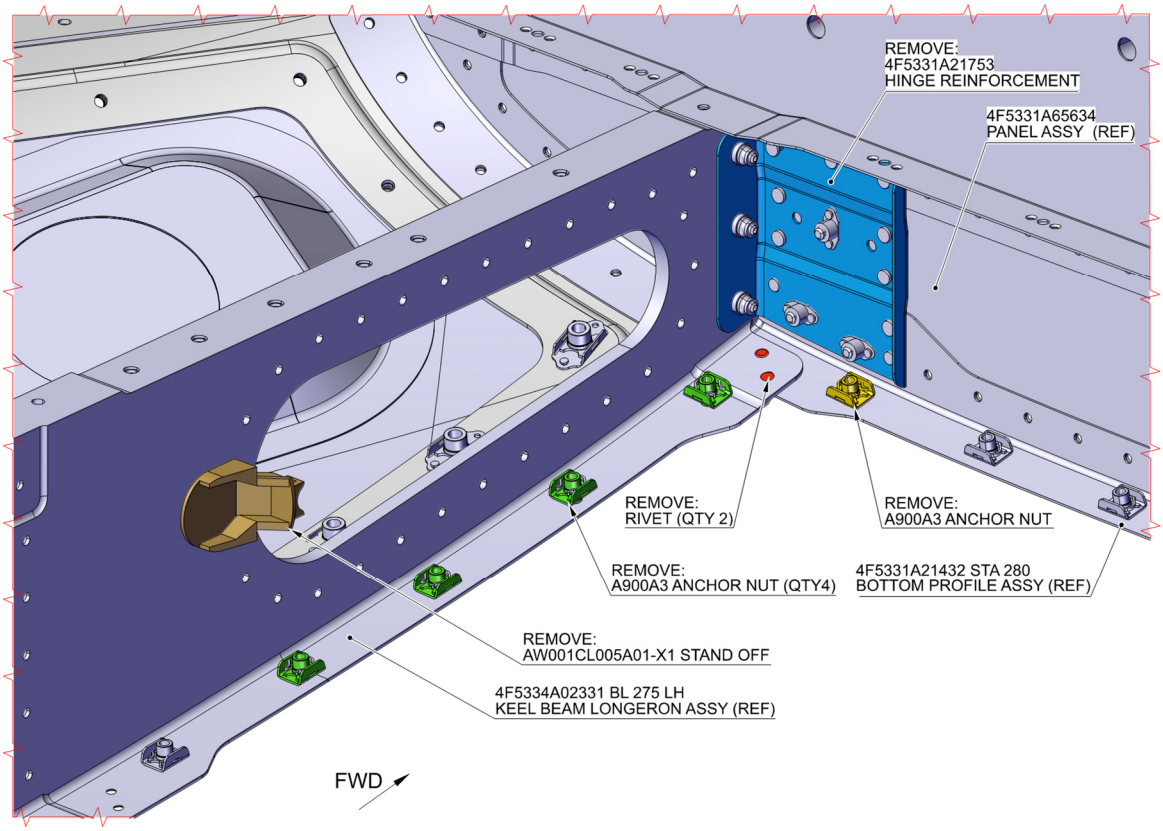
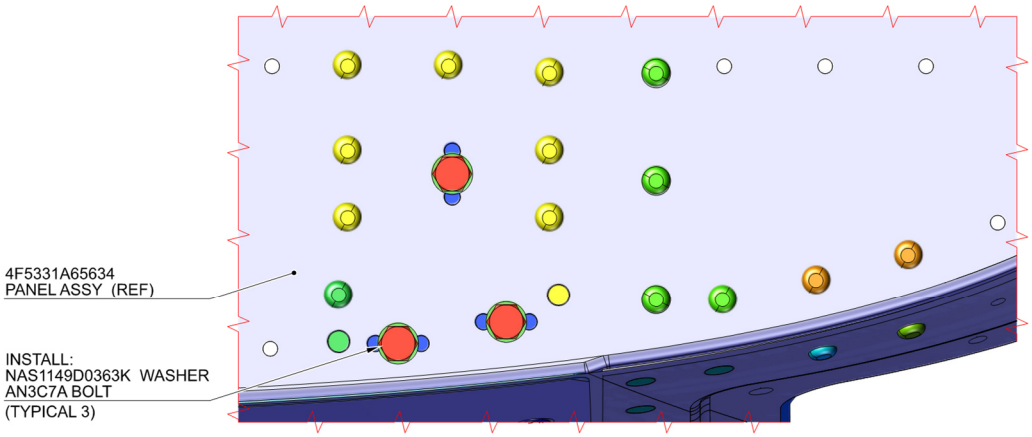

4F5331A65634
PANEL ASSY (REF)
4F5331A21432 STA 280
BOTTOM PROFILE ASSY (REF)
4F5334A02331 BL 275 LH
KEEL BEAM LONGERON ASSY
DRILL:
Ø 4.15÷4.18 HOLE
INSTALL:
HL18PB-5-6 PIN-RIVET
(TYPICAL 31)
DRILL:
Ø 4.17÷4.24 HOLE
INSTALL:
A298A05TW06
RIVET BLIND TITANIUM
(TYPICAL 3)
DRILL:
Ø 4.17÷4.24 HOLE
INSTALL:
A297A05TW06 RIVET BLIND
(TYPICAL 3)
INSTALL:
8G9540A02051
FITTING OUTBOARD
DRILL:
Ø 4.06÷4.17 HOLE
INSTALL:
NAS1720H5L6A RIVET
(TYPICAL 4)
DRILL:
Ø 4.06÷4.17 HOLE
INSTALL:
NAS1720H5L4A RIVET
(TYPICAL 2)
DRILL:
Ø 4.17÷4.24 HOLE
INSTALL:
A297A05TW05 RIVET BLIND
DRILL:
Ø 4.17÷4.24 HOLE
INSTALL:
A297A05TW07 RIVET BLIND
DRILL:
Ø 4.17÷4.24 HOLE
INSTALL:
A298A05TW07
RIVET BLIND TITANIUM
(TYPICAL 7)
DRILL:
Ø 6.71÷6.86 HOLE
INSTALL:
NAS1791A4-2 ANCHOR NUT
NAS1097AD3-7 RIVET (QTY 2)
(TYPICAL 2)
DRILL:
Ø 6.71÷6.86 HOLE
INSTALL:
NAS1791A4-2 ANCHOR NUT
NAS1097AD3-6 RIVET (QTY 2)
FWD
VIEW B2 (BECOMES)
(REFER TO FIGURE 1)

Figure 3



VIEW C1 (WAS)
STRUCTURE AND SYSTEMS ARE PARTIALLY OMITTED FOR BETTER CLARITY PURPOSE
(REFER TO FIGURE 1)



VIEW D
(REFER TO FIGURE 1)

Figure 4

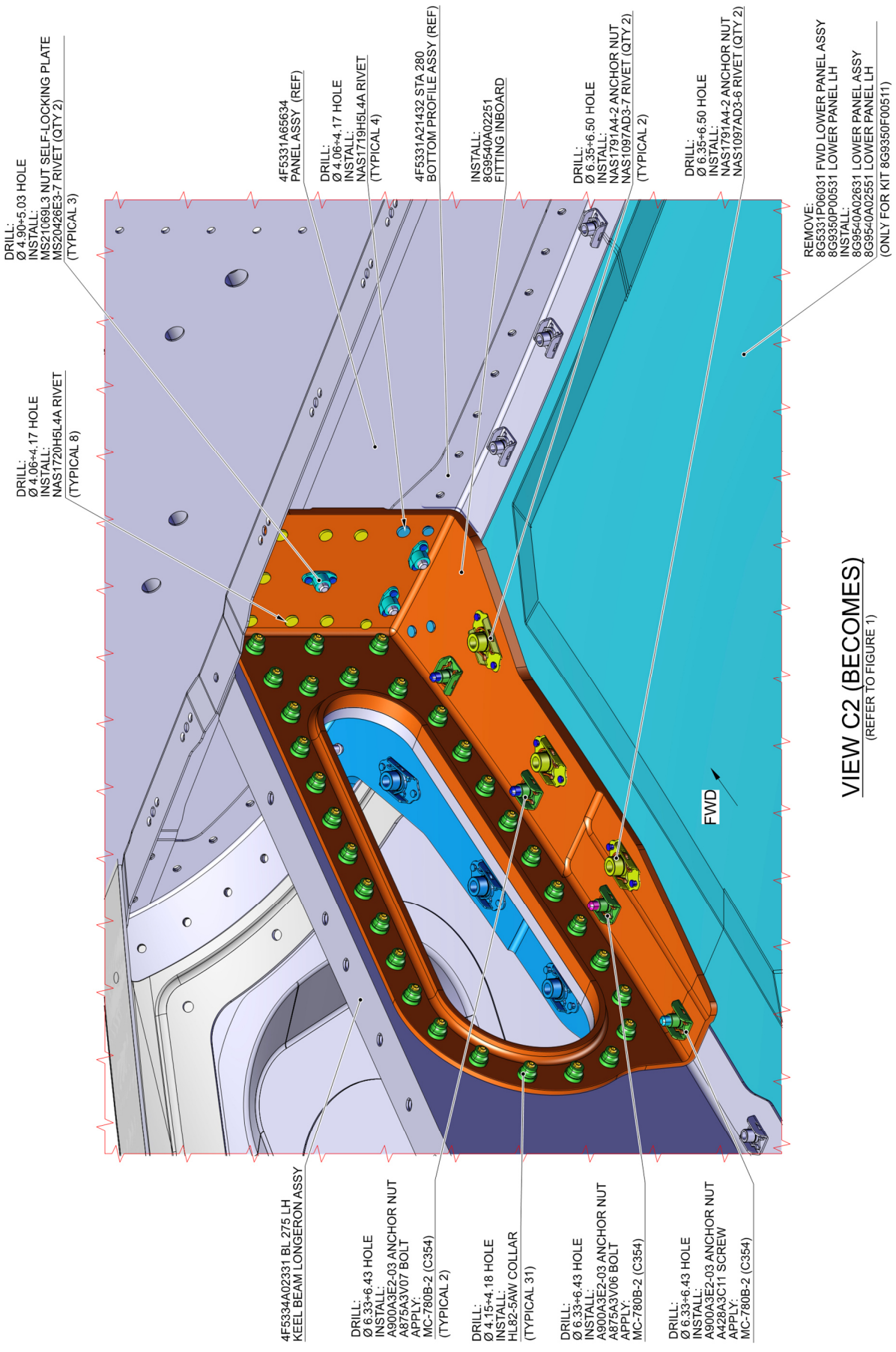

DRILL:
Ø 4.90÷5.03 HOLE
INSTALL:
MS21069L3 NUT SELF-LOCKING PLATE
MS20426E3-7 RIVET (QTY 2)
(TYPICAL 3)
DRILL:
Ø 4.06÷4.17 HOLE
INSTALL:
NAS1720H5L4A RIVET
(TYPICAL 8)
4F5331A65634
PANEL ASSY (REF)
DRILL:
Ø 4.06÷4.17 HOLE
INSTALL:
NAS1719H5L4A RIVET
(TYPICAL 4)
4F5331A21432 STA 280
BOTTOM PROFILE ASSY (REF)
INSTALL:
8G9540A02251
FITTING INBOARD
DRILL:
Ø 6.35÷6.50 HOLE
INSTALL:
NAS1791A4-2 ANCHOR NUT
NAS1097AD3-7 RIVET (QTY 2)
(TYPICAL 2)
DRILL:
Ø 6.35÷6.50 HOLE
INSTALL:
NAS1791A4-2 ANCHOR NUT
NAS1097AD3-6 RIVET (QTY 2)
REMOVE:
8G5331P06031 FWD LOWER PANEL ASSY
8G9350P00531 LOWER PANEL LH
INSTALL:
8G9540A02631 LOWER PANEL ASSY
8G9540A02551 LOWER PANEL LH
(ONLY FOR KIT 8G9350F00511)
4F5334A02331 BL 275 LH
KEEL BEAM LONGERON ASSY
DRILL:
Ø 6.33÷6.43 HOLE
INSTALL:
A900A3E2-03 ANCHOR NUT
A875A3V07 BOLT
APPLY:
MC-780B-2 (C354)
(TYPICAL 2)
DRILL:
Ø 4.15÷4.18 HOLE
INSTALL:
HL82-5AW COLLAR
(TYPICAL 31)
DRILL:
Ø 6.33÷6.43 HOLE
INSTALL:
A900A3E2-03 ANCHOR NUT
A875A3V06 BOLT
APPLY:
MC-780B-2 (C354)
DRILL:
Ø 6.33÷6.43 HOLE
INSTALL:
A900A3E2-03 ANCHOR NUT
A428A3C11 SCREW
APPLY:
MC-780B-2 (C354)
FWD
VIEW C2 (BECOMES)
(REFER TO FIGURE 1)

Figure 5

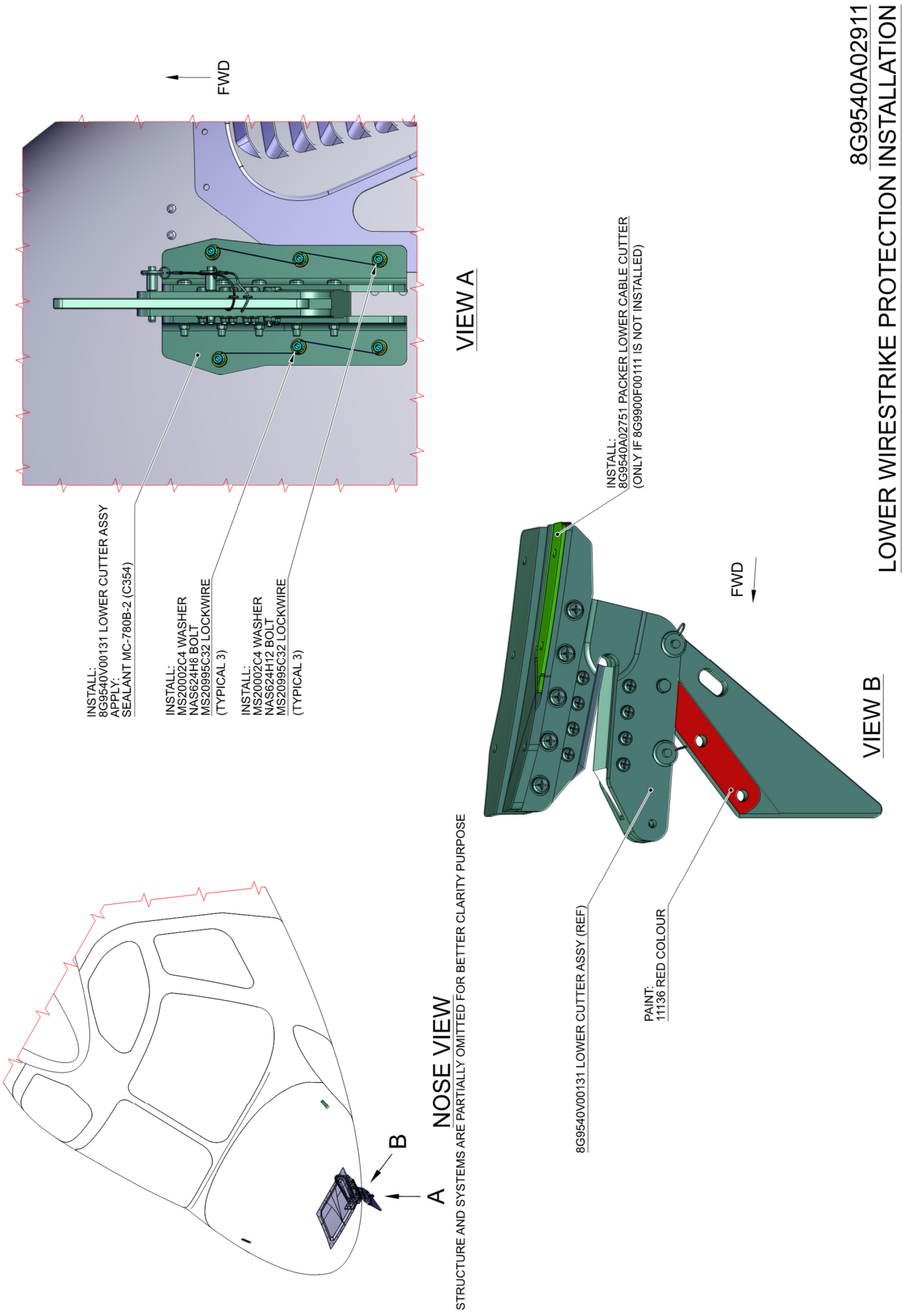

8G9540A02911
LOWER WIRESTRIKE PROTECTION INSTALLATION
FWD
VIEW A
INSTALL:
8G9540V00131 LOWER CUTTER ASSY
APPLY:
SEALANT MC-780B-2 (C354)
INSTALL:
MS20002C4 WASHER
NAS624H8 BOLT
MS20995C32 LOCKWIRE
(TYPICAL 3)
INSTALL:
MS20002C4 WASHER
NAS624H12 BOLT
MS20995C32 LOCKWIRE
(TYPICAL 3)
INSTALL:
8G9540A02751 PACKER LOWER CABLE CUTTER
(ONLY IF 8G9900F00111 IS NOT INSTALLED)
FWD
VIEW B
8G9540V00131 LOWER CUTTER ASSY (REF)
PAINT:
11136 RED COLOUR
B
A
NOSE VIEW
STRUCTURE AND SYSTEMS ARE PARTIALLY OMITTED FOR BETTER CLARITY PURPOSE

Figure 6

8G1120A18311
MARKINGS LOWER WIRE STRIKE

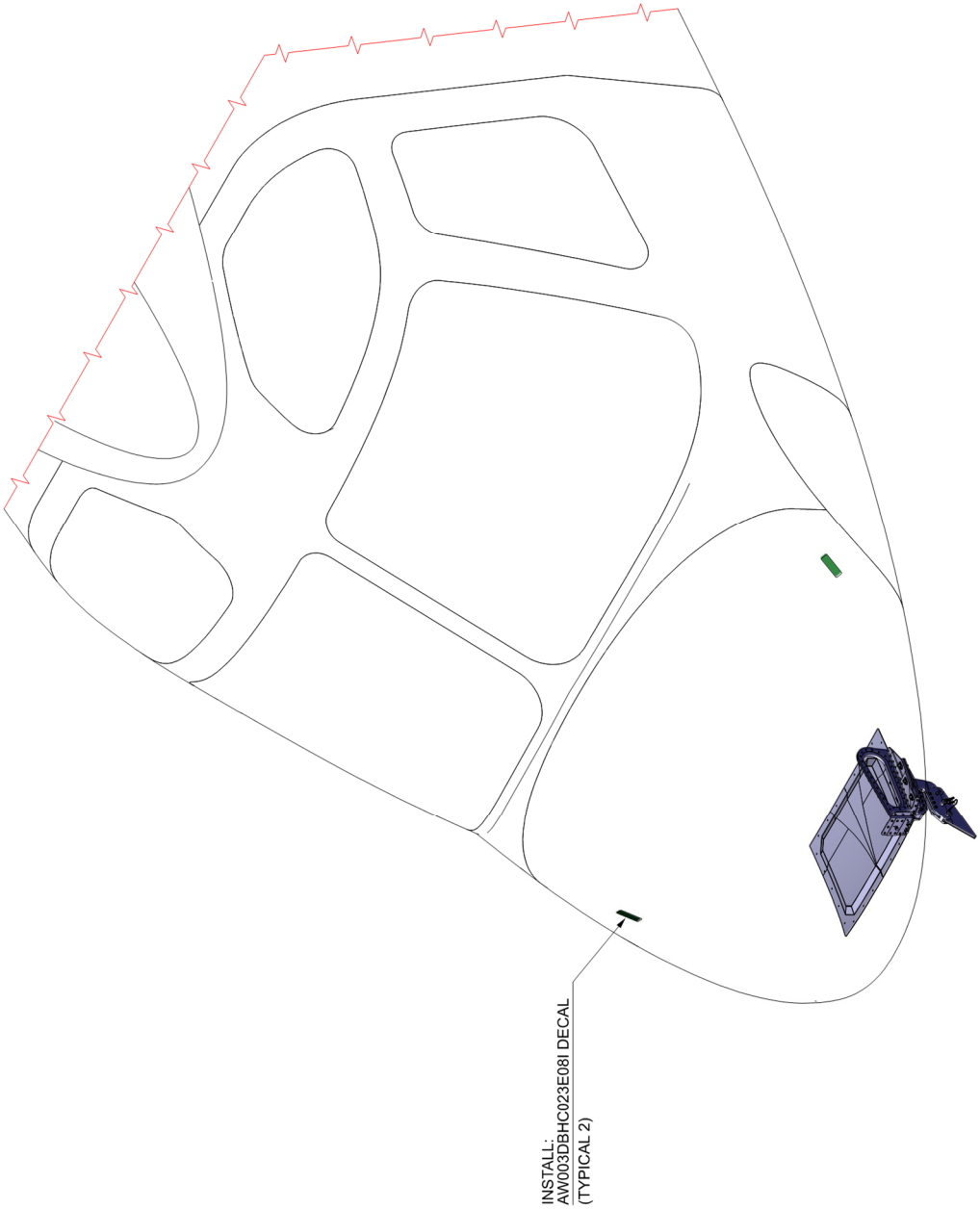


Figure 7

LEONARDO
HELICOPTERS

AgustaWestland Products

Please send to the following address: **LEONARDO S.p.A.** **CUSTOMER SUPPORT & SERVICES - ITALY** **PRODUCT SUPPORT ENGINEERING & LICENSES DEPT.** Via Giovanni Agusta, 520 21017 Cascina Costa di Samarate (VA) - ITALY Tel.: +39 0331 225036 Fax: +39 0331 225988	**SERVICE BULLETIN COMPLIANCE FORM**	Date:
	Number:	
	Revision:	

Customer Name and Address:	Telephone:
	Fax:
	B.T. Compliance Date:

Helicopter Model	S/N	Total Number	Total Hours	T.S.O.

Remarks:

Information:

We request your cooperation in filling this form, in order to keep out statistical data relevant to aircraft configuration up-to-date. The form should be filled in all its parts and sent to the above address or you can communicate the application also via Technical Bulletin Application Communication Section placed in Leonardo AW Customer Portal - MyCommunications Area. We thank you beforehand for the information given.