A/C TYPE: AC155B
A/C REGN: 9M-SAS S/N 6583
TYPE OF CHECK:

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EUROCOPTER SOUTH EAST ASIA PTE LTD
48 Loyang Way Singapore 508740

NO: 15281 FORM NO: M/003
(C/F FROM CRS A14 & T/O)
JOB NO.:
PAGE: OF:

DEFECT AND RECTIFICATION WORKSHEET/CERTIFICATE OF RELEASE TO SERVICE

ITEM NO.	REPORTED BY DATE	DEFECT/WORK REQUIRED	ACTION TAKEN	COMPONENT CHANGES S/N OFF	COMPONENT CHANGES S/N ON	G.I.N. NO. OR O/HAUL REPORT NO.	MAN HOURS	MECH	LIC/ APP NO. DATE
1	ANDREW	TO RE-INSTALL FMS CDU AFTER	FMS CDU RE-INSTALLED AND CARRIED						
	29/10/02	REPAIR.	OUT FUNCTIONAL CHECK. FOUND SATIS'						
			P/N: 1117-12	-	364	S1083/2002		[illegible] 29/10/02	
			REFER ADD ITEM 2.				1		29/10/02
2	ANDREW	TO REPLACE NO. 2 ND DISPLAY	NO. 2 ND DISPLAY REPLACED AND						
	29/10/02	DUE TO APPEARANCE OF A	CARRIED GROUND CHECK. FOUND SATIS'	481	625		1		29/10/02
		WHITE HORIZONTAL LINE.	P/N: C19209VG11	~~625~~	~~481~~	S1087/2002			
3	ANDREW	TO COMPLY WITH ALLT TX 31A005R1	ALERT TELEX 31A005R1 COMPLIED						
	29/10/02	- INDICATING AND RECORDING	WITH AND FOUND 'SATIS'.				/		29/10/02
		SYSTEMS (LANDING GEAR	REFER AWM/024/01 ITEM 20.						
		EXTENSION INOPERATIVE)	MOD. KIT, P/N: 365P08-8837-0071	-	-	N3228/02			
4	ANDREW	LANDING GEARS UNABLE TO	LANDING GEAR HYD. RETURN CHECK	23-6×3-15-2481	980590		—		
	29/10/02	RETRACT.	VALVE FOUND SEAL TORN,	MS28775-014	2310/02				A1719
			REPLACED, LANDING GEAR						29/10/02
			RETRACTION & EXTENSION CARRIED OUT FOUND SATIS.						

THE WORK RECORDED ABOVE HAS BEEN CARRIED OUT IN ACCORDANCE WITH THE FOLLOWING REQUIREMENTS FOR THE TIME BEING IN FORCE AND IN THAT RESPECT THE AIRCRAFT/ EQUIPMENT IS CONSIDERED FIT FOR RELEASE TO SERVICE:

- ☐ SINGAPORE AIR NAVIGATION ORDER
- ☐ JAR 145 NO : F-04E
- ☐ _____
- ☑ MALAYSIAN CIVIL AVIATION REGULATIONS
- ☐ MANUFACTURER'S PUBLICATIONS

R.R.D. 100 /02-87

FICHE MATRICULE *LOG CARD* •

Fiche n° *Log card n°*

Fiche suiveuse Matériel neuf *Equipment record card New equipment* •

1 Identification du matériel *Material identification*

Dénomination *Name* SMD HS

Nomenclature *Nomenclature*

Référence fabricant *Part number* C19 209 VGM

Code fabricant *Manuf code* F9M1

Numéro de série *Serial number* HB1

Amendements *Amendments* ABC

Marque *Trade mark* THALES AVIONICS

Type *Type*

Identification du support ou du sous-support de rattachement
Related assembly or sub assembly identification

Code *Code*

Dénomination *Name*

2 Marché ou commande / *Contract or order*

Référence *Reference*

Date *Date*

Organisme émetteur *Issuing agency*

N° ou lot *Kit n°*

Fournisseur *Contractor*

Adresse *Address*

3 Garantie *Guarantee*

Matériel *Material*	Date de livraison *Date of delivery*	Durée garantie de stockage *Guarantee storage period*	Date de mise en service *Service date*	Durée garantie de fonctionnement *Guarantee operation period*
Neuf/*New*				

4 Renseignements particuliers *Special information*

(Document réception usine, dérogations, précaution à la mise en service, etc...)
(Acceptance document, concessions, precaution before putting into service...)

5

Unité ou société *Contractor*	Date *Date*	Positions *Location*	Fonctionnement *Operation* Support *Support*	Partiel *Partial*	Total *Total*	
THALES AVIONICS	12 99					Fab/*Manuf.* - Récept/*Receipt.* •
						Déstockage/*Depreservation* •
						V.P. / *Periodic Overhaul* •
						Montage/*Fitting* •

Limite de fonctionnement *Operating limit*

Limite de vie *Life limit*

• Rayer la mention inutile • *Delete as necessary*

6 Contrôle d'exécution des modifications *Modification checking*

Numéro *N°*	Date décision *Date of decision*	Nature de la modification *Type of modification*	Unité ou société d'exécution *Performing unit or contractor*	Date d'exécution et tampon contrôle *Performing date inspection stamp*
		ORIGINAL PASSED TO		
		ANDREW ON 29-10-02.		

5

Positions successives [illegible] opérations d'entretien et de remise en état min[illegible]s et majeures
Successive locati[illegible] mi[illegible] and major maintenance and overhaul operations (

Unité ou société *Contractor*	Date *Date*	Positions *Location*	Fonctionnement *Operation*			Motif du mouvement (code, symptômes) - Travaux effectués - Pièces changées *Reason for transfert (code and symtoms) - Accomplished work - Replaced parts*
			Support *Support*	Partiel *Partial*	Total *Total*	
						Positions successives avant livraison avion, voir au recto rubrique n° 5 *Successive location before A / C delivery - See front page box number 5*
[illegible]ALES AVIONICS	[illegible] 11-01	Walulop	—	—		Modified to VGM [illegible] to VF11
ESEA	[illegible]	EC15[illegible] 65[illegible]	1[illegible]		15:31	INSTALLED (CRS # 13[illegible])
ESEA	29-10-02	EC155B-6583	188:50		185:46	REMOVED (CRS # 15281

6	Contrôle d'exécution des modifications et des Service Bulletins *Modification and Service Bulletin checking*		
Numéro *No.*	Nature de la modification *Type of modification*	Unité ou société d'exécution *Performing unit or contractor*	Date d'exécution et tampon de contrôle *Date of embodiement and inspection stamp*

5 Unité ou société *Contractor*	Date *Date*	Version et N° Appareil *Aircraft Version and S/N*	Fonctionnement *Operation* Support *Support*	Partiel *Partial*	Total *Total*	Attestation / *Certif*
SEXTANT AV.	06-2000				0h	Fab/*Manuf.* ~~Réceptⁿ/Receipt~~
						Destockage/*Depreservation*
						V.P/*Periodic Overhaul*
EC	09-2000	EC155B 6583	0h		0h	Montage/*Fitting*

GAF-15-04

FICHE MATRICULE / *LOG CARD* *

Fiche n° / *Log card n°* 1

~~Fiche suiveuse Matériel neuf Follow-up Sheet for new equipment~~ *

1 Identification du matériel *Material identification*

Dénomination *Name*	ECRAN 45H VIDEO MISSION SCREEN 45H VIDEO MISSION		
Nomenclature OTAN *NATO Nomenclature*			
Référence EC *EC Part number*	704A45751153		
Référence fabricant *Manufacturer's Part number*	C19209VF11	Code OTAN fabricant *NATO Manuf. code*	F9111
Configuration de livraison *Delivery configuration*		Version *Version*	
Numéro de série *Serial number*	481	Amendements *Amendments*	C
Fabricant *Supplier*	SEXTANT AVIONIQUE	Type *Type*	

2 Marché ou commande *Contract or order*

Référence *Reference*		Date *Date*	
Organisme émetteur *Issuing agency*		N° ou lot *Kit n°*	
Fournisseur *Contractor*		Adresse *Address*	

3 Garantie *Guarantee*

Matériel *Equipment*	Date de livraison *Date of delivery*	Durée garantie de stockage *Guaranteed storage period*	Date de mise en service *Service date*	Durée Garantie de fonctionnement *Guarantee operation period*
Fab/*Man*				

4 Renseignements particuliers *Special information*

Annexe au tableau 4 : *Appendix to table 4*	~~oui~~/non * *yes/no*	Nb de page I/__ *Number of pages*
Inventaire des pièces à Durée de Vie : *Inventory of lifed components*	~~oui~~/non * *yes/no*	Nb de page I/__ *Number of pages*
Fiche Matricule de Transfert : *Transfer sheet*	~~oui~~/non * *yes/no*	Nb de fiche I/__ *Number of sheets*

Limite de fonctionnement *Operating limit*	Limite de vie *Life limit*

* Rayer la mention inutile * Delete as necessary

Aia 102 . 31

5 **Positions successives et opérations d'entretien et de remise en état mineures et majeures**

Successive status and minor/major maintenance and overhaul operations

Unité ou société *Contractor*	Date *Date*	Version et N° Appareil *Aircraft Version and S/N*	Fonctionnement/*Operations* Unité / *Unit*: En Heures / In Hours — Support *Support*	Partiel *Partial*	Total *Total*	Unité / *Unit*: Support *Support*	Partiel *Partial*	Total *Total*	Motif du mouvement (code symptômes) - Travaux effectués - Pièces changées *Reason for transfer (code and symptoms) - Accomplished work - Replaced parts* Positions successives avant livraison avion - Voir recto tableau 5 *Successive status before A/C delivery - See front page of table 5*
ESEA	18-10-01	EC155B1/N6583	15:31		15:31				REMOVED FOR MODIFICATION AS PER AIRBUS SB CHANGE

AWM/024/01 Rev 0

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ALERT TELEX

No. 31A005 R1

AIRCRAFT: EC 155 Civil Versions: B-B1.

Corresponds to drawing: 365PCS 76 023

SUBJECT: INDICATING AND RECORDING SYSTEMS
Landing Gear Extension Inoperative

1. PLANNING INFORMATION

1.A. EFFECTIVITY

Helicopters equipped with ASU board 10 alpha 2 (P/N: SE07451) pre drawing 365PCS 76 023.

1.B. ASSOCIATED REQUIREMENTS

Not applicable.

1.C. REASON

Modify the electric circuit to preclude any risk of the landing gear not extending in the "NORMAL" and "EMERGENCY" extension modes following a failure of ASU board 10 alpha 2.

Revision 1 of this ALERT TELEX which supersedes Revision 0 of this ALERT TELEX:

- Incorporates in paragraph 2.B.2 and in the Appendix the procedure for the functional tests on ASU board 10 alpha 2, to replace the procedure in Subtask 31.60.00.721.004, which will be issued at a later date.
- Replaces the word "MOD" in the title box with the word "drawing".

1.D. DESCRIPTION

One of our operators has reported a case of landing gear not extending in "NORMAL" and "EMERGENCY" extension modes.

The investigation conducted jointly by EUROCOPTER and the equipment manufacturer SEFEE shows that this incident is due to a short-circuit on ASU board 10 alpha 2.
This short-circuit is probably caused by foreign matter located between 2 components of ASU board 10 alpha 2.

Pending a modification to the installation, EUROCOPTER render compliance with paragraph 1.E mandatory.

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1.E. COMPLIANCE

1.E.1. At the works:

- On aircraft: Compliance with drawing 365PCS 76 023, as from August 20, 2002.

- On Spares: Not applicable.

1.E.2. Retrofit action:

On aircraft: By the operator.

1.E.2.a) Within the next 15 flight hours following receipt of this ALERT TELEX:

- Comply with paragraph 2.B.2.a): Cleaning ancillary board 10 alpha 2.

1.E.2.b) Not later than within 1 month from receipt of this ALERT TELEX:

- Comply with paragraphs 2.B.2.b) to 2.B.2.e):
  . Modification to the wiring of connector 10 alpha 2 CP1.
  . Preparing and affixing the "LIMITATION" label.

On Spares: Not applicable.

1.F. APPROVAL

Not applicable.

1.G. MANPOWER

Qualification: 1 electrician.

Time required for compliance with paragraph 2.B.2.a): Approximately 1 hour.
Time required for compliance with paragraphs 2.B.2.b) to 2.B.2.e): Approximately 2 hours and 30 minutes.

1.H. WEIGHT AND BALANCE

Weight: Not applicable.
Moment: Not applicable.

1.I. EFFECT ON ELECTRICAL LOADS

Not applicable.

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1.J. SOFTWARE MODIFICATION EMBODIMENT STATE

Not applicable.

1.K. REFERENCES

Aircraft Maintenance Manual (AMM) Tasks:

. 24-00-00-911	Electrical Power – General Instructions
. 29-00-00-911	Hydraulic Power System – General Instructions
. 29-20-00-481	Hydraulic Assistance on the Ground
. 31-80-00-061	Ancillary System Unit Board – Removal / Installation
. Appendix	Functional Tests on Ancillary System Unit Boards
. 32-30-00-721	Adjustment / Tests (Landing gear retraction and extension tests)

Aircraft Maintenance Manual (AMM) Subtasks:

. 32-30-00-721-002 of Task 32-30-00-721 Helicopter Jacking-Up
. 07-10-00-581-002 of Task 07-10-00-581

1.L. OTHER DOCUMENTS CONCERNED

Not applicable.

1.M. INTERCHANGEABILITY OR MIXABILITY OF PARTS

Not applicable.

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2. ACCOMPLISHMENT INSTRUCTIONS

2.A. GENERAL

The modification to the electric wiring, defined in paragraph 2.B.2.b), is a temporary solution (two connectors and one diode) to ensure electrical connection between the power switch (that supplies the electric pump) and wiring 10 alpha 2 CP1, pending a modification to the system.

The purpose of this modification is:

- to supply the "EMERGENCY" control system of the landing gear installation separately from the "NORMAL" control system,
- to ensure a continuous ground at terminal 4 (coil negative) of power switch 19G, in "EMERGENCY" mode and in "TEST" mode,
- to inhibit an induction current between the connection terminals of the excitation coil of the power switch of the ancillary electric pump.

NOTE

This modification to the electric wiring leads to inhibition of the protective thermal switch of the electric pump. Limitations related to the use of the electric pump on the ground are listed in paragraph 2.D of this ALERT TELEX.

- Comply with the following general instructions:

  . Electric Power in Task 24.00.00.911,
  . Hydraulic Power in Task 29.00.00.911.

2.B. OPERATIONAL PROCEDURE

2.B.1. Preliminary steps (Figure 1)

- Open radome (A) to gain access as per Detail B to ancillary system unit 10 alpha (C), located on the right side of the helicopter, beneath the horizontal support member of electrical master box 4 alpha.

2.B.2. Procedures

a) Cleaning ancillary board 10 alpha 2 (Figure 1)

- Comply with Task 31-80-00-061: Removal of Ancillary Board 10 Alpha 2 from Ancillary System Unit 10 Alpha.

- Blast ancillary board 10 alpha 2 with dry, uncontaminated (filtered) air, using a suitable equipment.

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NOTE

This action will eliminate any chips or foreign matter, which can cause short-circuits on the various electrical systems that are related to the removed board.

- Carry out:

.Task 31-80-00-061: Insertion of Ancillary Board 10 Alpha 2 into Ancillary System Unit 10 Alpha.
. The functional tests on (ASU) ancillary boards 10 alpha 2 in compliance with the Appendix enclosed.

b) Modification to the electric wiring of connector 10 alpha 2 CP1

Tools required for the modification:

. Cutting pliers
. Metric stripping pliers, fitted with blue handles
. Crimping pliers M 22520/2-01
. Locating tool "2-02" for crimping contacts (2)
. Locating tool "2-06" for crimping contact (6)
. Pliers for cable clamps, fitted with black handles
. Extraction/insertion tools:
. Red/white plastic extraction/insertion tool for contacts (2), P/N: M 15570-22-1,
. Green/white extraction/insertion tool for contact (6), P/N: M 81969/14-01.
. 3.9 mm diameter Allen wrench.

Material required for the modification:

- Refer to paragraph 3.C.

Description of Figures 1 to 4

Figure 1: Drawing showing the location of ancillary system unit 10 alpha and overhead panel 12 alpha on aircraft.

Figure 2: Wiring diagram of connector 10 alpha 2 CP1 pre and post modification 365PCS 76 023.

Figure 3: Photo depicting the modification embodied to the wiring of connector 10 alpha 2 CP1:
. The upper picture shows the recommended distance between the location of the modification components relative to the rear of connector 10 alpha 2 CP1.
. The lower picture shows the routing principle for the wires that are to be connected post modification 365PCS 76 023 and identifies the components on wire harness 10 alpha 2 CP1.

Figure 4: Drawing:
. shows the location of the "limitation" label on the overhead panel in the cockpit,
. specifies the dimensions of the "limitation" label and the text to be marked on it.

As per Figure 1

- Unplug connector 10 alpha 2 (D) from fixed connector 10 alpha 2 (E).
- Remove:
  . Identification marker 10 alpha 2 CP1 and retain it.
  . Cable clamp (F) that retains the harness and green protecting sheath (G) on clamping block (H), at the rear of connector 10 alpha 2 (D).

- Remove the cable clamps which retain the green protecting sheaths (G) and (J) to move the wires (their electrical markers are listed below) aft of connector 10 ALPHA 2 CP1 and to remove them from wire harness 10 alpha 2 CP1, at intersection (P) of protecting sheaths (G) and (J) (as per Figures 1 and 3).

As per Figure 2 - PRE MODIFICATION:

- Unplug the following wires from connector 10 alpha 2 CP 1:

. Wire 1GA24NE from terminal 12,
. Wire 1GA6G from terminal 15,
. Wire 1GA18E from terminal 16,
. Wire 1GA6F from terminal 17,
. Wire 1GA13E from terminal 18.

NOTE 1

Remove and discard contacts (K). To this end, cut the five wires, listed above, as close as possible to contacts (K) so that these wires need not be replaced and can be re-connected to connector 10 alpha 2 CP1 when the future modification is embodied.

NOTE 2

As per Figure 3

Before you make the connections of wires 1GA6F, 1GA6G, 1GA13E, 1GA18E and 1GA24NE in accordance with the procedure specified below, coil wire overlength from intersection (P) of protecting sheaths (G) and (J), and route these wires on protecting sheath (G).

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As per Figure 2 POST MODIFICATION, and Figure 3

- Blank the ports of unequipped contacts 15, 16, 17 and 18 of connector 10 alpha 2 CP1 with blanking plugs (10).

- Take wire (1) and cut it into three 140 mm long pieces.
- Strip both ends of each wire (1).

NOTE 3

This wire (1) has no electrical identification marker engraved on the insulator. Consequently, as per Figure 2 post modification, wire (1) will be identified by REP5378A, REP5379A and REP5380A, in the text below, according to the connections to be made.

- Strip wires 1GA6F, 1GA6G, 1GA13E, 1GA18E and 1GA24NE.

- Crimp contact (2) to 1GA6F, 1GA6G, 1GA13E, 1GA18E, 1GA24NE, and to both ends of the two wires (1) REP5378A and REP5379A.

Preparation of the connection of wire (1) REP5380A:

- Crimp contact (2) to one end of wire (1) REP5380A and contact (6) to the other end.

As per Figure 2 - POST MODIFICATION and Figure 3

- Connect wires: 1GA6F, 1GA6G, 1GA13E, 1GA18E, 1GA24NE, REP5378A, REP5379A REP5380A.

NOTE 4

The pictures on Figure 3 show the connection loops (R) of connectors (3) and (5), and of diode (4).

• Connector 10 alpha 2 MD1 (3):

- REP5380A (1) terminal 1A
- REP5378A (1) terminal 1B
- 1GA13E terminal 2A
- 1GA24NE terminal 2B

• Connector 10 alpha 2 MD2 (5):

- 1GA6F terminal 1A
- 1GA6G terminal 1B
- 1GA18E terminal 2A
- REP5379A (1) terminal 2B

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- Diode connector (4):

  - Be careful to ensure the ANODE / CATHODE connections in accordance with the requirements.
    - REP5378A (1) ANODE,
    - REP5379A (1) CATHODE.

- Connector 10 alpha 2 CP1:

  - REP5380A (1) terminal 12.

c) Reconditioning of wire harness 10 alpha 2 CP1 (Figures 1 and 3)

As per Figure 3

- Correctly position protecting sheaths (G) and (J) on wire harness 10 alpha 2 CP1.
- Using clamps (7), hold sheaths (G) and (J) on the wires that make up harness 10 alpha 2 CP1.

As per Figure 1 and Figure 3

- Using a clamp (7), hold the wire harness on clamping block (H), which is located at the rear of connector 10 alpha 2 CP1 (D).
- Affix identification marker 10 alpha 2 CP1 (retained earlier).

As per Figure 3

- Attach connectors (3) and (5) and diode (4) to wire harness 10 alpha 2 CP1, using spacers (8) and clamps (7) and (9).

As per Figure 1

- Plug connector 10 alpha 2 (D) into connector 10 alpha 2 (E).
- Secure/screw connector 10 alpha 2 (D) using an Allen wrench.

d) Functional test of the landing gear installation

- Comply with the instructions specified in paragraph 2.D.

- Comply with:

. Task 32-30-00-721: Electric Pump Testing in "NORMAL" and "EMERGENCY Modes.
. Task 29-20-00-481: Electric Pump Testing in "TEST" Mode.

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e) Preparing and Affixing the "LIMITATION" label, Figures 1 and 4

1) Producing a "LIMITATION" label (11), as per Figure 4

- Make a self-adhesive label:
  . In accordance with required dimensions,
  . text in white letters on red background.

2) Affixing the "LIMITATION" label, as per Figures 1 and 4

- Affix the "LIMITATION" label (11) to overhead panel 12 alpha in the cockpit, as shown on Figure 4.

2.B.3. Final steps, Figure 1

- Lower the helicopter onto its wheels as per Subtask 07-10-00-581-002 of Task 07-10-00-581.
- Close radome (A).

2.C. IDENTIFICATION

- Record compliance with Revision 1 of this ALERT TELEX in the aircraft documents.

2.D. OPERATING AND MAINTENANCE INSTRUCTIONS

Reminder: The modification to the electric wiring, covered by Revision 1 of this ALERT TELEX, leads to inhibition of the protective thermal switch of the electric pump.

- Comply with the following on-ground operation time restrictions that are applicable to the electric pump:

1) The continuous operation time of the pump must not exceed 10 minutes.

2) Allow the electric pump to cool for 15 minutes, when the continuous operation time of the pump is 1 to 5 minutes.

3) Allow the electric pump to cool for at least 30 minutes, when the continuous operation time of the pump is 5 to 10 minutes.

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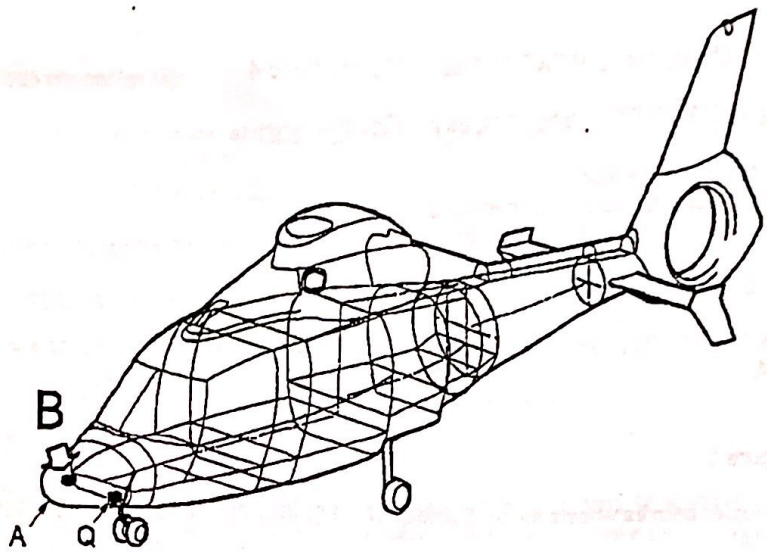

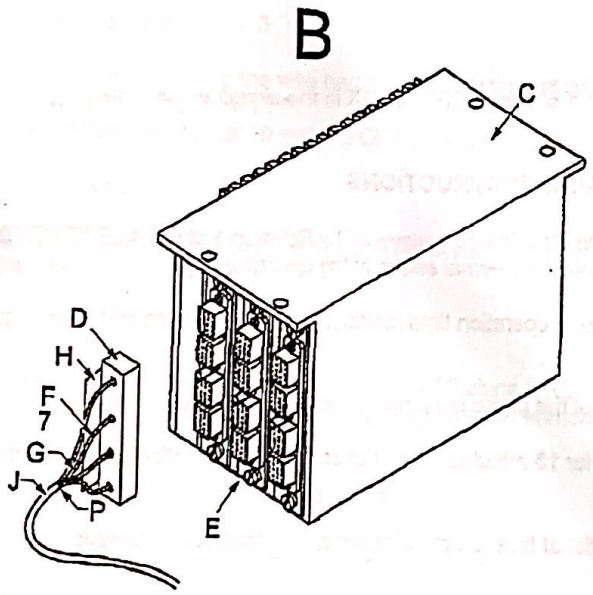

FIGURE 1

Location of ancillary system unit 10 alpha and overhead panel 12 alpha

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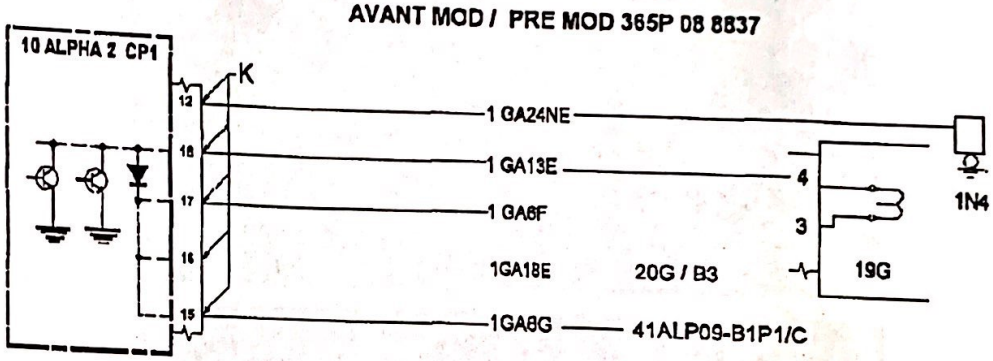
AVANT MOD / PRE MOD 365P 08 8837



APRES MOD / POST MOD 365P 08 8837

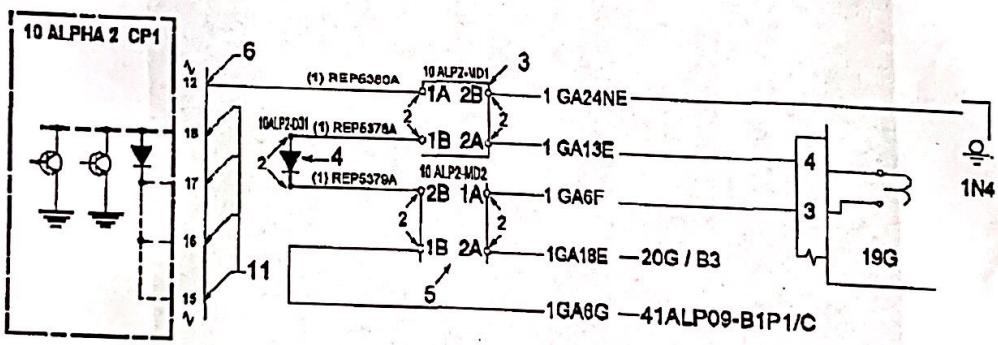


FIGURE 2

Wiring of connector10 alpha 2 CP1

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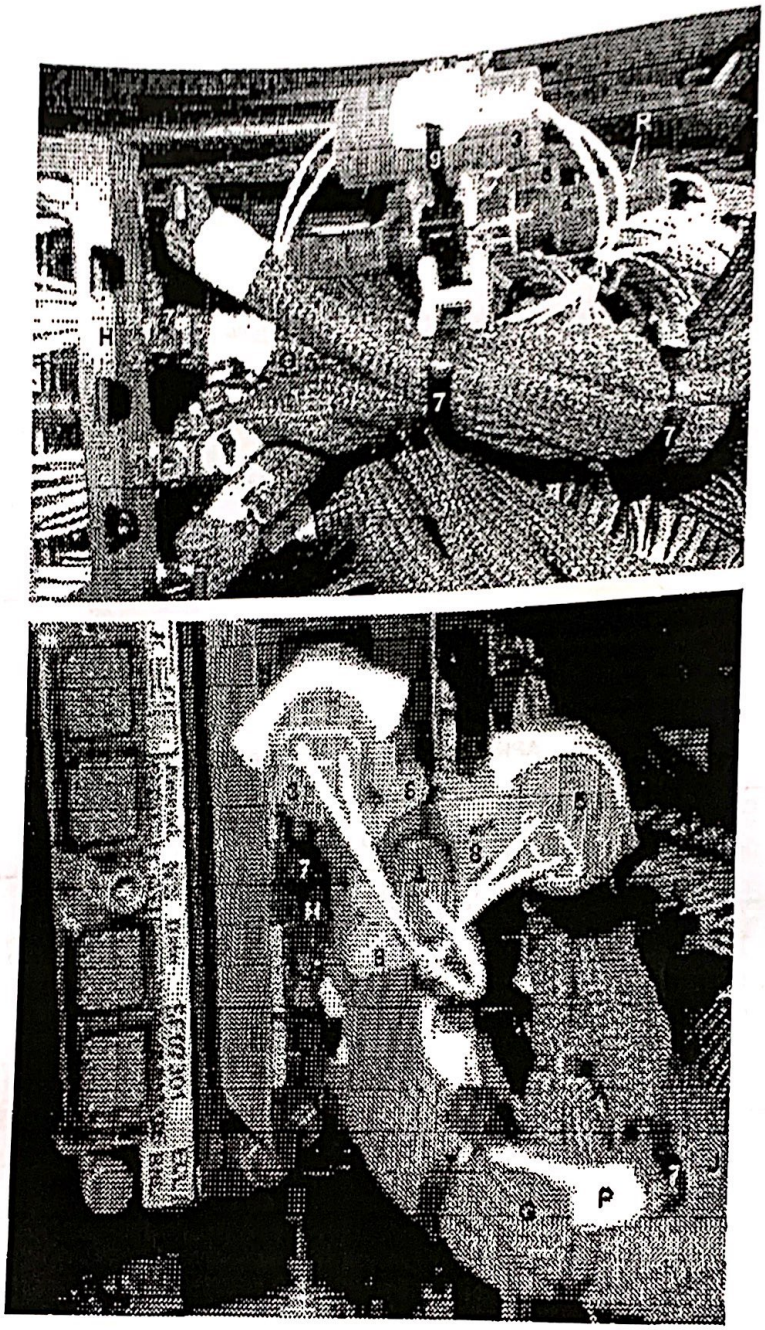

FIGURE 3

Representation of modification 365PCS 76 023

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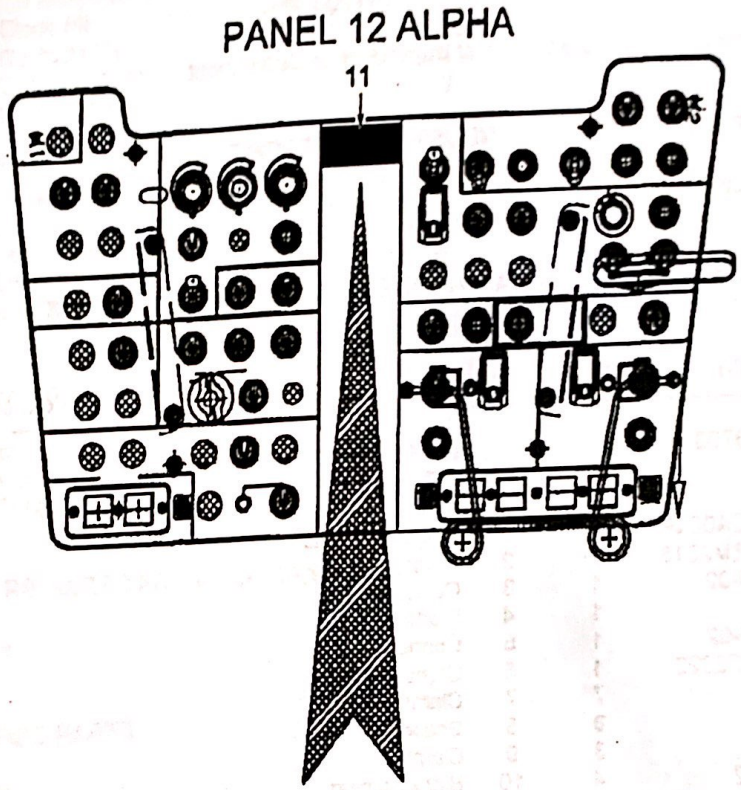

Location of the "LIMITATION" label

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3. MATERIAL INFORMATION

3.A. MATERIAL – COST - AVAILABILITY

For all information contact the Customer Support Sales Department.

3.B. INFORMATION CONCERNING INDUSTRIAL SUPPORT

Not applicable.

3.C. MATERIAL REQUIRED FOR EACH AIRCRAFT, ENGINE/COMPONENTS

New P/N	Qty	Item	Key Word	Former P/N	Instructions Disposition
365P 0888370071			LANDING GEAR MODIFICATION KIT		
EN2666-005A004P	1m	1	Single-conductor wire		
EN3155-016M2018	10	2	Crimp contact		
001119-224-02	1	3	Connector		
E0536-02	1	4	Diode		
001119-224-02	1	5	Connector		
EN3155003F2222	1	6	Crimp contact		
E0043-1C0	7	7	Clamp		
E0688-01	3	8	Spacer		
E0043-4C0	3	9	Clamp		
E0775.22.02	4	10	Blanking plug		

NOTE 1

The material listed above can be supplied locally, from a Supplier of your choice. Should you have difficulty in finding this material locally, you can order the landing gear modification kit 365P 0888370071 from EUROCOPTER (available as from September 20, 2002).

NOTE 2

The diode (4) required for the modification to the wiring of connector 10 alpha 2 CP1 (D) must have the following technical features:

- *Rectifier diode F 102 in accordance with NF C 93-462 standard.*

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For example:

Diode (4) is available:

. *from equipment manufacturer* AIR-LB:
. Diode (4) P/N: 001119 704 02
. Contacts (2) P/N: 001104 210 02
. Crimping pliers P/N: M 22520/2-01
. Locating tool P/N: M 22520/1-02

. *from equipment manufacturer* DEUTSCH:
. *Diode (4)* P/N: BJE 80
. Contacts (2) P/N: 1841-1-5620
. Insertion/extraction tool for contacts (2) P/N: M 15570-22-1
. Crimping pliers P/N: M 22520/2-01
. Locating tool P/N: K127-2

Connectors (3) and (5) are available:

. *from equipment manufacturer* AIR-LB:
. Connectors (3) and (5) P/N: 001119 224 02
. Contacts (2) P/N: 001119 210 02

3.D. MATERIAL REQUIRED FOR EACH SPARE PART

Not applicable.

3.E. RE-IDENTIFIED PARTS

Not applicable.

3.F. TOOLING – COST – AVAILABILITY

Not applicable.

3.G. PROCUREMENT CONDITIONS

Order the required quantity

from

EUROCOPTER
Etablissement de Marignane
Direction VENTES Service Client
S.C.
13725 MARIGNANE CEDEX
FRANCE

NOTE 1

For ALERT TELEXES, order by:
Telex: HELICOP 410 969F
Fax: +33 (0)4 42 85 99 96

NOTE 2

On the purchase order, please specify the mode of transport, the destination and the serial numbers of the aircraft to be modified.

4. APPENDIX

Procedure for functional tests on ASU board (10 ALPHA 2).

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T.F.S. No. 00000091 dated September 19, 2002
EUROCOPTER - MARIGNANE - TLX 425067

Functional Tests on ASU Board (10 ALPHA 2)

A) Functional Test of the Baggage Compartment Fire and Overheating Detection System

(PRE MOD 39B96 baggage compartment fire extinguishing system, refer to the aircraft individual inspection log book (RIC).

1. Configuration:

(a) Disengage circuit breaker 4ALPHA D1 (EXT.LT.CT.HOLD O/H).

(b) Check that 3ALPHA D1 (EXT.LT.CT.HOLD O/H) is engaged .

(c) Reset the "red warning light" repeater, if necessary.

(d) Check that the "C FIRE" warning light on the "red warning light" caution advisory panel is not lit.

(e) Check that there is no "OVERHEAT" warning message on the CAD.

2.Baggage Compartment Fire Detection Procedure (T>160°C):

(a) On the overhead panel, press and hold the "SERVO + CARGO" pushbutton and make sure that the "C FIRE" warning light comes on, and that the "red warning light" repeater flashes.

(b) Stop the test and make sure the "C FIRE" warning light goes out.

(c) Reset the "red warning light" repeater.

3. Baggage Compartment Overheating (T>110°C) Detection Procedure:

(a) Engage circuit breaker 4 ALPHA D1 (EXT.LT.CT.HOLD O/H) and disengage 3 ALPHA D1 (EXT.LT.CT.HOLD O/H).

(b) On the overhead panel, press and hold the "SERVO + CARGO" pushbutton and make sure the "OVERHEAT" warning light comes on.

(c) Stop the test and make sure the "OVERHEAT" warning light goes out.

(d) Engage circuit breaker 3 ALPHA D1 (EXT.LT.CT.HOLD O/H).

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EUROCOPTER - MARIGNANE - TLX 42506F

APPENDIX

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EUROCOPTER - MARIGNANE - TLX 42506F

(POST MOD 39B96 baggage compartment fire extinguishing system, refer to the aircraft individual inspection log book (RIC).

1. Configuration:

   (a) Engage circuit breakers 4ALPHA D1 (EXT.LT.CT.HOLD O/H) et 4ALPHA C1 (WARN EXT).

   (b) Disengage circuit breaker 5ALPHA A3 (EXT.LT.LC O/H).

   (c) Reset the "red warning light" repeater, if necessary.

   (d) Make sure the "FAIL" warning light on the percussion pushbuttons on the overhead panel are not lit.

   (e) Check that the "C FIRE" warning light on the "red warning light" caution advisory panel is not lit.

   (f) Check that there is no "OVERHEAT" warning message on the CAD.

2. Baggage Compartment Overheating (T>110°C) Detection Procedure:

   (a) Remove the fuse 36ALPHA1 F3 and make sure the "FAIL" warning lights on the percussion pushbuttons of the "LC FIRE" window, on the overhead panel, come on.

   (b) Re-install the fuse 36ALPHA1 F3 and make sure the "FAIL" warning lights on the pushbuttons of the "LC FIRE" window, on the overhead panel, go out.

   (c) Press and hold the "O/H" switch of the "LC FIRE" window , on the overhead panel, in the "FIRE" position and make sure the "OVERHEAT" message appears on the CAD.

   (d) Release the pushbutton and make sure the « OVERHEAT » message disappears on the CAD.

3. Baggage Compartment Fire (T>160°C) Detection Procedure:

   (a) Engage circuit breakers 5ALPHA F6 (WARN EXT) and 5ALPHA A3 (EXT.LT.LC O/H).

   (b) Disengage circuit breakers 4ALPHA D1 (EXT.LT.CT.HOLD O/H) and 4ALPHA C1 (WARN EXT).

   (c) Make sure the "CAR FIRE" warning light on the "red warning light" caution advisory panel is not lit.

   (d) Press and hold the « O/H » switch of the "LC FIRE" window , on the overhead panel, in the "FIRE" position, and make sure that the "CAR FIRE" warning light on the "red warning light" caution advisory panel comes on, that the red " FIRE" warning lights in the "LC FIRE" window, on the pushbuttons, come on and that the "red warning light" repeaters are flashing.

   (e) Press and hold the "O/H" switch of the "LC FIRE" window, on the overhead panel, in the "FAIL" position, and make sure that the amber "FAIL" lights in the "LC FIRE" window, on the pushbuttons, come on.

   (f) Engage circuit breakers 4ALPHA D1 (EXT.LT.CT.HOLD O/H) and 4ALPHA C1 (WARN EXT).

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T.F.S. No. 00000091 dated September 19, 2002
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B) Test of the Landing Light Control System

Depending on customizations:

Conduct the test on the swivel landing lights, as per AMM Task 33.43.02.721

or

Tasks 33.43.03.721 and 33.43.04.721

C) Test of the Landing Gear Retraction/Extension System in Normal and Emergency Extension Modes

Conduct the test on the landing gear retraction/extension systems as per AMM Task 32.30.00.721.

NOTE 1:

The test of the "Landing Gear Not Extended" warning function for IAS < 55 kt, specified in Task 32.30.00.721, is not required as this function is not managed by the 10ALPHA 2 board.

NOTE 2

During the test of the emergency landing gear extension, the L/G PUMP indicator light appears on the CAD at the end of the maneuver, and not as soon as the « L/G PUMP » selector is set to the "EMERG" position, as specified in Task 32.30.00.721. This task will be revised at a later date.

D) Test of the Hydraulic Indicating Systems

Conduct the test of the hydraulic indicating systems as per AMM Task 67.30.01.221.

APPENDIX

AWM/024/01 #20.

GSAC

AIRWORTHINESS DIRECTIVE

released by DIRECTION GENERALE DE L'AVIATION CIVILE

Inspection and/or modifications described below are mandatory. No person may operate a product to which this Airworthiness Directive applies except in accordance with the requirements of this Airworthiness Directive.

Translation of 'Consigne de Navigabilité' ref. : 2002-515(A)

In case of any difficulty, reference should be made to the French original issue.

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EC 155 helicopters

Landing gear indicating system (ATA 31)

1. EFFECTIVITY

EC 155 helicopters, versions B and B1, equipped with ASU board 10 alpha 2, P/N SE07451, delivered before August 20, 2002.

2. REASON

This Airworthiness Directive (AD) is issued following a case of landing gear not extending in "NORMAL" and "EMERGENCY" extension modes due to a short-circuit between two components of ASU board 10 alpha 2.

3. MANDATORY ACTIONS AND COMPLIANCE TIME

The following measures are rendered mandatory :

3.1.

Not later than within the next 15 flight hours following the effective date of this AD (unless already accomplished), clean ASU board 10 alpha 2 in accordance with the instructions described in paragraph 2.B.2.a of referenced EUROCOPTER EC 155 Alert Telex No. 31A005 R1.

3.2.

Not later than within one month following the effective date of this AD (unless already accomplished), modify the electric wiring, affix a limitation label and comply with the operating and maintenance instructions described in paragraphs 2.B.2.b to 2.B.2.e of the referenced Alert Telex.

REF. :
EUROCOPTER EC 155 Alert Telex No. 31A005 R1.

EFFECTIVE DATE :

On receipt from OCTOBER 02, 2002

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