AIRBUS

EC 155

AIRWORTHINESS LIMITATIONS SECTION (ALS)

EC 155
B1

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IMPORTANT NOTE

The practical value of this manual depends entirely upon it being updated correctly by the operator.

The successive revisions should be recorded on the relevant page of the manual.

RECORD OF REVISIONS

Rev. No.	Issue Date	Inserted Date	Inserted Name

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APPROVAL

1 EASA - Approval

The Airworthiness Limitations Section is approved and any change must also be approved by The EASA

Revision	04-00 LOEP Identification	Approval Date	EASA nb
REV 000	2003.07.03	EASA : 2003.10.03	-
REV 001	2004.01.02	EASA : 2004.05.03	2004-4665
REV 002	2004.06.10	EASA : 2004.11.17	2004-11069
REV 003	2005.01.20	EASA : 2005.04.07	2005-3141
REV 004	2005.06.09	EASA : 2005.09.27	R.A 01040
REV 005	2005.12.08	EASA : 2006.02.24	R.A 01090
REV 006	2006.04.07	EASA : 2006.07.24	R.C 01832
REV 007	2006.11.14	EASA : 2007.03.02	R.C 02183
REV 008	2007.03.23	EASA : 2007.04.23	R.C 02154
REV 009	2007.07.20	EASA : 2007.08.01	R.C 02545
REV 010	2008.06.27	EASA : 2008.10.09	R.C 03064
REV 011	2009.01.08	EASA : 2009.04.31	R.C 03380
REV 012	2010.04.08	EASA : 2010.06.17	10030472
REV 013	2010.07.29	EASA : 2010.12.07	10032897
REV 014	2011.06.09	EASA : 2012.01.03	10037733.R1

Revision	04-00 LOEDU Identification	Approval Date	EASA nb
New REV 000	2012.11.08	EASA : 2012.10.22	10041894
New REV 001	2013.09.12	EASA : 2014.01.20	10047763
New REV 002	2014.05.29	EASA : 2014.11.10	10051101

Revision	LOEDU Identification	N° TFSAC	Approval date	N°
New REV 003	2016.09.12	13.155.0003.a 14.155.0001.a 15.155.0002.a 15.155.0005.a 16.155.0001.a	2015.10.01 2015.11.04 2015.10.15 2015.11.25 2016.05.06	10054937 10055307 REV.1 10055158 10055640 10057793
New REV 004	2017.09.11	16.155.0003.b 16.155.0004.b	2016.12.12 2016.11.30	EASA: 10060423 EASA: 10060209 REV. 1
New REV 005	2018.09.10	17.155.0002.a	2018.06.19	AH DOA: DO/2018/034
New REV 006	2019.09.16	15.155.0001.a 18.155.0001.a 19.155.0003.a	2018.11.20 2018.06.22 2019.09.20	10067575 10065949 AH DOA : DO/2019/056
New REV 007	2021.01.22	19.155.0001.a 19.155.0004.a 20.155.0001.a	2020.05.18 2020.05.29 2021.01.12	10073297 AH DOA: DO/2017/106 AH DOA: DO/2017/024
New REV 008	2022.02.01	20.155.0002.a 20.155.0003.a 21.155.0001.a	2020.10.26 2020.10.27 2021.06.18	10074654 AH DOA : DO/2020/112 10076719

The technical content of these revisions are approved under DOA N°EASA.21J.700 authority. This approval means that the revisions have been prepared with already EASA approved data and minor changes approved under DOA N°EASA.21J.700. The revisions are issued under DOA procedure.

The list of chapter 04 effective pages is given in the "LOEDU" which is identified by 2022.02.01.

The "LOEDU" identification does not correspond to the approval date, it is a documentation handling code provided at the bottom of all pages.

End of the Document Unit

LOEDU

1 Update

Refer to the **"Update of the ALS" Document Unit**

2 "Type" Column Code

N-New, to be inserted
R-Revised, to be replaced
D-To be removed from the manual
- (No code) Unchanged with respect to the previous issue

D.U.	Type	Date	D.U.	Type	Date
TITLE	N	2022.02.01	04-10-00 ATA 53	-	2013.09.12
APPROVAL	N	2022.02.01	04-10-00 ATA 62	-	2021.01.22
LOEDU	N	2022.02.01	04-10-00 ATA 63	R	2022.02.01
TABLE OF CONTENTS	N	2022.02.01	04-10-00 ATA 64	-	2016.09.12
HIGHLIGHTS OF THE REVISION	N	2022.02.01	04-10-00 ATA 65	-	2018.09.10
			04-10-00 ATA 67	-	2019.09.16
GLOSSARY	-	2016.09.12	04-20-00 ATA 25	-	2017.09.11
STRUCTURE OF THE ALS	-	2019.09.16	04-20-00 ATA 26	-	2012.11.08
			04-20-00 ATA 53	-	2013.09.12
UPDATE OF THE ALS	-	2013.09.12	04-20-00 ATA 62	-	2016.09.12
BREAKDOWN OF THE ALS	-	2019.09.16	04-20-00 ATA 63	R	2022.02.01
USE OF THE ALS	R	2022.02.01	04-20-00 ATA 64	-	2017.09.11
SB INCORPORATED	-	2019.09.16	04-20-00 ATA 65	-	2013.09.12

End of the Document Unit

TABLE OF CONTENTS

TITLE

TITLE

End of the Document Unit

HIGHLIGHTS OF THE REVISION

1 GENERAL

- Check that the content of the sections is in accordance with the List Of Effective Document Units (LOEDU).
- Return the acknowledgment card.

2 OUTLINE OF THE REVISION

The revision is codified as follows:

- Revision 008 2022.02.01

3 DETAILED DESCRIPTION

3.1 Section USE OF THE ALS

§ 5.5.4 “Specific cases” integrated

3.2 Section 04.10.00

63-10 ENGINE/MGB COUPLING

- 63/10/00/000/000/100 — 3700 FH
  Front gimbal joint equipped
  New task integrated
- 63/10/00/000/000/105 — 4300 FH
  Rear gimbal joint equipped
  New task integrated
- 63/10/00/000/000/110 — 11000 FH
  Gimbal joint ring
  New task integrated
- 63/10/00/000/000/115 — 1100 FH
  Gimbal joint pin
  New task integrated
- 63/10/00/000/000/200 — 40000 FH
  Gimbal joint pin
  New task integrated

63-20 MAIN GEARBOX

- 63/20/00/000/000/300 — 40000 FH
  Front right housing equipped
  New task integrated
- 63/20/00/000/000/310 — 40000 FH
  Rear right housing equipped
  New task integrated
- 63/20/00/000/000/400 — 40000 FH
  Main bevel wheel
  New task integrated
- 63/20/00/000/000/450 — 9647 FH
  Fixed rear output flange
  New task integrated

- 63/20/00/000/000/600
  Main housing
  New task integrated
  25000 FH // 125000 TC
- 63/20/00/000/000/605
  Right input housing
  New task integrated
  40000 FH // 200000 TC
- 63/20/00/000/000/610
  Left input housing
  New task integrated
  40000 FH // 200000 TC
- 63/20/00/000/000/615
  Bevel wheel support equipped
  New task integrated
  40000 FH
- 63/20/00/000/000/620
  Main bevel wheel
  New task integrated
  1100 FH
- 63/20/00/000/000/625
  Tail output wheel
  New task integrated
  40000 FH
- 63/20/00/000/000/630
  Sun gear
  New task integrated
  40000 FH
- 63/20/00/000/000/635
  Servocontrol fitting
  New task integrated
  5330 FH
- 63/20/00/000/000/640
  Servocontrol axis and link servocontrol fitting to main housing
  New task integrated
  40000 FH
- 63/20/00/000/000/645
  Rear output bevel gear
  New task integrated
  25800 FH
- 63/20/00/000/000/650
  Fixed rear output flange
  New task integrated
  1638 FH
- 63/20/00/000/000/655
  Sliding rear output flange
  New task integrated
  15600 FH
- 63/20/00/000/000/660
  Tail output flexible coupling
  New task integrated
  28800 FH
- 63/20/00/000/000/665
  Planet gear
  New task integrated
  40000 FH
- 63/20/00/000/000/670
  Fixed gear
  New task integrated
  40000 FH
- 63/20/00/000/000/675
  Planet carrier
  New task integrated
  9060 FH // 45300 TC

- 63/20/00/000/000/680 — 3350 FH
  Upper cover
  New task integrated
- 63/20/00/000/000/685 — 40000 FH
  Right seat hydraulic pump
  New task integrated
- 63/20/00/000/000/690 — 40000 FH
  Left seat hydraulic pump
  New task integrated

3.3 Section 04.20.00

63-10 ENGINE/MGB COUPLING

- 63/10/00/000/000/010 — 15 FH
  Flexible coupling (MGB side)
  Task deleted
- 63/10/00/000/000/120 — 800 FH
  Front half gimbal joints
  New task integrated
- 63/10/00/000/000/125 — 540 FH
  Rear half gimbal joints
  New task integrated
- 63/10/00/000/000/130 — 600 FH
  Gimbal joint ring
  New task integrated
- 63/10/00/000/000/135 — 100 FH
  Link front gimbal joint to input housing
  New task integrated
- 63/10/00/000/000/140 — 50 FH
  Link torque tube to rear gimbal joint
  New task integrated

63-20 MAIN GEARBOX

- 63/20/00/000/000/000 — 10 FH
  MGB – Electrical or non-electrical magnetic plug
  Task deleted
- 63/20/00/000/000/001 — 10 FH
  MGB - Chip detector
  New task integrated
- 63/20/00/000/000/055 — 50 FH
  Planet gear carrier
  Task deleted
- 63/20/00/000/000/056 — 50 FH
  Planet gear carrier
  New task integrated
- 63/20/00/000/000/548 — 15 FH
  MGB bottom
  New task integrated
- 63/20/00/000/000/549 — 15 FH
  MGB bottom
  Task deleted

- 63/20/00/000/000/700 MGB - Chip detector New task integrated	100 FH
- 63/20/00/000/000/710 MGB plate (Bottom plate) New task integrated	40 FH
- 63/20/00/000/000/720 Upper cover New task integrated	447 FH
- 63/20/00/000/000/730 Main housing New task integrated	2722 FH // 13600 TC
- 63/20/00/000/000/740 Servocontrol axis New task integrated	1620 FH
- 63/20/00/000/000/750 Main actuator fitting New task integrated	280 FH
- 63/20/00/000/000/760 Servocontrol axis New task integrated	8130 FH
- 63/20/00/000/000/770 MGB - Oil filter New task integrated	50 FH
- 63/20/00/000/000/780 Input flexible coupling (MGB side) New task integrated	15 FH
- 63/20/00/000/000/790 Tail output flexible coupling (MGB side) New task integrated	15 FH
- 63/20/00/000/000/800 MGB housing links New task integrated	1425 FH // 7125 TC
- 63/20/00/000/000/810 Link sliding rear output flange to flexible coupling New task integrated	6100 FH
- 63/20/00/000/000/820 All links New task integrated	1100 FH
- 63/20/00/000/000/830 Thermocontact New task integrated	1100 FH
- 63/20/00/000/000/840 Thermocontact New task integrated	1100 FH

- 63/20/00/000/000/850 — 2180 FH
  Upper cover
  New task integrated
- 63/20/00/000/000/860 — 1639 FH
  Fixed rear output flange
  New task integrated

End of the Document Unit

GLOSSARY

(-) :	Not indicated
(P)	Provisional limitation
& :	And
// :	Or
§ :	Paragraph
ALL MP/N :	All Manufacturer Part Numbers
ALS :	Airworthiness Limitations Section
AMM :	Aircraft Maintenance Manual
ATA	Air Transport Association
CHK :	CHecK
D :	Day
DU:	Document Unit
EASA :	European Aviation Safety Agency
EXC :	EXCept
FH :	Flight Hour
FM :	Log card
FS :	Frequency Start
HC :	Hoist Cycle
LC :	Landing Cycle
LOEDU :	List Of Effective Document Units
M :	Month
MGB :	Main Gear Box
MP/N :	Manufacturer Part/Number
MRH :	Main Rotor Hub
MSM :	Master Servicing Manual (Maintenance Program)
OC :	On Condition
OEM :	Original Equipment Manufacturer
OPT :	OPTion
PN :	Part Number
PO :	Perform Once
POST :	After (POST MOD, POST SB)
PRE :	Before (PRE MOD, PRE SB)
RC :	Roping Cycle
SB :	Service Bulletin
SBV :	Vendor's Service Bulletin
SC :	Sling Cycle
SLL :	Service Life Limit
S/N :	Serial Number
TC :	Torque Cycle
TGB:	Tail Gear Box
TL :	Type Limit

TRH :	Tail Rotor Hub
TSI :	Time Since Installation
TSM :	Time Since Manufacture
WC :	Work Card
Y :	Year

End of the Document Unit

STRUCTURE OF THE ALS

1 GENERAL STRUCTURE OF THE ALS

The Airworthiness Limitations Section (ALS) is broken down into two parts.

The first part comprises general information which explains how the ALS operates and how it must be used.

It is composed of the following DUs:

- Glossary,
- Structure of the ALS,
- Update of the ALS,
- Breakdown of the ALS,
- Use of the ALS.

The second part which defines the limits comprises the following sections:

- 04-10-00, which specifies the components subject to a Service Life Limit (SLL).
- 04-20-00, which specifies the mandatory airworthiness inspections (CHK) and provides various specific information to be performed on the helicopter, periodically or a limited number of times, regardless of the operating conditions.

2 BREAKDOWN OF THE ALS

The various parts of the ALS are presented in the form of Document Units (DUs).

2.1 Identification of the DUs

Each DU is unique and is identified by the following:

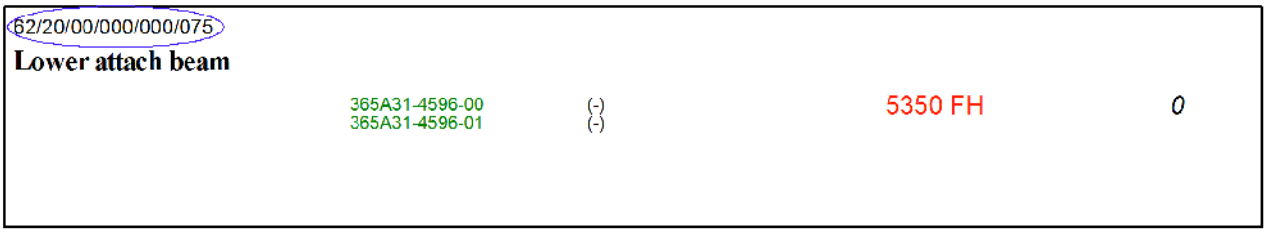
- the title of the manual at the top right hand side of the page,
- the effectivity (helicopter type and version) marked at the bottom left hand side of the page,
- the date code of the DU at the bottom left hand side of the page below the effectivity (format: year.month.day),
- the number or title of the DU at the bottom right hand side of the page,
- the ATA number at the bottom of the page only for the sections 04-10-00 and 04-20-00,
- the page number at the bottom right hand side of the page.

The end of each Document Unit is identified by "End of the Document Unit" on the last page.

2.2 Task codes

The DUs in sections 04-10-00 and 04-20-00 comprise maintenance tasks. These tasks are identified by a unique code specific to Airbus, comprising a group of 15 characters.

Example :



The first four characters (**62/20**/00/000/000/075) correspond to the breakdown into ATA system/sub-system.

The fifth and sixth characters (62/20/**00**/000/000/075) are used to regroup an ATA system comprising several subjects into individual subjects.

The seventh to fifteenth characters (62/20/00/**000/000/075**), according to the ALS section, are used to organize the tasks in an ATA system/sub-system/subject.

2.3 List Of Effective Document Units (LOEDU)

The LOEDU specifies the following information, for each DU, in a table:

- the Document Unit: this column specifies the number of the section, the ATA number or the title of the DU (example: 04-20-00, ATA 62).
- the type: this column is coded with the letter "R" if the DU is revised, "D" if the DU is deleted, "N" if the DU is new, or has no code (-) for DUs which are unchanged.
- the revision date: this column indicates the DU date code.

3 CLASSIFICATION OF INFORMATION

The DUs in sections 04-10-00 and 04-20-00 are classed in increasing order according to ATA 100 numbering.

End of the Document Unit

UPDATE OF THE ALS

1 REVISION OF THE ALS

The ALS is revised approximately every 6 months.

The ALS cover page bears the issue date code (first issue), the current revision number taken in chronological order followed by the revision date code.

Example:

- issue date : 2012.01.30,
- revision 002 : 2012.03.28.

The last applicable revision is revision 002 with a date code of 2012.03.28.

The date code marked on the new or revised DUs corresponds to the revision date code marked on the cover page.

2 UPDATE OF THE ALS

2.1 Update procedure

The user is responsible for updating his documentation each time he receives a revision or a new issue from the manufacturer. The ALS is updated using the new LOEDU which specifies which DU must be inserted (new DU), replaced (revised DU) or removed (deleted DU). A deleted DU is issued with the wording "DU TO BE REMOVED FROM THE MANUAL" in the revision concerned.

Once the ALS has been updated, the date code of each DU indicated at the bottom left hand side of each page must correspond with the LOEDU.

The beginning of the ALS includes a "HIGHLIGHTS" page which explicitly summarizes the modifications made to each revision or new issue.

NOTE

After updating, the "RECORD OF REVISIONS" page at the beginning of the ALS must be completed (date, name and signature).

2.2 Identification of the modifications

Modifications are identified by a revision mark in the form of a vertical line in the right hand margin of the page.

Revision marks are not used in the following DUs:

- LOEDU,
- Table of Contents,
- Highlights.

3 EFFECTIVITY

The operator must check the effectivity of the ALS used (correctly updated) for the helicopter concerned.

The effectivity (helicopter type and version) is indicated on the cover page and is repeated at the bottom of each page comprising the ALS DUs.

End of the Document Unit

BREAKDOWN OF THE ALS

1 PURPOSE OF THE ALS

The Airworthiness Limitations Section (ALS) drafted by the manufacturer, draws up a list of all the components subject to one or more airworthiness limitations and defines these limits.

It specifies the mandatory operations and limits to ensure the continued airworthiness of the helicopter (*), in all its varied operating missions.

For the airworthiness limitations and the life limits assigned to the engine components, refer to the engine manufacturer's documentation.

(*) helicopter (type/version) defined by Airbus.

2 CHANGES TO THE ALS

The ALS section is modified by the manufacturer according to changes to the helicopter definition.

Each modification to the ALS is subject to approval by the European Aviation Safety Agency (EASA).

3 AIRWORTHINESS LIMITATIONS SPECIFICATION

The Airworthiness Limitations can be expressed in the form of a life limit, an airworthiness inspection interval or various specific information.

3.1 Components subject to a Service Life Limit (SLL)

Life-limited components are components that are exposed to fatigue deterioration due to in-service stress, and the failure could cause catastrophic effect to the helicopter.

The loads applied to these components cause the latter to deteriorate according to one or more types of fatigue:

- vibration fatigue, which is related to the number of flight hours logged (very large number of low-amplitude load variations),
- low-cycle fatigue, which is related to a limited number of large load variations (example: centrifugal load variations, related to the number of rotor starts, or large torque variations related to takeoff, etc.),
- fatigue related to the number of times certain components are used.

These components must be removed from service when the specified limit is reached.

NOTE

The service life limits assigned to components guarantee flight safety with regard to the flight load fatigue alteration of components with no external deterioration. These service life limits are not a commercial guarantee because a component may be removed due to wear, fretting, corrosion or scoring, etc., before expiry of the service life.

The only warranty applicable to the helicopter and any component is the warranty included in the Sales Contract for the helicopter and the components.

3.2 Components subject to an airworthiness inspection

A component subject to an airworthiness inspection must be inspected periodically to confirm:

- that there is no damage,
- or that the damage found is within the removal or maintenance criteria specified in the documentation.

In both cases, the component is kept in service until the next inspection. The component must be removed from service when it reaches the criterion for removal, or made serviceable again as per the instructions given in the documentation.

The visual inspection is aimed at the overall external appearance of the component (no distortion, failure, cracks, scratches, signs of heating or wear, etc.), which would modify its original condition.

STANDING INSTRUCTIONS RELATIVE TO CONDITION: Surface damage and damage to protection and paint on all helicopter components must be examined and treated without delay in accordance with the applicable instructions (criteria, then reworking).

WHEN AN AIRWORTHINESS INSPECTION REQUIRES THE MEASUREMENT OF A PHYSICAL VALUE (CLEARANCE, VIBRATION LEVEL, TIGHTENING TORQUE, BONDING SEPARATION SURFACE AREA, ETC.), THE VALUE MEASURED AT EACH INSPECTION MUST BE RECORDED ON AN APPROPRIATE MEDIUM (E.G.: FOLLOW UP SHEET).

4 EFFECTIVITY - RESPONSIBILITY

The limits assigned to the component part numbers listed in this section are applicable to material:

- acquired directly from Airbus or through the Airbus subsidiaries or distribution network,
- purchased from the equipment vendors listed in our spare part catalogs, either directly or through their own distribution network.

In all cases, the manufacturing source is specified in the "airworthiness document", depending on the regulations (JAA Form One or equivalent).

Airbus will not carry out repairs on service life limited parts which were procured other than through the Airbus distribution network.

IT IS PROHIBITED TO REUSE PARTS, EQUIPMENT OR ASSEMBLIES ORIGINATING FROM A HELICOPTER WHICH HAS BEEN INVOLVED IN AN ACCIDENT, WITHOUT FORMAL TECHNICAL APPROVAL FROM THE AIRBUS CUSTOMER TECHNICAL SUPPORT DEPARTMENT.

REMINDER: THE AIR ACCIDENT INVESTIGATION BOARD OF THE COUNTRY CONCERNED IS RESPONSIBLE FOR CLASSIFYING AN EVENT AS AN ACCIDENT.

WARNING

AN AIRCRAFT CONSIDERED DESTROYED CANNOT BE REPAIRED OR REBUILT, WITHOUT FORMAL TECHNICAL APPROVAL FROM THE AIRBUS CUSTOMER TECHNICAL SUPPORT DEPARTMENT.

AN AIRCRAFT IS CONSIDERED TO BE DESTROYED WHEN THE OEM-DEFINED MAIN STRUCTURE IS DAMAGED (DEFORMED, CRUSHED. CORRODED, CONSUMED BY FIRE. ETC.) TO SUCH AN EXTENT THAT ITS RETURN TO AN AIRWORTHY CONDITION CAN ONLY BE ACCOMPLISHED BY UTILIZING ANOTHER MAIN STRUCTURE OR BY REPAIRING THE DAMAGED MAIN STRUCTURE UTILIZING AN ORIGINAL EQUIPMENT MANUFACTURER APPROVED RIG OF REFERENCE.

End of the Document Unit

USE OF THE ALS

1 COMPLIANCE WITH THE ALS

The Airworthiness Limitations Section (ALS) specifies all the components which are subject to airworthiness limitations, which must be complied with in addition to the information given in the maintenance program (MSM).

WARNING

THE LIMITS MUST BE COMPLIED WITH TO ENSURE THE AIRWORTHINESS OF THE HELICOPTER. THE TIME LIMITS MUST NOT BE EXCEEDED AND ANY CHANGE MUST BE SUBJECT TO PRIOR APPROVAL BY THE EASA AND AIRBUS.

2 OCCURRENCE REPORTING

Reporting of any malfunction, failure, fault or other occurrence which had or may have an adverse effect on the airworthiness of the helicopter is a key factor to keeping the helicopter airworthy and for the continuous improvement of flight safety.

2.1 Reporting to the airworthiness authorities

The operator shall inform his relevant Airworthiness Authorities of any occurrence, in accordance with the local regulations he is subject to.

2.2 Reporting to the helicopter manufacturer

As part of the Continued Airworthiness Program, any technical incident, discovered in operation or during maintenance work, must be reported by the operator to the helicopter manufacturer.

A technical incident is a technical event which is unexpected.

A few examples of technical incidents are given below. This list is not exhaustive. It is given as an illustration, intended to help the operator to understand the definition of a technical incident which is to be reported to the helicopter manufacturer.

In case of doubt on the need to report an event, the event shall be reported.

Examples of incidents to be reported (non-exhaustive list):

- technical difficulty in accomplishing the mission as planned, including but not limited to, difficulty resulting in an aborted mission,
- technical event increasing the crew workload,
- technical event that may jeopardize the airworthiness of the helicopter for the continuation of the mission or the safety of persons onboard or on the ground,
- in-flight loss or rupture of any part,
- abnormal deterioration (wear, scratch, crack, corrosion, delamination, bonding failure, indentation, etc.) to any structural or mechanical part, especially when exceeding the Maintenance Manual (AMM) removal/repair criteria or in case no maintenance criteria exist,
- system malfunction or loss (including partial or temporary incorrect operation) and interference within or between systems,
- incorrect operation of a system intended to minimize the effects of an accident, including when being used for maintenance or test purposes,
- incorrect operation or malfunction of a system, or deterioration of a part found when complying with an Airworthiness Directive,

- non-conformity of a product, part or equipment with respect to its approved design, or a product of unknown or doubtful origin,
- missing, misleading, incorrect or unclear procedure that could lead to a flight crew or maintenance error,
- abnormal vibration,
- abnormal heating, smoke, fire,
- repetitive instances of any technical event which in isolation would not be considered "reportable" but which due to the frequency at which they arise, form a potential hazard,
- any other occurrence which has resulted in or may result in an unsafe condition.

In case of doubt concerning the need to report an event, the event shall be reported.

3 IN-SERVICE MONITORING OF COMPONENTS

All the components listed in section 04-10-00 of the ALS are to be monitored in service, unless "not to be monitored in service" is specifically mentioned in the task.

The in-service monitoring of components consists in recording the number of flight hours and/or the calendar time and/or the cycles logged on the equipment log card (**FM**).

If the component does not have a log card (component monitored via the log card of its higher assembly, etc.), before installation, the operator must organize his documentation in order to comply with the limits relating to this component (refer to Standard Practices Manual Work Card 20.08.05.101).

These recordings must be done in the various allocations of the component: case of transfer from one helicopter to another (refer to § 6).

All the documentation must be organized to enable recording of all the parameters used to comply with the limits.

4 COMPONENT PART NUMBERS

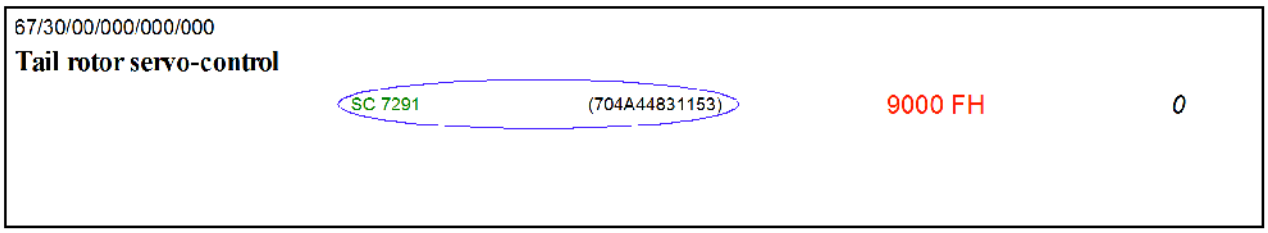
4.1 Definition

- **Manufacturer Part Number (MP/N)**: This is the identification number of the component marked on the identification plate or directly on the component.
- **Airbus Part Number (PN)**: This is the Airbus part number which is associated with the manufacturer's part number (MP/N) for components which are not manufactured by Airbus.

The part number of the components which is given in the ALS is the manufacturer's part number (MP/N).

In the case of components which are not manufactured by Airbus, the MP/N is followed by the associated PN (in brackets).

Example :



In some cases the PN is not specified, in this case it is replaced by a hyphen.

4.2 Part number followed by a suffix

In some cases, the MP/N can be assigned a suffix (example: MP/N 365A33-6005-08**M**), used for spares to indicate the installation of an optional component or a specific configuration.

These MP/Ns with a suffix generally do not appear in the ALS.

The limit applicable to an MP/N with a suffix is identical to that of the same MP/N without a suffix.

Example: the limit given for assembly MP/N 365A33-6005-08 is valid for assembly MP/N 365A33-6005-08 and assembly MP/N 365A33-6005-08M.

4.3 Applicability of the limits

The limit(s) specified for the first MP/N are valid for all the other MP/Ns listed in the same task.

Example :

63/20/00/000/000/135

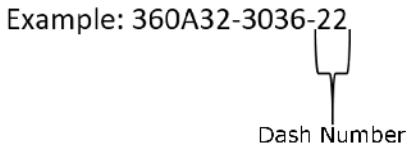
Main bevel gear wheel

360A32-3256-25 360A32-3256-26	(-) (-)	20000 FH	0

4.4 Changes to part numbers

Unless otherwise specified, the service life and airworthiness inspections specified in the ALS apply to the components whose part numbers are specified in this section as well as to their successive dash numbers if these are not yet specified in this section.

In this case, check the installation effectivity of the dash number with the Airbus technical support department.



5 EXPRESSION OF THE LIMITS

A component or an airworthiness inspection can be subject to one or more airworthiness limitations, expressed in:

- flight hours,
- calendar time,
- cycles with large centrifugal load or torque variations,
- number of operations (for certain components).

5.1 Flight Hours

FH (Flight Hours), are counted from takeoff to landing.

5.2 Calendar Time

Calendar time can be expressed:

- in years (Y: Years),
- in months (M: Months),
- in days (D: Days).

5.2.1 Initial aging date for a component

The initial aging date for a component to be taken into account is generally the date of first installation or first removal from storage, to be recorded on the equipment log card.

The storage time before installation is therefore not to be taken into account on the condition that storage was in accordance with the applicable documentation, except for the following specific cases.

For some components, the aging date to be taken into account is the date of manufacture marked on the Log Card (FM). In this case, TSM (Time Since Manufacture) is specified in each task concerned in the ALS.

5.3 Cycles

5.3.1 Cycles with large centrifugal load variation

To limit the number of parameters to be monitored during aircraft operation, the service life limit of the components likely to be damaged by large centrifugal load variations has been calculated based on 5 rotor starts/stops per flight hour.

If aircraft operation involves on average more than 5 rotor starts/stops per flight hour over a period of more than 600 FH, the actual service life of some components could be less than that specified in this document.

In this case, you must inform Airbus.

5.3.2 Cycles with large torque variation

The number of torque variation cycles varies significantly with the missions that can be performed by a helicopter (example: external load carrying operations, such as logging, can lead to a large number of torque cycles), up to 60 per hour, whereas passenger transport flights of one hour on average lead only to one torque cycle per hour.

Therefore, no fixed number of cycles has been specified in order to avoid premature removal of components that are subject to few torque variations. The life limit of the components concerned is indicated in torque cycles (TC).

Consequently, the number of torque variation cycles logged must be monitored carefully and counted as follows:

- **1 landing with or without stopping the rotor = 1 cycle (1 TC)**
- **1 external load carrying operation = 1 cycle (1 TC)**

The cycles associated with external load carrying operations must be added to the landing cycles.

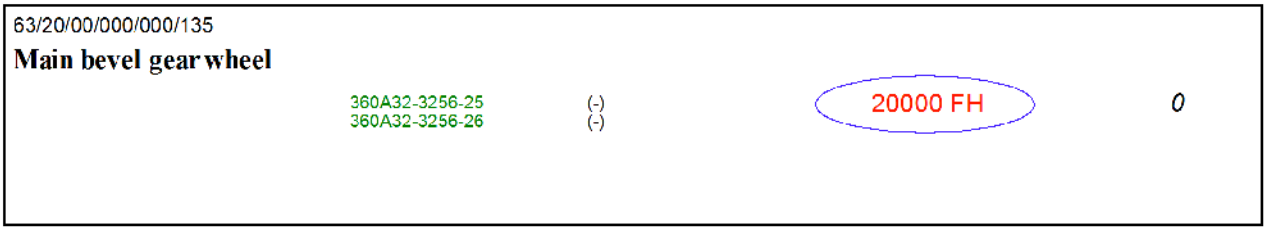
5.4 Number of operations for certain components

Certain components may be subject to fatigue limits related to the number of operations. In this case, the type of use to be counted is defined directly in the tasks concerned with the permissible number.

5.5 Airworthiness limitation units - Description and counting

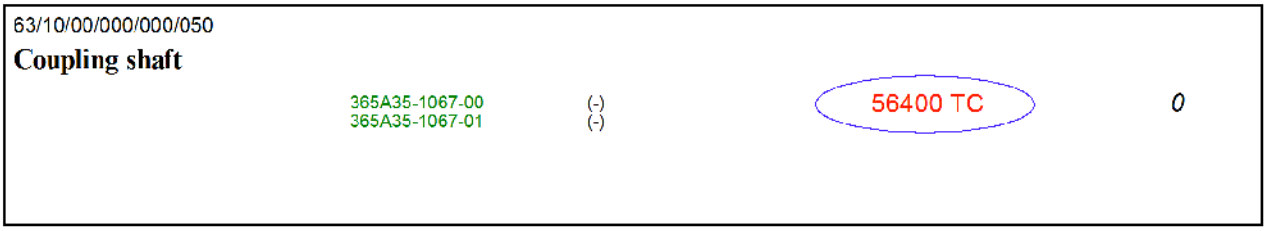
5.5.1 Airworthiness limitation expressed in Flight Hours

Tasks for which the Airworthiness Limitation is expressed in flight hours: count the number of flight hours.



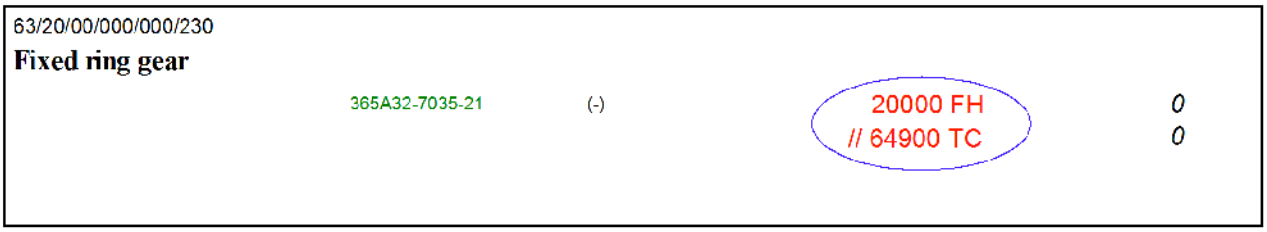
5.5.2 Airworthiness limitation expressed in Torque Cycles

Tasks for which the Airworthiness Limitation is expressed in torque cycles: count the torque cycles as per § 5.3.2 or 5.4.



5.5.3 Airworthiness limitations expressed in Flight Hours and Torque Cycles

Tasks for which the Airworthiness Limitation is expressed in flight hours and torque cycles: count the flight hours and the torque cycles logged.



5.5.4 Specific cases

5.5.4.1 Use of the helicopter in cold and very cold weather

Operation of helicopters when the outside air temperature is between - 15°C and - 40°C gives rise to higher fatigue loads on some components. As these loads are different according to the components, a corrective multiplication factor is to be applied to the flying hours (FH) and/or Torque Cycles (TC) after each flight carried out in cold and very cold weather. The components concerned are identified by the special annotation: “Operation in cold and very cold weather, corrective multiplication factor: X”.

Example for application of the corrective multiplication factor on Flight Hours (FH): For the parts followed in flight hours (FH), a penalty factor must be mandatorily applied whenever flight in cold weather conditions, so in a temperature domain below - 15°C , are performed. Flight hours (FH) spent in cold weather conditions must be multiplied by the specified factors. Example: for a component for which penalty factor is equal to 2.0, if the flight time is spent 2 hours in a temperature domain above - 15°C and 1 hour in a temperature domain below - 15°C , the total accumulated hours is 2H+1H x 2,0 = 4H. Example for application of the corrective multiplication factor on Torque Cycles (TC): For the parts, followed in torque cycles (TC), if the take off and/or landing is performed in cold weather conditions, whatever the temperature flight conditions, the torque cycles (without link with flight hours performed) must be multiplied by the specified factors. Example: For a component for which penalty factor is equal to 14, if the number of torque cycles performed in a temperature domain above - 15°C is equal to 200 TC and if 25 takes off and/or landings are performed in a temperature domain below - 15°C, the total accumulated TC is 200 TC+25 TC x 14 = 550 TC.

5.6 Removal of life-limited components

When a limit is reached, the component must be rejected and finally withdrawn from service. When a component is assigned several limits, it must be rejected when the first of the limits is reached (refer to § 5.10).

5.7 Provisional limitations

The limits for which additional fatigue tests indicate the possibility of an extension are identified by code "P" (Provisional). This code (P) follows the value of the limit in the tasks concerned.

The operator should therefore store the component in question after removal, pending a service life extension.

5.8 Version identification of components

Generally, new components can be installed on several Dauphin family helicopter types or versions. As the operating conditions are sometimes different from one version to another, components must be identified according to the version to which they have been assigned in order to know the versions on which the components have been used.

All the components listed in section 04-10-00 are subject to version identification.

5.8.1 Principle

The version is identified by marking one or more letters or figures in a box (corresponding to the version to which the component is assigned), in an indelible manner, near to the part number, in accordance with the instructions of MTC Work Card 20.08.05.103.

Example:

[Y] - 365A35-1067-00, component installed on helicopter version X (original assignment).

[Y][X] - 365A35-1067-00, component transferred from helicopter version X to version Y; version Y must be struck out and followed by the new assigned version X.

5.8.2 Marking the version

The version is marked on the components at the latest when they are removed for the first time.

The equipment log card of the component concerned or of its next higher assembly must be filled in as soon as the component is installed on a helicopter.

5.9 Periodic and perform once airworthiness inspections

The maintenance operations associated with the airworthiness inspections indicated in section 04-20-00 of the ALS are to be performed:

- **Periodically:** the maintenance operation must be performed at the latest when the indicated limit is reached.

NOTE

For operational reasons, the maintenance operation can be performed before the limit is reached. However, the maximum interval between two maintenance operations must be complied with.

or

- **Once only:** the maintenance operation must be performed once only, when the indicated limit is reached. It must not be performed before the limit is reached.

These maintenance operations are identified by a dotted line around the task and by code PO (Perform Once) before the limit value.

These operations can be performed either:

- After installation of a component removed from the same helicopter,

- After the introduction to service of a new, overhauled or repaired component,

- After installation of a component originating from another helicopter,

- Each time after the component is installed.

The start of the maintenance operation is specified in each task concerned.

Example :

62/20/00/605/000/000

Mast and/or hub

AMM	*62.20.00.22*	PO	3 FH	7 FH

Check tightening torque loads.
Each time after the component is installed.

5.10 Performance interval

Some components with service life limit (SLL) and airworthiness inspections (CHK) are assigned several performance limits separated by the "**//**" or "**&**" sign.

In the case of limits separated by the "//" sign, at the **first limit reached, the component must be removed and rejected or the airworthiness inspection must be performed.**

The next limit at which the component is to be rejected or the maintenance operation is to be performed will once again be the first limit reached.

In the case of airworthiness inspections separated by the "&" sign, the maintenance operation must be performed at each limit indicated.

Example 1:

63/20/00/000/000/230

Fixed ring gear

365A32-7035-21	(-)	20000 FH		*0*
		// 64900 TC		*0*

A fixed ring gear has logged only 50000 **TC** and 20000 **FH**. The fixed ring gear has reached the FH limit; it must therefore be removed and rejected.

Example 2:

25/62/00/000/000/090

Float

CMM 25.69.35 PO 6 Y 180 D
& 9 Y 180 D
& 12 Y 180 D

216064-0 (-)
216166-0 (-)
216167-0 (-)
216080-0 (-)

Proof pressure test and detailed check.

The maintenance operation must be carried out once at 6 Y, once at 9 Y and once at 12 Y. This maintenance operation shall be carried out again only if the component is removed then reinstalled or replaced.

5.11 Specific and severe atmospheric operating conditions

Certain specific and severe climatic conditions are considered as specific operating conditions. This concerns the use of helicopters in the following conditions:

- **Tropical and damp atmosphere**

  Combination of high ambient temperatures and high humidity level (from +28 °C (+82.4 °F) and 75% relative humidity).

- **Salt-laden atmosphere**

  Aircraft based on a ship or based more than 50% of the time inland at less than 1 km (.54NM) from the coast or,

  flying more than 50% of the time over the sea at low altitude (less than 1000 feet).

- **Sand and/or dust-laden atmosphere**

  Sand-laden wind, landing on sandy ground.

- **Cold weather**

  encountered or scheduled temperatures from -15 °C (5 °F) to -25 °C (-13°F).

- **Very cold weather**

  encountered or scheduled temperatures from -25 °C (-13°F) to -40 °C (-40°F).

The limitation relating to these operating conditions is covered with all the ALS tasks.

5.12 Airworthiness limitations value

The airworthiness limitation value represents the limitation not to be exceeded. No tolerance is permitted on the airworthiness limitation value.

The airworthiness limitation value equals to a limit value + margin value.

5.12.1 Definition of the margin

To introduce “flexibility” into maintenance planning in order to compensate for unpredictable situations (e.g.: unforeseeable increase in the helicopter utilization rate), Airbus defines a value, called "margin" which can be added to the limit value.

NOTE

The margin concept only concerns airworthiness checks (CHK).

No margin is allowed for SLL type limitations, therefore the margin value displayed is 0.

This margin, added to the limit value displayed in the column "CHK" to which it is applicable, results in the airworthiness limitation value (limit value + margin value) to be taken into account for two consecutive inspections.

5.12.2 Application of the margin

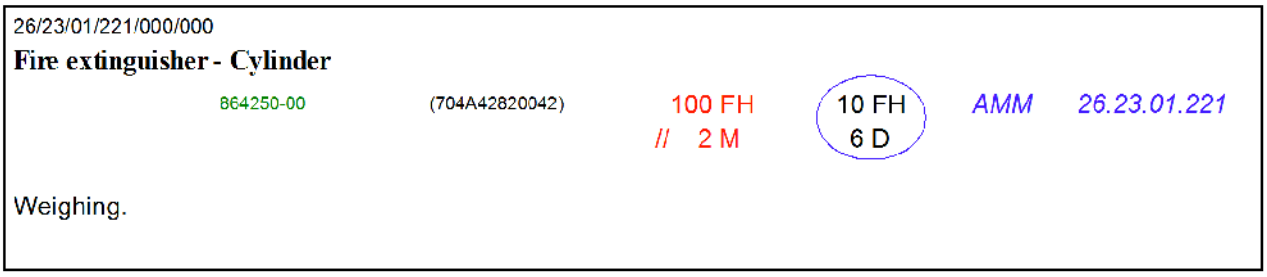
The margin can be used repetitively for each limit value interval (see “Example of use of the margin” to § 5.12.4).

However, to maintain certain "flexibility" in maintenance planning, Airbus recommends to plan airworthiness checks using the limit value without taking the margin into account.

5.12.3 Specification of the margin

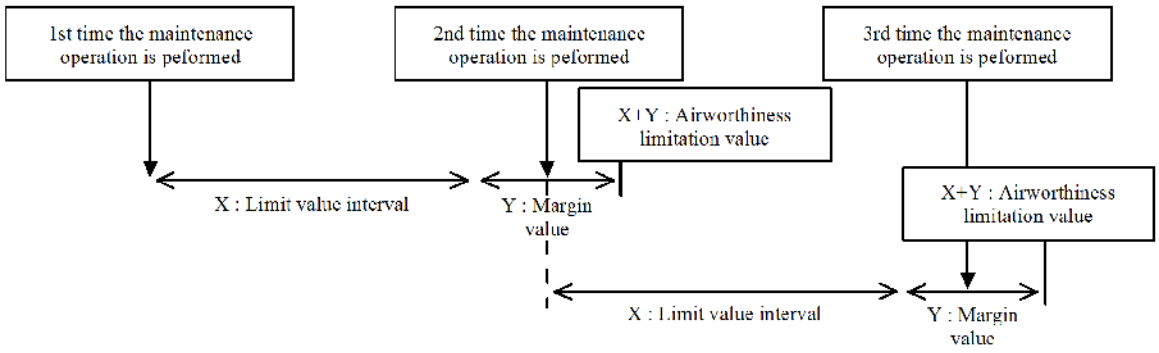
The margin value and its units are specified as shown in the following example.

Example:



No unit specified when the margin equal zero.

5.12.4 Example of use of the margin



X is the limit value for performing the maintenance operation and Y is the margin value added to the limit value.

After using the first margin value (partially or completely), the next maintenance operation is to be performed again within the X+Y interval, that is to say before the next airworthiness limitation value.

6 TRANSFER OF A LIFE-LIMITED COMPONENT INCLUDED IN ALS SECTION 04-10-00 FROM ONE HELICOPTER VERSION TO ANOTHER HELICOPTER VERSION

6.1 Dauphin family helicopter versions concerned by the transfer

The versions concerned by the transfer of a component are identified in the following table:

Dauphin Family	Civil versions
SA 365	C, C1, C2, C3
AS 365	N, N1, N2, N3
EC 155	B, B1

CAUTION

IT IS FORBIDDEN TO TRANSFER COMPONENTS BETWEEN CIVIL AND MILITARY VERSION HELICOPTERS.

6.2 Authorization to transfer

The transfer of a component (part or assembly) listed in section 04-10-00 from helicopter version "x" to helicopter version "y", is authorized only if the following conditions are met simultaneously:

- both helicopter versions "x" and "y" must be listed in the table above. The transfer of components originating from any other version or any other type of helicopter is forbidden,
- the part number of the component to be transferred is listed both in section 04-10-00 of version "x" and in section 04-10-00 of version "y",
- the component to be transferred does not have a provisional limitation (neither in section 04-10-00 of version "x" nor in section 04-10-00 of version "y").

6.3 Application rules

If the transfer is authorized, comply with the following application rules:

6.3.1 If the component Service Life Limits are exactly the same on the aircraft version "x" and "y", you must continue to count the flight hours and/or torque cycles on the aircraft version "y".

6.3.2 If the component Service Life Limits on the aircraft versions "x" and "y" are different, convert the flight hours and/or torque cycles already logged by applying the calculation rules given in paragraph 6.4.

6.4 Calculation rules

6.4.1 Information required for the calculations

Calculating the equivalent flight time on version "y" when the component has already flown on version "x":

- flight time logged on version "x" = tx
- component service life limit on version "x" = SLLx
- component service life limit on version "y" = SLLy
- equivalent flight time on version "y" = ty
- remaining flight time available = T

6.4.2 Calculating ty

ty = tx X (SLLy : SLLx)

6.4.3 Calculating T

Remaining flight time available: T = SLLy – ty

NOTE

This type of calculation also applies for components with a service life limit expressed in torque cycles.

Examples:

Component ’A’ that has logged 2000 FH on a version B1 helicopter and is transferred to a version N2 helicopter:

- service life limit of component 'A' on version B1: 6000 FH.
- service life limit of component 'A' on version N2: 3000 FH.
- equivalent flight time on version N2: 2000 x (3000 : 6000) = 1000 FH
- remaining flight time on version N2: 3000 - 1000 = 2000 FH.

Component 'B' that has logged 4000 FH on a version N1 helicopter and is transferred to a version N2 helicopter:

- service life limit of component 'B' on version N1: 8000 FH.
- service life limit of component 'B' on version N2: 10000 FH.
- equivalent flight time on version N2: 4000 x (10000 : 8000) = 5000 FH
- remaining flight time on version N2: 10000 - 5000 = 5000 FH.

Component 'C' that has logged 1500 FH on a version B1 helicopter and is transferred to a version N2 helicopter:

- service life limit of component 'C' on version B1: 4500 FH.
- service life limit of component 'C' on version N2: 3000 FH. **provisional limitation** (no indication authorizing transfer)
- In this case the transfer is not authorized.

NOTE

The examples above can be adapted directly to service life limits expressed in torque cycles by replacing flight hours (FH) with torque cycles (TC).

6.5 Counting

Then count the flight hours and/or torque cycles and add them to the equivalent flight time "ty".

6.6 Recording

In all cases record the transfer information and calculation on the equipment log card of the component concerned (refer to § 5.8).

An individual log card must be drawn up if the component has been monitored via the log card of its higher assembly.

7 TRANSFER OF A COMPONENT SUBJECT TO AN AIRWORTHINESS INSPECTION INCLUDED IN SECTION 04-20-00 FROM ONE HELICOPTER VERSION TO ANOTHER HELICOPTER VERSION

7.1 Dauphin family helicopter versions concerned by the transfer

The versions concerned by the transfer of a component are identified in the following table:

Dauphin Family	Civil versions
SA 365	C, C1, C2, C3
AS 365	N, N1, N2, N3
EC 155	B, B1

7.2 Authorization to transfer

The transfer of a component included in section 04-20-00 of helicopter version "x" to helicopter version "y", is authorized only if the following conditions are met simultaneously:

- both helicopter versions "x" and "y" must be listed in the table above. The transfer of components originating from any other version or any other family of helicopter is forbidden,
- the part number of the component to be transferred is listed both in section 04-20-00 of version "x" and in section 04-20-00 of version "y".

7.3 Transfer rules for components subject to an airworthiness inspection

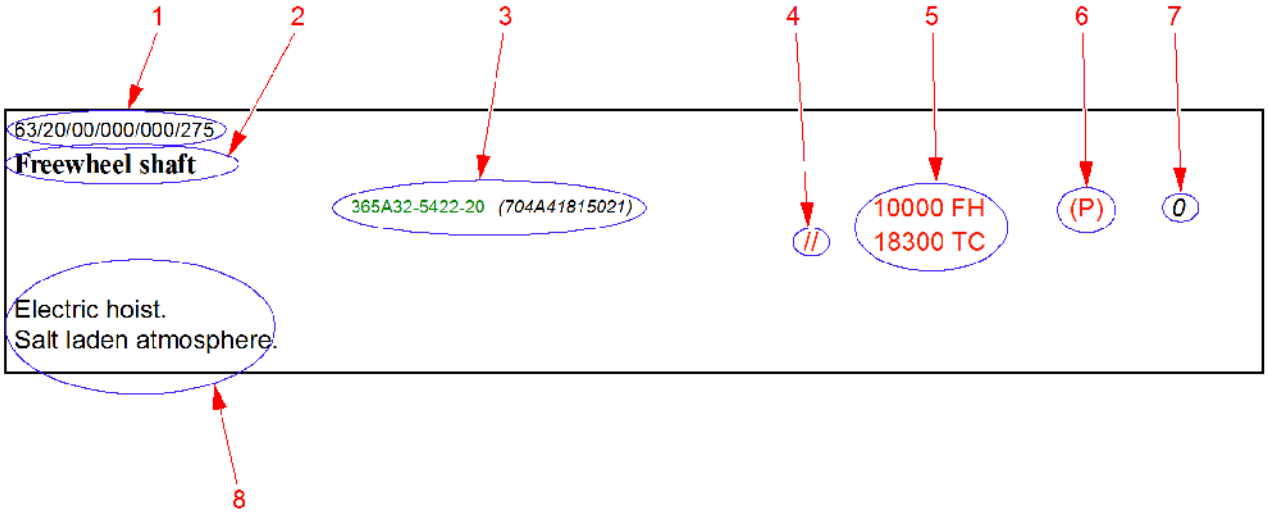
If the inspection intervals (flight hours and/or torque cycles) for the component are identical on version "x" and version "y", continue counting the flight hours and/or torque cycles on version "y".

If the inspection intervals are different on version "x" and version "y", perform the airworthiness inspection when the component is transferred, then comply with the inspection interval specific to version "y".

8 CONTENT OF A TASK

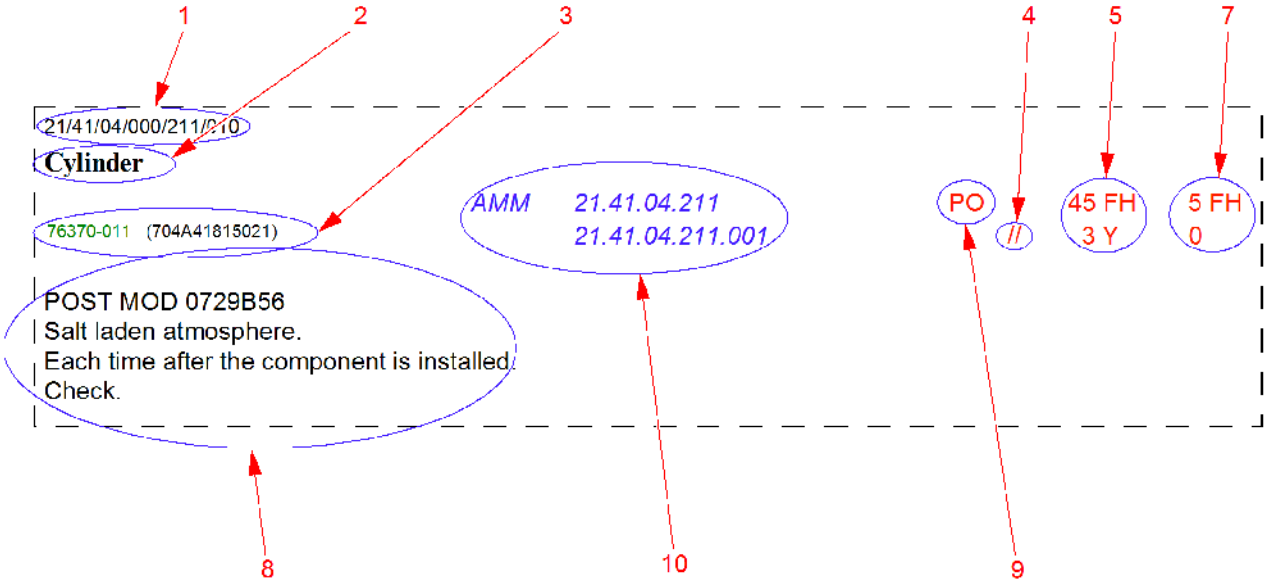
Section 04-10-00

Fictitious example.



Section 04-20-00

Fictitious example.



- 1 - Task number (refer to § 2.2 of the DU “Structure of the ALS”)
- 2 - designation of the component
- 3 - equipment manufacturer's part number and the Airbus management number (refer to § 4)
- 4 - // (or), & (and). (refer to § 5.10)
- 5 - limit value(s) and unit(s) (refer to § 5.1, 5.2, 5.3)
- 6 - P: provisional limitation for the service life limit in flight hours (refer to § 5.7)
- 7 - margin value(s) and unit(s) (refer to § 5.12)
- 8 - comments if necessary
- 9 - PO (perform once): one-off operation (refer to § 5.9)
- 10 - Reference of the documentation: AMM, CMM, etc.

End of the Document Unit

SB INCORPORATED

Airbus is required to issue Service Bulletins (ASB/SB) subsequent to new events (incidents, etc).

The information contained in the SBs is incorporated in the ALS if, and only if:

The SB contains new periodic checks, or modifies the existing ALS limitations and its incorporation is approved by EASA.

The table below specifies for each SB incorporated:

- the number of the SB and its revision number,
- the subject of the SB,
- the ATA chapter/section concerned,
- the number of the associated EASA Airworthiness Directive (AD),
- the date on which the SB is incorporated in the ALS.

SB N°	Subject	ATA	AD number	Incorporation date
04A003 R0	Reduced service life for the tail rotor pitch change control rod.	65–20	2001-444-003(A)	2001.11.30
04A005 R0	Tail Rotor Pitch Change Control Rod.	65–20	2001-473-006(A)	2003.07.03
05A006 R1	Check and Replacement of Baggage Compartment Fire Extinguisher and Engine Fire Extinguishers.	26–21	F-2004-180	2005.01.20
05A007 R0	Check of the MGB Planet Gear Carrier.	63–20	F-2005-181	2005.01.20
62A006 R0	Check for Correct Positioning of the Filler Wedge of the Tenon at the Main Rotor Blade Tip.	62-10	F-2004-106	2005.01.20
64-003 R0	Tail Rotor Blade Protection Against Sand Erosion.	64–10	Not Applicable	2005.01.20
05A005 R1	Check of the Main Gearbox (MGB) Base Plate.	63–20	F-2004-023	2005.06.09
05A011 R0	Tail Rotor Blade Check.	64–10	AD-2006-0099	2006.11.14
62A014 R0	MAIN ROTOR - Check of the Rotor Mast / Hub Nut.	62–20	EAD-2006-0368-E	2008.06.27
05A007 R2	Check of the MGB Planet Gear Carrier. - This SB revision leads to no change the ALS.	63–20	2007-0288-E	2010.04.08
05A007 R3	Check of the MGB Planet Gear Carrier. - This SB revision leads to no change the ALS.	63–20	2007-0288-R1	2013.09.12
05A027 R1	Check of the deployment system of the external life rafts located in the footsteps.	25–69	2014-0214	2017.09.11
05A021 R1	Visual check of the 10-blade TRH hub.	64–21	2012-0227-1	2017.09.11

End of the Document Unit

Task Number Description/Remarks	MP/N (PN)	Type of limit: SLL	*Margin*

ATA 53-FUSELAGE

53-10 PRIMARY STRUCTURE

53/10/00/000/000/005

Transmission deck - MGB forward fitting

365A21-8117-00	(-)	16100 FH	*0*
365A21-8117-01	(-)		

53/10/00/000/000/010

Transmission deck - MGB fitting assy

365A21-8201-00	(-)	20000 FH	*0*

53/10/00/000/000/020

Transmission deck - MGB aft fitting

365A21-3355-04	(-)	3200 FH	*0*
365A21-3355-05	(-)		

53/10/00/000/000/030

Carbon tube - TGB mount

365A23-4211-00	(-)	30000 FH	*0*
365A23-4211-02	(-)		

53/10/00/000/000/040

Carbon tube - TGB mount

365A23-4210-02	(-)	20000 FH	*0*
365A23-4213-00	(-)		

End of the Document Unit

Task Number Description/Remarks	MP/N (PN)		Type of limit: SLL	*Margin*

ATA 62-ROTOR (S)

62-10 MAIN ROTOR BLADES

62/10/00/000/000/005 **Main rotor blade**	365A11-0080-00	(-)	10000 FH	*0*
62/10/00/000/000/007 **Main rotor blade**	365A11-0080-01	(-)	20000 FH	*0*

62-20 MAIN ROTOR HEAD

62/20/00/000/000/075 **Lower attach beam**	365A31-4596-00 365A31-4596-01	(-) (-)	5350 FH	*0*
62/20/00/000/000/097 **Upper attach beam**	365A31-4597-01 365A31-4706-00 365A31-4706-01	(-) (-) (-)	5350 FH	*0*
62/20/00/000/000/120 **Spherical thrust bearing**	57910600	(704A33633213)	4300 FH	*0*
Service life limit for the metal section only.				
62/20/00/000/000/143 **Frequency adapter end-fitting**	12-11505P 12-11506P	(704A33633218) (704A33633217)	20000 FH	*0*
62/20/00/000/000/150 **Frequency adapter**	F4T410001	(704A33640090)	20000 FH	*0*
Service life limit for the metal section only.				
62/20/00/000/000/170 **Unequipped rotor blade pin**	365A31-2031-20	(-)	2800 FH	*0*

Task Number Description/Remarks	MP/N (PN)		Type of limit: SLL	*Margin*
62/20/00/000/000/190				
Blade horn				
	365A31-4561-00 365A31-4561-01	(-) (-)	20000 FH	*0*
62/20/00/000/000/200				
Spherical thrust bearing - Attach beam bolt				
	704A33-633-224	(-)	8000 FH	*0*
62/20/00/000/000/220				
Frequency adapter - Attach beam bolt				
	704A33-633-242	(-)	8050 FH	*0*
62/20/00/000/000/230				
Scissors drive adapter				
	365A31-1171-00	(-)	5000 FH	*0*
62/20/00/000/000/300				
Non-rotating swashplate				
	365A31-4531-01 365A31-4531-02	(-) (-)	3800 FH	*0*
62/20/00/000/000/310				
Rotating swashplate				
	365A31-4541-01 365A31-4541-02	(-) (-)	20000 FH	*0*
62/20/00/000/000/330				
Pitch - Change rod body				
	T24497	(704A33633216)	1000 FH	*0*
62/20/00/000/000/337				
Pitch - Change rod body				
	T25274V TB212100-004	(704A33633259) (704A33633268)	20000 FH	*0*
62/20/00/000/000/340				
Pitch - Change rod end-fitting				
	1211507P 1211508P XMRE10RSF14 XMRE10RSF14G	(704A33633220) (704A33633221) (704A33633222) (704A33633219)	20000 FH	*0*
62/20/00/000/000/350				
Scissors lower link				
	365A31-1116-03 365A31-1116-04	(-) (-)	20000 FH	*0*
62/20/00/000/000/360				
Scissors upper link				
	365A31-1115-03 365A31-1115-04	(-) (-)	20000 FH	*0*

Task Number Description/Remarks	MP/N (PN)		Type of limit: SLL	Margin
62/20/00/000/000/370				
Flared housing				
	365A31-4591-00	(-)	6400 FH	0
	365A31-4591-01	(-)		
	365A31-4591-02	(-)		
	365A31-4591-03	(-)		
62/20/00/000/000/380				
Rotor mast bearing spacer				
	365A31-4527-00	(-)	1300 FH	0
62/20/00/000/000/382				
Rotor mast bearing spacer				
	365A31-4527-01	(-)	13700 FH	0
62/20/00/000/000/392				
Rotor shaft				
	365A31-4521-00	(-)	2470 FH	0
	365A31-4521-02	(-)	// 25795 TC	0
	365A31-4521-03	(-)		
	365A31-4521-04	(-)		
	365A31-4521-05	(-)		
62/20/00/000/000/395				
Rotor shaft				
	365A31-4521-06	(-)	7850 FH	0
	365A31-4521-07	(-)	// 30637 TC	0
62/20/00/410/000/100				
Scissors upper link				
	365A31-1115-05	(-)	10900 FH	0
62/20/00/605/000/030				
Mast and/or hub - Nut				
	365A31-2060-20	(-)	6200 FH	0
			// 31100 TC	0
62/20/00/605/000/040				
Mast and/or hub - Nut				
	365A31-2060-21	(-)	13800 FH	0
			// 69400 TC	0

End of the Document Unit

Task Number Description/Remarks	MP/N (PN)		Type of limit: SLL	*Margin*

ATA 63-ROTOR DRIVE (S)

63-10 ENGINE/MGB COUPLING

63/10/00/000/000/025

Coupling shaft forward flared housing

365A35-1010-24	(-)	6500 FH	*0*
365A35-1010-25	(-)		

63/10/00/000/000/035

Coupling shaft aft flared housing

365A35-1011-28	(-)	2900 FH	*0*

63/10/00/000/000/050

Coupling shaft

365A35-1067-00	(-)	56400 TC	*0*
365A35-1067-01	(-)		
365A35-1067-02	(-)		
365A35-1067-03	(-)		

63/10/00/000/000/070

Engine/MGB coupling shaft

365A35-1064-00	(-)	56400 TC	*0*
365A35-1064-01	(-)		
365A35-1064-02	(-)		
365A35-1064-03	(-)		
365A35-1073-00	(-)		
365A35-1073-01	(-)		
365A35-1073-02	(-)		
365A35-1073-03	(-)		

63/10/00/000/000/080

MGB input flange

365A32-5427-21	(-)	49400 TC	*0*
365A32-5427-22	(-)		

63/10/00/000/000/100

Front gimbal joint equipped

368A32-0624-00	(-)	3700 FH	*0*

POST MOD 0763C88

63/10/00/000/000/105

Rear gimbal joint equipped

368A32-0630-00	(-)	4300 FH	*0*

POST MOD 0763C88

63/10/00/000/000/110

Gimbal joint ring

368A32-0625-00	(-)	11000 FH	*0*

POST MOD 0763C88

63/10/00/000/000/115

Gimbal joint pin

368A32-0985-20	(-)	1100 FH	*0*

POST MOD 0763C88

Task Number Description/Remarks	MP/N (PN)		Type of limit: SLL	Margin
63/10/00/000/000/200 **Gimbal joint pin**				
	368A32-0985-21	(-)	40000 FH	0
POST MOD 0763D08				

63-20 MAIN GEARBOX

63/20/00/000/000/030 **Servo-control attachment fitting**				
	360A32-1163-03	(-)	20000 FH	0
	360A32-1163-04	(-)		
63/20/00/000/000/050 **Housing**				
	365A32-2046-15	(-)	10000 FH	0
			// 29300 TC	0
63/20/00/000/000/060 **Planet gear carrier**				
	360A32-3036-20	(-)	25900 TC	0
	360A32-3036-21	(-)		
	360A32-3036-22	(-)		
63/20/00/000/000/095 **Tail rotor drive shaft bevel gear wheel**				
	365A32-1058-22	(-)	12500 FH	0
63/20/00/000/000/135 **Main bevel gear wheel**				
	360A32-3256-25	(-)	20000 FH	0
	360A32-3256-26	(-)		
63/20/00/000/000/155 **Tail rotor drive shaft bevel pinion**				
	360A32-1029-26	(-)	2800 FH	0
	360A32-1029-29	(-)		
Part : assembly 360A32-1029-26 : 360A32-1218-06 360A32-1029-29 : 360A32-1218-09				
63/20/00/000/000/160 **First reduction pinion**				
	365A32-4129-03	(-)	20000 FH	0
63/20/00/000/000/180 **Double pinion**				
	365A32-7100-20	(-)	20000 FH	0
63/20/00/000/000/230 **Fixed ring gear**				
	365A32-7035-21	(-)	20000 FH	0
			// 64900 TC	0

Task Number Description/Remarks	MP/N (PN)		Type of limit: SLL	*Margin*
63/20/00/000/000/260 **Sun gear vertical shaft**	365A32-7021-23	(-)	20000 FH	*0*
63/20/00/000/000/261 **Sun gear vertical shaft**	365A32-7021-24	(-)	7200 FH	*0*
63/20/00/000/000/275 **Freewheel shaft**	365A32-5424-01	(-)	101600 TC	*0*
63/20/00/000/000/280 **Freewheel driven head shaft**	365A32-5422-20	(-)	18300 TC	*0*
63/20/00/000/000/300 **Front right housing equipped**	365A32-4340-02	(-)	40000 FH	*0*
POST MOD OP-63C95				
63/20/00/000/000/310 **Rear right housing equipped**	365A32-4342-03	(-)	40000 FH	*0*
POST MOD OP-63C95				
63/20/00/000/000/320 **Planet gear**	365A32-7023-21	(-)	5000 FH	*0*
Part: assembly 365A32-7023-21 : 365A32-9011-00, 365A32-9013-00.				
63/20/00/000/000/340 **First reduction gear housing**	365A32-7108-00	(-)	20000 FH	*0*
63/20/00/000/000/400 **Main bevel wheel**	368A32-0240-02 368A32-0240-03	(-) (-)	40000 FH	*0*
POST MOD 0763D09				
63/20/00/000/000/450 **Fixed rear output flange**	368A32-0405-21	(-)	9647 FH	*0*
POST MOD 0763D10				
63/20/00/000/000/525 **Half-gimbal joint**	365A32-7110-00 365A32-7112-00	(-) (-)	20000 FH	*0*

Task Number Description/Remarks	MP/N (PN)		Type of limit: SLL	Margin
63/20/00/000/000/530				
Upper cover				
	365A32-7102-00	(-)	20000 FH	0
	365A32-7824-00	(-)		
	365A32-7824-01	(-)		
63/20/00/000/000/545				
Gimbal ring				
	365A32-7113-20	(-)	20000 FH	0
	365A32-7113-21	(-)		
Part : assembly 365A32-7113-20 : 365A32-7114-00 365A32-7113-21 : 365A32-7114-01				
63/20/00/000/000/550				
MGB bottom				
	366A32-1060-26	(-)	52600 TC	0
Part : assembly 366A32-1060-26 : 366A32-1062-06				
63/20/00/000/000/555				
MGB bottom				
	366A32-1060-29	(-)	17000 TC	0
	366A32-1060-30	(-)		
Part : assembly 366A32-1060-29 : 366A32-1062-09 366A32-1060-30 : 366A32-1062-10				
63/20/00/000/000/560				
Fixed flange (tail rotor drive)				
	360A32-2034-20	(-)	2800 FH	0
	360A32-2034-21	(-)		
	360A32-2034-22	(-)		
	360A32-2034-23	(-)		
63/20/00/000/000/570				
Tail rotor drive shaft flange				
	360A32-1035-20	(-)	20000 FH	0
	360A32-1035-21	(-)		
63/20/00/000/000/575				
Accessory drive pinion				
	365A32-4059-20	(-)	10600 FH	0
Service life limit applicable only to aircraft equipped with OP63B98.				
63/20/00/000/000/600				
Main housing				
	368A32-0100-01	(-)	25000 FH	0
			// 125000 TC	0
POST MOD 0763C88 Operation in cold and very cold weather, corrective multiplication factor 1.65 for FH only.				

Task Number Description/Remarks	MP/N (PN)		Type of limit: SLL	Margin
63/20/00/000/000/605				
Right input housing				
	368A32-0121-00	(-)	40000 FH	0
			// 200000 TC	0
POST MOD 0763C88				
63/20/00/000/000/610				
Left input housing				
	368A32-0131-00	(-)	40000 FH	0
			// 200000 TC	0
POST MOD 0763C88				
63/20/00/000/000/615				
Bevel wheel support equipped				
	368A32-0140-00	(-)	40000 FH	0
POST MOD 0763C88				
63/20/00/000/000/620				
Main bevel wheel				
	368A32-0240-00	(-)	1100 FH	0
	368A32-0240-01	(-)		
POST MOD 0763C88				
63/20/00/000/000/625				
Tail output wheel				
	368A32-0250-20	(-)	40000 FH	0
POST MOD 0763C88				
63/20/00/000/000/630				
Sun gear				
	368A32-0270-20	(-)	40000 FH	0
POST MOD 0763C88				
63/20/00/000/000/635				
Servocontrol fitting				
	368A32-0600-00	(-)	5330 FH	0
POST MOD 0763C88				
63/20/00/000/000/640				
Servocontrol axis and link servocontrol fitting to main housing				
	360A32-3173-20	(-)	40000 FH	0
	360A32-3173-21	(-)		
	368A32-0957-20	(-)		
POST MOD 0763C88				
63/20/00/000/000/645				
Rear output bevel gear				
	368A32-0262-20	(-)	25800 FH	0
POST MOD 0763C88				

Task Number Description/Remarks	MP/N (PN)		Type of limit: SLL	Margin
63/20/00/000/000/650 **Fixed rear output flange**				
	368A32-0405-20	(-)	1638 FH	0
POST MOD 0763C88				
63/20/00/000/000/655 **Sliding rear output flange**				
	368A32-0410-20	(-)	15600 FH	0
POST MOD 0763C88				
63/20/00/000/000/660 **Tail output flexible coupling**				
	365A34-1033-01	(-)	28800 FH	0
POST MOD 0763C88				
63/20/00/000/000/665 **Planet gear**				
	368A32-0291-20	(-)	40000 FH	0
POST MOD 0763C88 Part followed according to equipped planet gear MP/N 368A32-0290-01.				
63/20/00/000/000/670 **Fixed gear**				
	368A32-0300-20	(-)	40000 FH	0
POST MOD 0763C88				
63/20/00/000/000/675 **Planet carrier**				
	368A32-0280-21	(-)	9060 FH	0
	368A32-0280-22	(-)	// 45300 TC	0
POST MOD 0763C88				
63/20/00/000/000/680 **Upper cover**				
	368A32-0150-00	(-)	3350 FH	0
POST MOD 0763C88				
63/20/00/000/000/685 **Right seat hydraulic pump**				
	368A32-0170-01	(-)	40000 FH	0
POST MOD 0763C89				
63/20/00/000/000/690 **Left seat hydraulic pump**				
	368A32-0180-01	(-)	40000 FH	0
POST MOD 0763C90				

63-30 MOUNTS AND ATTACHMENTS

63/30/00/000/000/071 **MGB forward suspension bar**				
	365A38-4520-01	(-)	5560 FH	0

Task Number Description/Remarks	MP/N (PN)		Type of limit: SLL	*Margin*
63/30/00/000/000/076 **MGB aft right suspension bar**	365A38-4521-02	(-)	8600 FH	*0*
63/30/00/000/000/081 **MGB aft left suspension bar**	365A38-4521-03	(-)	8600 FH	*0*
63/30/00/000/000/105 **Suspension bar/fitting bolt**	704A33-633-233	(-)	20000 FH	*0*

End of the Document Unit

Task Number Description/Remarks	MP/N (PN)		Type of limit: SLL	*Margin*

ATA 64-TAIL ROTOR

64-10 ROTOR BLADES

64/10/00/000/000/000

Tail rotor blade

365A12-0060-01	(-)	1500 FH	*0*

This part number must not be mixed with the part numbers 365A12-0070-00 and 365A12-0070-01.

64/10/00/000/000/005

Tail rotor blade

365A12-0070-00	(-)	5000 FH	*0*
365A12-0070-01	(-)		

These part numbers must not be mixed with the part number 365A12-0060-01.

64-20 TAIL ROTOR HEAD

64/20/00/000/000/225

Tail rotor hub

365A33-3511-00	(-)	13000 FH	*0*
365A33-3511-01	(-)		
365A33-3511-02	(-)		
365A33-3511-03	(-)		

64/20/00/000/000/230

Plate

365A33-3522-20	(-)	20000 FH	*0*

64/20/00/000/000/240

Splined flange

365A33-3523-00	(-)	16000 FH	*0*

64/20/00/000/000/250

Torsion bar bolt

365A33-3525-20	(-)	20000 FH	*0*

64/20/00/000/000/270

Blade attaching bolt

365A33-3535-20	(-)	20000 FH	*0*

End of the Document Unit

Task Number Description/Remarks	MP/N (PN)		Type of limit: SLL	*Margin*

ATA 65-TAIL ROTOR DRIVE

65-10 TAIL ROTOR DRIVE SHAFT

65/10/00/000/000/006 **Flexible coupling**				
	365A34-1033-01	(-)	20000 FH	*0*
65/10/00/000/000/020 **Forward shaft element**				
	365A34-4510-00	(-)	6300 FH	*0*
65/10/00/000/000/035 **Center shaft element**				
	365A34-4520-00	(-)	6300 FH	*0*
65/10/00/000/000/055 **Aft shaft element**				
	365A34-4530-00	(-)	10000 FH	*0*

65-20 TAIL GEARBOX

65/20/00/000/000/025 **Input pinion**				
	365A33-6123-22	(-)	11700 FH	*0*
	365A33-6123-24	(-)		
65/20/00/000/000/035 **Bevel gear wheel**				
	365A33-6104-00	(-)	20000 FH	*0*
	365A33-6104-01	(-)		
	365A33-6104-02	(-)		
	365A33-6104-03	(-)		
	365A33-6104-04	(-)		
	365A33-6104-05	(-)		
65/20/00/000/000/055 **Sliding flange**				
	365A33-6197-21	(-)	8400 FH	*0*
	365A33-6197-22	(-)		
65/20/00/000/000/065 **Input flange**				
	365A33-6196-21	(-)	11700 FH	*0*
	365A33-6196-22	(-)		
	365A33-6196-23	(-)		
	365A33-6196-24	(-)		
65/20/00/000/000/090 **Control shaft**				
	365A33-6214-20	(-)	20000 FH	*0*
	365A33-6214-21	(-)		
	365A33-6214-22	(-)		
	365A33-6214-23	(-)		

Task Number Description/Remarks	MP/N (PN)		Type of limit: SLL	*Margin*
65/20/00/000/000/105				
Control rod				
	365A33-6145-21	(-)	20000 FH	*0*
65/20/00/000/000/115				
Input housing				
	365A33-6128-00	(-)	20000 FH	*0*
	365A33-6128-01	(-)		
For blades on which the optional MOD OP61B33 has been embodied, should one or more polyurethane protective plates be lost, each flying hour logged without a protective plate shall be counted double.				
65/20/00/000/000/125				
Housing				
	365A33-6137-02	(-)	20000 FH	*0*
	365A33-6137-03	(-)		
65/20/00/000/000/135				
Flared coupling tube				
	365A33-6141-20	(-)	20000 FH	*0*
	365A33-6141-21	(-)		
	365A33-6141-22	(-)		
	365A33-6141-23	(-)		
Element followed through its own log card. For blades on which the optional MOD OP61B33 has been embodied, should one or more polyurethane protective plates be lost, each flying hour logged without a protective plate shall be counted double.				

End of the Document Unit

Task Number Description/Remarks	MP/N (PN)	Type of limit: SLL	*Margin*

ATA 67-ROTORS FLIGHT CONTROL

67-30 SERVO-CONTROLS

67/30/00/000/000/000 **Tail rotor servo-control**				
	SC7291	(704A44831153)	9000 FH	*0*
67/30/00/000/000/010 **Main rotor servo-control**				
	SC8037 SC8037-1	(704A44831149) (704A44831155)	10900 FH	*0*
67/30/00/000/000/020 **Servo-control end-fitting**				
	1211350P 1211351P	(704A33632062) (704A33632061)	3400 FH	*0*

End of the Document Unit

Task Number Description/ Remarks	MP/N (PN)	Type of limit: CHK	*Margin*	Documentation

ATA 25-EQUIPMENTS/FURNISHINGS

25-69 OPT - LIFE RAFTS

25/69/06/200/000/100

Life raft

200-188	(704A34130230)	600 FH	*0*	*AMM 25.69.01.222*
		// 1 Y	*0*	

PRE MOD 0725C69

Check of absence of corrosion inside the junction box.

End of the Document Unit

Task Number Description/ Remarks	MP/N (PN)	Type of limit: CHK	*Margin*	Documentation

ATA 26-FIRE PROTECTION

26-21 ENGINE FIRE EXTINGUISHING

26/21/00/222/000/005 **Engine fire extinguisher - Cylinder**				
1023000	(704A42820043)	6 M	*0*	*AMM 26.21.00.222*
Weighing.				

26-23 BAGGAGE COMPARTMENT FIRE EXTINGUISHING

26/23/01/221/000/000 **Fire extinguisher - Cylinder**				
864250-00	(704A42820042)	100 FH // 2 M	*10 FH* *0*	*AMM 26.23.01.221*
Weighing.				

End of the Document Unit

Task Number Description/ Remarks	MP/N (PN)	Type of limit: CHK	*Margin*	Documentation
53/10/00/000/000/000 **Transmission deck - MGB forward fitting** Check	365A21-8117-00 (-) 365A21-8117-01 (-)	600 FH	*60 FH*	*AMM 53.41.00.991* *53.41.00.991.001* *GRT 06*
53/10/00/000/000/015 **Transmission deck - MGB aft fitting** Check.	365A21-3355-04 (-) 365A21-3355-05 (-)	600 FH	*60 FH*	*AMM 53.41.00.991* *53.41.00.991.002* *GRT 06*

ATA 53-FUSELAGE

53-10 PRIMARY STRUCTURE

End of the Document Unit

Task Number Description/ Remarks	MP/N (PN)	Type of limit: CHK	Margin	Documentation

ATA 62-ROTOR (S)

62-10 MAIN ROTOR BLADES

62/10/00/000/000/000

Main rotor blade - Tip cap

365A11-0080-00	(-)	600 FH	*60 FH*	*AMM 62.10.00.223*

Remove the blade tip cap and do a check of the blade-to-tip cap attachment area.

62-20 MAIN ROTOR HEAD

62/20/00/000/000/110

Spherical thrust bearing

ALL MP/N	(-)	15 FH	*0*	*AMM 05.22.00.213* *05.22.00.213.003*

Check the elastomer section.

62/20/00/000/000/111

Spherical thrust bearing

57910600	(704A33633213)	600 FH	*60 FH*	*AMM 62.24.02.991* *62.24.02.991.001* *GRT 01*

Check the reinforcements.
This check can be performed every 1320 FH provided that the interval between two checks doesn't exceed three years.

62/20/00/000/000/160

Frequency adapter

ALL MP/N	(-)	15 FH	*0*	*AMM 05.22.00.213* *05.22.00.213.003*

Check the elastomer section.

62/20/00/000/000/185

Blade horn - Attach beam bolt

704A33-633-226 704A33-633-227	(704A33633226) (704A33633227)	1200 FH	*120 FH*	*AMM 62.24.03.991* *62.24.03.991.001*

Check.

62/20/00/000/000/196

Pitch change rod - Blade horn bolt

704A33-633-228	(-)	1200 FH	*120 FH*	*AMM 62.26.01.991* *62.26.01.991.001*

Check.

Task Number Description/ Remarks	MP/N (PN)	Type of limit: CHK	Margin	Documentation
62/20/00/000/000/205 **Spherical thrust bearing - Attach beam bolt** Check.	704A33-633-224 (-)	600 FH	60 FH	AMM 62.24.02.991 62.24.02.991.002 GRT 01
62/20/00/000/000/215 **Frequency adapter - Attach beam bolt** Check.	704A33-633-242 (-)	1200 FH	120 FH	AMM 62.25.01.991 62.25.01.991.001
62/20/00/000/000/290 **Swashplate bearing** Grease.	ALL MP/N (-)	100 FH	10 FH	AMM 62.23.02.641 62.23.02.641.001
62/20/00/000/000/295 **Swashplate bearing** Grease.	ALL MP/N (-)	600 FH	60 FH	AMM 62.23.02.641 62.23.02.641.002 GRT 01
62/20/00/000/000/296 **Non-rotating swashplate** Check.	365A31-4531-01 (-) 365A31-4531-02 (-)	600 FH // 2 Y	60 FH 0	AMM 62.23.02.991 62.23.02.991.001 GRT 01
62/20/00/000/000/305 **Rotating swashplate** Check.	365A31-4541-01 (-) 365A31-4541-02 (-)	1200 FH	120 FH	AMM 62.23.02.991 62.23.02.991.002
62/20/00/000/000/320 **Pitch - Change rod** Check.	ALL MP/N (-)	15 FH	0	AMM 05.22.00.213 05.22.00.213.003

Task Number Description/ Remarks	MP/N (PN)	Type of limit: CHK	*Margin*	Documentation
62/20/00/000/000/348 **Scissors lower link** Check.	365A31-1116-03 (-) 365A31-1116-04 (-)	600 FH	*60 FH*	*AMM 62.23.01.991* *62.23.01.991.001* *GRT 01*
62/20/00/000/000/358 **Scissors upper link** Check.	365A31-1115-03 (-) 365A31-1115-04 (-) 365A31-1115-05 (-)	600 FH	*60 FH*	*AMM 62.23.01.991* *62.23.01.991.002* *GRT 01*
62/20/00/000/000/366 **Flared housing - Forward bar attachment** Check.	365A31-4591-00 (-) 365A31-4591-01 (-) 365A31-4591-02 (-) 365A31-4591-03 (-)	600 FH	*60 FH*	*AMM 62.22.00.991* *62.22.00.991.001* *GRT 01*
62/20/00/000/000/368 **Flared housing - Skin** Check.	365A31-4591-00 (-) 365A31-4591-01 (-) 365A31-4591-02 (-) 365A31-4591-03 (-)	600 FH	*60 FH*	*AMM 62.22.00.991* *62.22.00.991.002* *GRT 01*
62/20/00/000/000/390 **Rotor shaft** Check.	365A31-4521-00 (-) 365A31-4521-02 (-) 365A31-4521-03 (-) 365A31-4521-04 (-) 365A31-4521-05 (-) 365A31-4521-06 (-) 365A31-4521-07 (-)	1200 FH	*120 FH*	*AMM 62.21.00.991* *62.21.00.991.001*
62/20/00/605/000/020 **Mast and/or hub** PRE MOD 0762C42 Check the nut MP/N 360A31-1020-20.	365A31-4500-00 (-) 365A31-4500-01 (-) 365A31-4500-03 (-)	600 FH	*60 FH*	*AMM 62.20.00.291* *GRT 01*

End of the Document Unit

Task Number Description/ Remarks	MP/N (PN)	Type of limit: CHK	*Margin*	Documentation

ATA 63-ROTOR DRIVE (S)

63-10 ENGINE/MGB COUPLING

63/10/00/000/000/120 **Front half gimbal joints** 368A32-0624-00 (-) POST MOD 0763C88 Detailed inspection.		800 FH	*0*	*AMM 63.10.64.222* *63.10.64.222.004*
63/10/00/000/000/125 **Rear half gimbal joints** 368A32-0630-00 (-) POST MOD 0763C88 Detailed inspection.		540 FH	*0*	*AMM 63.10.64.222* *63.10.64.222.003*
63/10/00/000/000/130 **Gimbal joint ring** 368A32-0625-00 (-) POST MOD 0763C88 Detailed inspection.		600 FH	*60 FH*	*AMM 63.10.64.222* *63.10.64.222.002*
63/10/00/000/000/135 **Link front gimbal joint to input housing** POST MOD 0763C88 Tightening torque check.		100 FH	*0*	*AMM 63.10.64.721* *63.10.64.721.002*
63/10/00/000/000/140 **Link torque tube to rear gimbal joint** POST MOD 0763C88 Tightening torque check.		50 FH	*0*	*AMM 63.10.64.721* *63.10.64.721.001*

63-20 MAIN GEARBOX

Task Number Description/ Remarks	MP/N (PN)	Type of limit: CHK	Margin	Documentation
63/20/00/000/000/001 **MGB - Chip detector**	ALL MP/N (-)	10 FH	0	AMM 12.20.00.211
Check of absence of particles on chip detectors.				
63/20/00/000/000/056 **Planet gear carrier**	360A32-3036-20 (-) 360A32-3036-21 (-)	50 FH	0	AMM 63.20.00.291
PRE MOD 0763C88 Borescope inspection.				
63/20/00/000/000/548 **MGB bottom**	366A32-1060-26 (-)	15 FH	0	AMM 63.20.00.212
PRE MOD 0763C88 Visual check. Operation to be carried out from 2600 cycles onwards. Part : assembly 366A32-1060-26 : 366A32-1062-06				
63/20/00/000/000/700 **MGB - Chip detector**		100 FH	0	AMM 63.40.62.711
POST MOD 0763C88 Operational check of the chip detector.				
63/20/00/000/000/710 **MGB plate (Bottom plate)**		40 FH	0	AMM 63.20.60.281
POST MOD 0763C88 Check the absence of particles in MGB bottom plate.				
63/20/00/000/000/720 **Upper cover**	368A32-0150-00 (-)	447 FH	0	AMM 63.20.60.212
POST MOD 0763C88 Check.				

Task Number Description/ Remarks	MP/N (PN)	Type of limit: CHK	Margin	Documentation
63/20/00/000/000/730 **Main housing** POST MOD 0763C88 Check.	368A32-0100-01 (-)	2722 FH // 13600 TC	0 0	AMM 63.20.60.217
63/20/00/000/000/740 **Servocontrol axis** POST MOD 0763C88 Tightening torque check. Interval since MGB new or overhauled.	368A32-0957-20 (-)	1620 FH	0	AMM 63.20.60.722
63/20/00/000/000/750 **Main actuator fitting** POST MOD 0763C88 Detailed inspection.	368A32-0600-00 (-)	280 FH	0	AMM 63.20.60.222 63.20.60.222.001
63/20/00/000/000/760 **Servocontrol axis** POST MOD 0763C88 Detailed inspection. Interval since MGB new or overhauled.	368A32-0957-20 (-)	8130 FH	0	AMM 63.20.60.222 63.20.60.222.002
63/20/00/000/000/770 **MGB - Oil filter** POST MOD 0763C88 Filter clogging check.		50 FH	0	AMM 63.21.62.211
63/20/00/000/000/780 **Input flexible coupling (MGB side)** PRE MOD 0763C88 Detailed visual check.	365A32-2872-00 (-) 365A32-6530-00 (-)	15 FH	0	AMM 05.22.00.213 05.22.00.213.003
63/20/00/000/000/790 **Tail output flexible coupling (MGB side)** POST MOD 0763C88 General visual inspection.		15 FH	0	AMM 63.20.60.219 63.20.60.219.004

Task Number Description/ Remarks	MP/N (PN)	Type of limit: CHK	Margin	Documentation
63/20/00/000/000/800 **MGB housing links** POST MOD 0763C88 Tightening torque check. Interval since MGB new or overhauled.		1425 FH // 7125 TC	0 0	AMM 63.20.60.721
63/20/00/000/000/810 **Link sliding rear output flange to flexible coupling** POST MOD 0763C88 Tightening torque check. Interval since MGB new or overhauled.		6100 FH	0	AMM 65.10.00.222 65.10.00.222.001
63/20/00/000/000/820 **All links** POST MOD 0763C88 Discard all links at overhaul.		1100 FH	0	AMM 63.20.60.061
63/20/00/000/000/830 **Thermocontact** POST MOD 0763C88 Functional test of the temperature switch.		1100 FH	0	AMM 63.40.63.721 63.40.63.721.001
63/20/00/000/000/840 **Thermocontact** POST MOD 0763C88 Functional test of the temperature switch electrical line.		1100 FH	0	AMM 63.40.63.721 63.40.63.721.002
63/20/00/000/000/850 **Upper cover** POST MOD 0763C88 Tightening torque check of the upper cover links. Interval since MGB new or overhauled.	368A32-0150-00 (-)	2180 FH	0	AMM 63.20.60.722
63/20/00/000/000/860 **Fixed rear output flange** POST MOD 0763D10 Check the state of the part.	368A32-0405-21 (-)	1639 FH	0	AMM 63.24.61.211

Task Number Description/ Remarks	MP/N (PN)	Type of limit: CHK	*Margin*	Documentation

63-30 MOUNTS AND ATTACHMENTS

63/30/00/000/000/069 **MGB forward suspension bar** Check.	365A38-4520-01 (-)	1200 FH	*120 FH*	*AMM 63.30.01.991* *63.30.01.991.001*
63/30/00/000/000/074 **MGB aft right suspension bar** Check.	365A38-4521-02 (-)	1200 FH	*120 FH*	*AMM 63.30.01.991* *63.30.01.991.002*
63/30/00/000/000/079 **MGB aft left suspension bar** Check.	365A38-4521-03 (-)	1200 FH	*120 FH*	*AMM 63.30.01.991* *63.30.01.991.002*

End of the Document Unit

Task Number Description/ Remarks	MP/N (PN)	Type of limit: CHK	Margin	Documentation

ATA 64-TAIL ROTOR

64-10 ROTOR BLADES

64/10/00/000/000/020

Tail rotor blade - Stainless steel ring

365A12-0060-01 365A12-0070-00	(-) (-)	15 FH	0	AMM 05.22.00.213 05.22.00.213.004

Visual check.
Without blade removal.

64-20 TAIL ROTOR HEAD

64/20/00/000/000/224

Tail rotor hub

365A33-3511-00 365A33-3511-01 365A33-3511-02 365A33-3511-03	(-) (-) (-) (-)	1200 FH	120 FH	AMM 64.20.04.991 64.20.04.991.001

Check.

64/20/00/000/000/238

Splined flange

365A33-3523-00	(-)	1200 FH	120 FH	AMM 64.20.00.992 64.20.00.992.001

Check.

64/20/00/000/000/245

Torsion bar

365A33-3527-00	(-)	2000 FH	0	AMM 64.20.00.225 64.20.00.225.008

Check.

64/20/00/420/211/000

Tail rotor hub

365A33-3511-00 365A33-3511-01 365A33-3511-02	(-) (-) (-)	50 FH	5 FH	AMM 64.20.04.211

Detailed visual inspection.

End of the Document Unit

Task Number Description/ Remarks	MP/N (PN)	Type of limit: CHK	Margin	Documentation

ATA 65-TAIL ROTOR DRIVE

65-10 TAIL ROTOR DRIVE SHAFT

65/10/00/000/000/004

Flexible coupling

365A34-1033-01	(-)	15 FH	0	AMM 05.22.00.213 05.22.00.213.003

MGB output tail rotor drive shaft flexible coupling.
Detailed visual check.

65/10/00/000/000/005

Flexible coupling

365A34-1033-01	(-)	50 FH	5 FH	AMM 65.10.00.211 65.10.00.211.001

All tail rotor drive shaft flexible couplings.
Detailed visual check without removal.

65/10/00/000/000/018

Forward shaft element

365A34-4510-00	(-)	600 FH	60 FH	AMM 65.10.01.991 65.10.01.991.001 GRT 02

Check.

65/10/00/000/000/033

Center shaft element

365A34-4520-00	(-)	600 FH	60 FH	AMM 65.10.02.991 65.10.02.991.001 GRT 02

Check.

65/10/00/000/000/053

Aft shaft element

365A34-4530-00	(-)	600 FH	60 FH	AMM 65.10.03.991 65.10.03.991.001 GRT 02

Check.

65-20 TAIL GEARBOX

65/20/00/000/000/015

Flexible coupling

365A33-6213-00	(-)	50 FH	5 FH	AMM 65.10.00.211 65.10.00.211.001

Detailed visual check.

End of the Document Unit