



Airworthiness Directive

AD No.: 2018-0235

Issued: 05 November 2018

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

PILATUS AIRCRAFT Ltd

Type/Model designation(s):

PC-6 aeroplanes

Effective Date: 19 November 2018

TCDS Number(s): Switzerland No. F 56-10

Foreign AD: Not applicable

Supersedure: None

ATA 27 – Flight Controls – Flap Actuator / Pushrod Assembly Taper Pins – Inspection

Manufacturer(s):

Pilatus Aircraft Ltd; and Fairchild Republic Company, formerly Fairchild Industries, Fairchild Heli Porter, Fairchild-Hiller Corporation

Applicability:

PC-6 aeroplanes, all models, all manufacturer serial numbers (MSN), except those equipped with electrically operated flaps.

Definitions:

For the purpose of this AD, the following definitions apply:

The SB: Pilatus Aircraft Ltd Service Bulletin (SB) No. 27-005.

Affected part: Left-hand and right-hand flap actuator assemblies, having Part Number (P/N) 6132.0039.51 and P/N 6132.0039.52, and pushrod assemblies, having P/N 6132.0040.00.

Serviceable part: An affected part supplied as spare by Pilatus after 02 July 2018, or an affected part that, before installation, has passed an inspection (no defects found) in accordance with the instructions of Section 3.B of the SB, or has been corrected in accordance with the instructions of Section 3.C of the SB.



Reason:

During a recent overhaul, two new flap actuators were found to have taper pins installed that, apparently, had not been swaged. Investigation results identified that the taper pins had been incorrectly swaged during the manufacturing process.

This condition, if not detected and corrected, could lead to loss of one or both taper pins, consequent asymmetric flap deployment or flap surface flutter, possibly resulting in loss of control of the aeroplane.

To address this potential unsafe condition, Pilatus issued the SB to provide inspection instructions.

For the reason described above, this AD requires a one-time inspection of the taper pins of the affected parts for correct installation and, depending on findings, accomplishment of applicable corrective action(s). This AD also requires inspection of, and, depending on findings, corrective action(s) on, affected parts held as spare, prior to installation.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Inspection(s):

- (1) During the next scheduled 100 flight hours maintenance or annual inspection, or within 12 months after the effective date of this AD, whichever occurs first, inspect the taper pins of each affected part in accordance with the instructions of section 3.B of the SB.

Corrective Action(s):

- (2) If, during the inspection as required by paragraph (1) of this AD, any taper pin is found damaged, incorrectly swaged, or not swaged, before next flight, accomplish the applicable corrective action(s) in accordance with the instructions of section 3.C of the SB.

Parts Installation:

- (3) From the effective date of this AD, it is allowed to install on any aeroplane an affected part, provided it is a serviceable part, as defined in this AD.

Ref. Publications:

Pilatus Aircraft Ltd SB No. 27-005 original issue dated 02 July 2018.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 25 September 2018 as PAD 18-130 for consultation until 23 October 2018. No comments were received during the consultation period.



3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#).
5. For any question concerning the technical content of the requirements in this AD, please contact: Pilatus Aircraft Ltd, Customer Support Manager, CH-6371 Stans, Switzerland
Telephone: +41 41 619 33 33, Fax: +41 41 619 73 11
E-mail: SupportPC12@pilatus-aircraft.com, Website: www.pilatus-aircraft.com.

