	AIRWORTHINESS DIRECTIVE No F-2002-606 R1		Distribution: A	Issue date: March 02, 2005	Page : 1/2
	This Airworthiness Directive is published by the DGAC on behalf of EASA, Airworthiness Authority of the State of Design for the affected product, part or appliance.			<i>Translation of « Consigne de Navigabilité » of same number. In case of difficulty, reference should be made to the French issue.</i>	
Direction générale de l'aviation civile France GSAC publication	No person may operate an aircraft to which an Airworthiness Directive applies, except in accordance with the requirements of that Airworthiness Directive, unless otherwise agreed with the Authority of the State of Registry.				
Corresponding foreign Airworthiness Directive(s): Not applicable			Airworthiness Directive(s) replaced: 2002-606 original issue		
Person in charge of airworthiness: EUROCOPTER			Type(s): EC 120 helicopters		
Type certificate(s) No. 189 TCDS No 189					
ATA chapter: 67	Subject: Rotor flight controls - Collective pitch lever				

1. **EFFECTIVITY:**

EC120 B helicopters with serial numbers below 1344.

Note: This Airworthiness Directive (AD) is intended for maintenance personnel and crews.

2. **REASONS:**

This AD is issued following a case of stiffening of the collective pitch lever control encountered in flight due to an increase of the friction factor of the friction mechanism pads.

The purpose of Revision 1 of this AD is to cover the conversion of EUROCOPTER EC 120 Alert Telex No. 67A009 into an Alert Service Bulletin (ASB) bearing the same reference number, with no change to the technical content. The effectivity according to the serial number of the helicopter is specified.


3. **MANDATORY ACTIONS AND COMPLIANCE TIMES:**

The following measures are rendered mandatory:

3.1. For crews, as from the next flight following the effective date of the original issue of this AD, and until compliance is ensured with § 3.2. below:

- Only slightly tighten the collective pitch lever friction,
- If stiff operation is felt in flight with the collective pitch lever, loosen the collective pitch lever friction immediately.

3.2. No later than within 110 flying hours or 3 months following the effective date of the original issue of this AD (the first limit reached is applicable), secure each pad + spherical bearing element to the lever using adhesive in compliance with the instructions specified in § 2.B.2.a. of referenced EUROCOPTER EC120 Alert Service Bulletin (ASB) No. 67A009.

	<p style="text-align: center;">AIRWORTHINESS DIRECTIVE No F-2002-606 R1</p>	<p>Distribution: A</p>	<p>Issue date: March 02, 2005</p>	<p>Page: 2/2</p>
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3.3. No later than within 110 flying hours or 13 months following compliance with the instructions specified in § 3.2. above (the first limit reached is applicable), check the bonding quality by complying once with the instructions specified in § 2.B.2.b. of the referenced ASB.

3.4. Before installing a replacement friction mechanism on an aircraft, comply with the instructions specified in § 2.B.2.a. of the referenced ASB.

3.5. No later than within 110 flying hours or 13 months following compliance with the instructions specified in § 3.4. above (the first limit reached is applicable), check the bonding quality by complying once with the instructions specified in § 2.B.2.b. of the referenced ASB.

4. REFERENCE PUBLICATION:

EUROCOPTER EC120 Alert Service Bulletin No. 67A009
(Any subsequent approved revision of this ASB is acceptable).

5. EFFECTIVE DATES:

Original issue : Upon receipt as from December 11, 2002
Revision 1 : March 12, 2005.

6. REMARK:

For any questions concerning the technical content of the requirements in this AD, please contact:

EUROCOPTER (STXI) - Aéroport de Marseille Provence, 13725 Marignane Cedex - France
Phone: +33 (0) 4 42 85 97 97 - Fax: +33 (0) 4 42 85 99 66
E-Mail: Directive.technical-support@eurocopter.com

7. APPROVAL:

This AD Revision is approved under EASA reference No 2005-1917 dated February 22, 2005.