
BOLLETTINO TECNICO

The technical content of this document is approved
under the authority of DOA nr. EASA.21J.005

N° **139-311**

DATE February 21, 2014

REV. **B** – June 4, 2014

Compliance with
this bulletin is:

MANDATORY

SUBJECT

**SUBFLOOR FRAME STA2105 RETROMODIFICATION P/N 3G5306P47211,
INSTALLATION OF.**

REASON

To reinforce the Tail Rotor (TR) pedals support frame, performing interim repetitive inspections until implementation of the retromodification.

HELICOPTERS AFFECTED

All AB139/AW139 helicopters from 31005 (except S/N 31007) to S/N 31517 (except S/N's 31415, 31431, 31491, 31500, 31508, 31516) and from S/N 41001 to S/N 41356 (except S/N 41355).

NOTE

Implementation of this revision is not required for helicopters that have already successfully complied with Part II of this Bollettino Tecnico.

An appropriate entry should be made in the aircraft log book upon accomplishment.
If ownership of aircraft has changed, please, forward to new owner.

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COMPLIANCE

Part I: within thirty (30) flight hours or two (2) months, whichever comes first, and then every three hundred (300) flight hours or six (6) months until accomplishment of part II.

Part II: immediately after the inspection, if cracks are detected while complying with part I, or within twelve hundred (1200) flight hours, if cracks are not identified.

DESCRIPTION

Cracks on the TR pedals support frames have been reported on some aircraft in service. Part II of this Bollettino Tecnico provides the instructions to implement appropriate structural reinforcements acting both as a preventive measure and as a repair in case of cracks. Moreover specific periodic inspections are introduced in Part I in order to monitor aircraft in service up to the introduction of the reinforcements. These retromodifications have been already introduced on new build aircraft.

Revision A introduces some changes to the installation fixing hardware related to the central reinforcement repair while in revision B some editorial changes have been made in the compliance instructions to make them clearer.

REQUIRED MANPOWER

It is estimated that, in order to accomplish this Bollettino Tecnico, the following maintenance man-hours (MMH) are necessary:

Part I: two (2) MMH;

Part II: two hundred and forty (240) MMH.

Man-hours are based on hands-on time and may vary with personnel and facilities available.

WARRANTY

- Owners/Operators who comply with the instructions in this Bulletin, no later than the applicable date in the "Compliance section" will be eligible to receive replacements parts on a "free of charge basis", with the exception of consumable materials.

NOTE: Customers who fail to comply with the instructions in this bulletin before the compliance date are not eligible for the aforementioned special

- Please destroy locally the removed parts.

REQUIRED MATERIALS

The following materials, necessary to comply with part II of this Bollettino Tecnico, must be requested to AgustaWestland:

P/N	DESCRIPTION	Q.TY	NOTE
3G5306P47211	SUBFLOOR FRAME STA2105 RETROMOD	Ref	
Composed of:			
3G5306P47251	Shim	1	
3G5331A45851	Profile RH	1	
3G5331A45951	Profile LH	1	
3G5331A46051	Reinforcement	2	
3G5331A46152	Central Reinforcement	1	(1)
3G5331A46252	Central Profile	1	
3G0203R10511	SUBFLOOR FRAME STA2105 REPAIR	Ref	(2)
Composed of:			
3G5331A46451	Shim	2	(2)
3G5331A46551	Doubler	2	(2)
3G5331A46651	Doubler	1	(2)
MS21069L4	Nut, self-locking, plate	6	
MS21071L4	Nut, self-locking, plate	2	
NAS6604H9	Bolt	4	

Moreover, the following consumable materials are necessary to comply with part II of this Bollettino Tecnico:

<u>SPEC./AW Code Number</u>	<u>DESCRIPTION</u>	<u>Q.TY</u>	<u>NOTE</u>
MS20427M4-5	Rivet, solid	AR	Local supply

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<u>SPEC./AW Code Number</u>	<u>DESCRIPTION</u>	<u>Q.TY</u>	<u>NOTE</u>
MS20427M4-7	Rivet, solid	AR	Local supply
MS20470AD4-4	Rivet, solid	AR	Local supply
MS20470AD4-5	Rivet, solid	AR	Local supply
MS20470AD4-6	Rivet, solid	AR	Local supply (2)
MS20615-4M3	Rivet, solid	AR	Local supply
MS20615-4M5	Rivet, solid	AR	Local supply
MS20615-4M6	Rivet, solid	AR	Local supply
NAS1097AD4-4	Rivet, solid	AR	Local supply
NAS9301B-4-02	Rivet, blind	AR	Local supply
NAS9301B-4-03	Rivet, blind	AR	Local supply (3)
NAS9301B-4-04	Rivet, blind	AR	Local supply
NAS9301B-5-05	Rivet, blind	AR	Local supply
NAS9303B-4-03	Rivet, blind	AR	Local supply (2)
NAS9307M-4-02	Rivet, blind	AR	Local supply
NAS9307M-4-03	Rivet, blind	AR	Local supply (2)
NAS9307M-4-04	Rivet, blind	AR	Local supply
NAS9308M-4-03	Rivet, blind	AR	Local supply
MS20426AD3-6	Rivet	AR	Local supply
MS20426AD3-7	Rivet	AR	Local supply
MS20426AD4-4	Rivet	AR	Local supply
199-05-004 Ty. II Cl. B-2 MIL-S-8802 cod. 900001586	Sealant, Proseal 890B2 (C153)	AR	
199-05-002 Ty. I Cl. 2 cod. 900000581	Adhesive, Epoxy, EA9309NA (C231)	AR	(2)
TT-M-261	Solvent, Methyl-ethyl-ketone (MEK) (C005)	AR	(2)

Notes

(1) May be supplied with productive P/N 3G5331A46152A.

- (2) Required only if cracks are found performing inspection described in part I of this Bollettino Tecnico or if reinforcements/doublers had been previously applied according to ASRP or repair drawings.
- (3) Rivet P/N NAS9301BNS-4-03 may be supplied as an alternative.

SPECIAL TOOLS

The following special tools are necessary to accomplish this Bollettino Tecnico:

Part I

<u>P/N</u>	<u>DESCRIPTION</u>	<u>Q.TY</u>	<u>NOTES</u>
Local supply	Flashlight	1	

Part II

<u>P/N</u>	<u>DESCRIPTION</u>	<u>Q.TY</u>	<u>NOTES</u>
3G0705G00232 or an approved alternative	Jack, hydraulic (HA-01-00)	4	

WEIGHT AND BALANCE CHANGES

Compliance with part II of this Bollettino Tecnico has the following effect on weight and balance:

<u>P/N</u>	<u>WEIGHT</u>	<u>ARM</u>	<u>MOMENT</u>
3G5306P47211 Subfloor frame STA2105 retro-mod	0.689 Kg	2120 mm	1460.7 Kgmm
3G0203R10511 Subfloor frame STA2105 repair	0.877 Kg	2117 mm	1856.6 Kgmm (1)

Notes:

- (1) Weight and balance data of the repair already cover also the retromod application.

REFERENCES

- ✓ AW139 Aircraft Material Data Information (AMDI)
- ✓ AW139 Aircraft Maintenance Publication (AMP)

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- ✓ AW139 Aircraft Structural Repair Publication (ASRP)
- ✓ AW139 Illustrated Tools and Equipment Publication (ITEP)

PUBLICATIONS AFFECTED

- ✓ AW139 Illustrated Parts Data (IPD)

COMPLIANCE INSTRUCTIONS:

Part I

1. Prepare the helicopter for a safe ground maintenance. Disconnect the battery and all the electrical power sources and/or the external power supply. Refer to AMP, see DM n° 39-A-00-20-00-00A-120A-A.
2. With reference to figures 1 and 2, to get access to the area shown in figures 10 and 11, remove/open access doors/panels that follow:

NOTE

With reference to figure 2, it is possible to remove access panels 131AL (6) and 132AR (5) without removing the LH sheet skin assy (7) and the RH sheet skin assy (3) or the FWD float assemblies (where installed), just removing screws (4).

- ✓ access panel 130AL;
- ✓ access panel 131AL;
- ✓ access panel 132AR.

Refer to AMP, see DM n° 39-A-06-41-00-00A-010A-A.

3. With reference to figures 1, do a visual inspection for cracks of the subfloor frames at STA 2105 (LH frame, middle frame and RH frame, shown in figures 1 and 10 thru 13), from both sides (front and rear), using a flashlight.
 - 3.1 If no cracks are found, go to step 4.

- 3.2 If cracks are found, contact Product Support Engineering (aw139.mbx@agustawestland.com) to report about the inspection findings and perform Part II of this Bollettino Tecnico.
4. Return helicopter to flight configuration and record for compliance with part I of this Bollettino Tecnico on the helicopter logbook.

Part II

1. Prepare the helicopter for a safe ground maintenance. Disconnect the battery and all the electrical power sources and/or the external power supply. Refer to AMP, see DM n° 39-A-00-20-00-00A-120A-A.
2. Remove from the cockpit, to get access to the area shown in figures 10 and 11, the elements that follow:
 - ✓ cockpit equipments, liners, floor panels as necessary.
 - ✓ the pilot pedal set. Refer to AMP, see DM n° 39-A-67-21-01-00A-520A-A.
 - ✓ the copilot pedal set. Refer to AMP, see DM n° 39-A-67-21-02-00A-520A-A.
3. With reference to figure 3, remove the LH and RH nose windows. Hold the hardware for further re-use.

CAUTION

Be careful when you remove the central panel at step 4. There are studs and clamps fixing cables to the panel. Make sure to disconnect studs and clamps before pulling down the central panel.

4. With reference to figure 1, remove the central panel from the helicopter.
5. With reference to figures 1 and 2, remove from the helicopter the elements that follow:

NOTE

It is possible to remove access panels 131AL (6) and 132AR (5) without removing the LH sheet skin assy (7) and the RH sheet skin assy (3) or the FWD float

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assemblies (where installed), just removing screws (4).

- ✓ the access panel 131AL. Refer to AMP, see DM n° 39-A-06-41-00-00A-010A-A.
- ✓ the access panel 132AR. Refer to AMP, see DM n° 39-A-06-41-00-00A-010A-A.
- ✓ the control rod Y2. Refer to AMP, see DM n° 39-A-67-21-03-00A-520A-A.
- ✓ the control rod Y3. Refer to AMP, see DM n° 39-A-67-21-04-00A-520A-A.
- ✓ the bellcrank Y2-Y3. Refer to AMP, see DM n° 39-A-67-21-05-00A-520A-A.
- ✓ the control rod Y4. Refer to AMP, see DM n° 39-A-67-21-06-00A-520A-A.

NOTE

Nose landing gear removal at following step 6 is optional. It is possible to apply Subfloor Frame STA2105 Retro-Mod P/N 3G5306P47211 with the nose landing gear installed, provided that the helicopter is lifted on jacks and that operators adjust nose landing gear position while installing central reinforcement P/N 3G5331A46152 and central profile P/N 3G5331A46251 at steps 7.5 and 7.6. To lift the helicopter on jacks, refer to AMP, see data module n° 39-A-07-11-00-00A-000A-A.

With reference to figure 19, it is also allowed to remove AFT channel to get an easier access to the work area. In this case, use rivets P/N NAS1097AD4-4 to re-install AFT channel.

6. Remove the nose landing gear, if required, to get an easier access to the work area. Refer to AMP, see DM n° 39-A-32-21-01-00A-520A-A or DM n° 39-A-32-21-01-00A-520B-A.

WARNING

THE MATERIALS THAT FOLLOW ARE DANGEROUS. BEFORE YOU PERFORM STEP 7 MAKE SURE THAT YOU KNOW ALL THE SAFETY PRECAUTIONS AND FIRST AID INSTRUCTIONS FOR THESE MATERIALS:

- ✓ METHYL-ETHYL-KETONE (C005);
- ✓ PROSEAL 890B2 (C153);

✓ **ADHESIVE (C231).**
REFER TO AMDI, SEE DM N° 39-A-00-50-00-00A-074A-D.

NOTE

As an alternative to rivets indicated in step 7 and in related figures, it is allowed to use high lock pins with related collars. Refer to ASRP (Section 51-40) for details about hi-locks installation preparation.

NOTE

Figures 4 thru 9 show how to apply reinforcement/doublers on areas where cracks have been found. Figures 20 and 21 show subfloor frame STA 2105 after retro-mod installation.

NOTE

Productive P/N may be supplied with pilot holes or without holes in order to allow a field countermark action before finale riveted installation.

7. With reference to figures 4 thru 21, apply Subfloor Frame STA2105 Retro-Mod P/N 3G5306P47211 as follows:

NOTE

If reinforcements/doublers had been previously applied, according to ASRP or repair drawings, in the areas concerning this modification, remove them before proceeding with following step 7.1.

- 7.1. With reference to figure 12, remove existing rivets and self-locking nuts from RH frame in indicated positions.
- 7.2. With reference to figures 12 and 15, install reinforcements on RH frame as follows:

NOTE

Following steps 7.2.1 and 7.2.2 are applicable only if cracks have been found on RH frame (STA2105).

- 7.2.1. Perform cut-out of the affected area according to figure 7 and to figure 9 section D-D.

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NOTE

It is allowed to use a peeling shim at following step 7.2.2 if it is necessary to ensure the flat surface.

- 7.2.2. Put in position shim P/N 3G5331A46451 and doubler P/N 3G5331A46551 on RH profile P/N 3G5331A45851 according to figure 8 and to figure 9 section E-E.
- 7.2.3. Install RH profile P/N 3G5331A45851 using Proseal 890B2 (C153). Fix it to the frame (STA 2105) with rivets P/N MS20470AD4-4 and P/N NAS9301B-5-05 according to figure 12 and with rivets P/N NAS9301B-5-04, P/N NAS9301B-5-05 and P/N NAS9301B-5-06 according to figure 16.

NOTE

Performing following step 7.2.4, make sure to put reinforcement P/N 3G5331A46051 in position to drill the lateral holes at the same distance from both sides. The dimensions indicated in figure 12 are for reference only.

- 7.2.4. Install reinforcement P/N 3G5331A46051 using rivets P/N MS20470AD4-5 according to figure 12 and rivets P/N NAS9301B-4-02 according to figure 15.
 - 7.2.5. With reference to figure 12, install self-locking nuts P/N MS21069L4 according to existing holes on the structure, using rivets P/N MS20426AD3-6.
- 7.3. With reference to figure 13, remove existing rivets and self-locking nuts from LH frame in indicated positions.
 - 7.4. With reference to figures 14 and 15, install reinforcements on LH frame as follows:

NOTE

Following steps 7.4.1 and 7.4.2 are applicable only if cracks have been found on LH frame (STA2105).

- 7.4.1. Perform cut-out of the affected area according to figure 7 and to figure 9 section D-D.

NOTE

It is allowed to use a peeling shim at following step 7.4.2 if it is necessary to ensure the flat surface.

- 7.4.2. Put in position shim P/N 3G5331A46451 and doubler P/N 3G5331A46551 on LH profile P/N 3G5331A45951 according to figure 8 and to figure 9 section E-E.
- 7.4.3. Install LH profile P/N 3G5331A45951 using Proseal 890B2 (C153). Fix it to the frame (STA 2105) with rivets P/N MS20470AD4-4 and P/N NAS9301B-5-05 according to figure 14 and with rivets P/N NAS9301B-5-04, P/N NAS9301B-5-05 and P/N NAS9301B-5-06 according to figure 16.

NOTE

Performing following step 7.4.4, make sure to put reinforcement P/N 3G5331A46051 in position to drill the lateral holes at the same distance from both sides. The dimensions indicated in figure 14 are for reference only.

- 7.4.4. Install reinforcement P/N 3G5331A46051 using rivets P/N MS20470AD4-5 according to figure 14 and rivets P/N NAS9301B-4-02 according to figure 15.
- 7.4.5. With reference to figure 14, install self-locking nuts P/N MS21069L4 according to existing holes on the structure, using rivets P/N MS20426AD3-6.

NOTE

If nose landing gear has not been removed at step 6, adjust its position as required, while performing following steps 7.5 and 7.6.

- 7.5. With reference to figure 17, remove existing omega shape P/N 3P5331A47152 and related rivets and self-locking nuts.

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7.6. Install on middle frame P/N 3P5331A05531 central reinforcement P/N 3G5331A46152 and central profile P/N 3G5331A46252, as follows:

7.6.1. With reference to figures 17 and 18, remove the cover, cut it and rework it as shown to move the self-locking nut in the new position. Seal the existing self-locking nut hole using Proseal 890B2 (C153).

NOTE

Following steps 7.6.2 thru 7.6.4 are applicable only if cracks have been found on middle frame (STA2105).

7.6.2. Make a stop drill at both ends of the crack.

7.6.3. Clean with Methyl-Ethyl-Ketone (C005).

7.6.4. Put in position doubler P/N 3G5331A46651 according to figure 6 and to figure 4 section F-F and countermark holes position for self-locking nuts and rivets installation, from the middle frame on the doubler. Drill 6.50-6.65 mm diameter holes on the doubler.

7.6.5. Put in position reinforcement P/N 3G5331A46152 and central profile P/N 3G5331A46252 according to figures 18 and 19.

NOTE

Following steps 7.6.6 and 7.6.7 are applicable only if cracks have been found on middle frame (STA2105).

7.6.6. Install doubler P/N 3G5331A46651 using adhesive (C231).

7.6.7. Fix the doubler to the middle frame (STA2105) with eight rivets P/N NAS9307M4-02 and to the floor with seven rivets P/N MS20470AD4-6. Seal using Proseal 890B2 (C153) all over the doubler contour.

7.6.8. Fix central profile P/N 3G5331A46252 to the middle frame (STA2105) with rivets P/N NAS9307M-4-04 and P/N NAS9307M-4-02 according to figures 18 and 21.

NOTE

Performing following step 7.6.9 on helicopters where doubler P/N 3G5331A46651 has been installed at step 7.6.6, use six rivets P/N NAS9307M-4-03

instead of P/N MS20427M4-5 to fix the reinforcement on the frame and the doubler, according to figure 6.

- 7.6.9. Fix reinforcement P/N 3G5331A46152 to the middle frame (STA2105) and to the central profile with five rivets P/N MS20615-4M6, four rivets P/N MS20615-4M5, eight rivets P/N MS20427M4-5, seven rivets P/N MS20427M4-7 and five rivets P/N NAS9307M-4-04, using existing holes according to figures 18, 19 and 21.
- 7.7. With reference to figure 18, install self-locking nuts on the central reinforcement as follows:
- 7.7.1. Drill 6.50-6.65 mm diameter holes on central reinforcement according to existing holes on the longeron.

NOTE

On helicopters where doubler P/N 3G5331A46651 has been installed at step 7.6.6, use P/N MS20426AD3-7 instead of P/N MS20426AD3-6 to install self-locking nuts at following steps 7.7.2 and 7.7.3.

- 7.7.2. Install two self-locking nuts P/N MS21069L4 using rivets P/N MS20426AD3-6.
- 7.7.3. Install two self-locking nuts P/N MS21071L4 using rivets P/N MS20426AD3-6.
- 7.8. With reference to figure 19 and 21, fix central profile P/N 3G5331A46251 on the AFT channel using rivets P/N NAS9308-M-4-3 and on the middle lower panel using rivets P/N MS20427M4-5.
- 7.9. Install reworked cover P/N 3G5306P47253 on the middle frame and install removed self-locking nuts P/N MS21069L08 using rivets P/N MS20426AD3-6 according to figure 18.
- 7.10. On helicopters where doubler P/N 3G5331A46651 has been installed, install also self-locking nuts P/N MS21069L08K using two rivets P/N MS20426AD3-6, according to figure 6.

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8. Install the nose landing gear, if you have removed it at step 6. Refer to AMP, see DM n° 39-A-32-21-01-00A-720A-A or DM n° 39-A-32-21-01-00A-720B-A.
9. Install on the helicopter the elements that follow:
 - ✓ the bellcrank Y2-Y3. Refer to AMP, see DM n° 39-A-67-21-05-00A-720A-A.
 - ✓ the control rod Y2. Refer to AMP, see DM n° 39-A-67-21-03-00A-720A-A.
 - ✓ the control rod Y3. Refer to AMP, see DM n° 39-A-67-21-04-00A-720A-A.
 - ✓ the control rod Y4. Refer to AMP, see DM n° 39-A-67-21-06-00A-720A-A.
10. Install in the cockpit the elements that follow:

NOTE

To install pilot/copilot pedal set on the frame portion where reinforcement have been added, use bolts P/N NAS6604H9 instead of the old ones (P/N NAS6604H7).

- ✓ the pilot pedal set. Refer to AMP, see DM n° 39-A-67-21-01-00A-720A-A.
 - ✓ the copilot pedal set. Refer to AMP, see DM n° 39-A-67-21-02-00A-720A-A.
11. Install the central panel using hardware shown in figure 22.
 12. Install other access panels, floor panels, liners and equipments removed at step 2 and step 5.
 13. With reference to figure 3, install the LH and RH nose windows as follows:
 - 13.1. Install the LH nose window (1) using twenty-seven screws (2) P/N MS27039-1-10 and four screws (4) P/N MS27039-1-9 (in the corners) with washers (3).
 - 13.2. Install the RH nose window (5) using twenty-seven screws (6) P/N MS27039-1-10 and four screws (8) P/N MS27039-1-9 (in the corners) with washers (7).

WARNING

THE PROSEAL 890B2 (C153) IS DANGEROUS. MAKE SURE THAT YOU KNOW ALL THE SAFETY PRECAUTIONS AND FIRST AID INSTRUCTIONS FOR THIS MATERIAL. REFER TO AMDI, SEE DM N° 39-A-00-50-00-00A-074A-D.

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- 13.3. Seal the head of the screws and the contour of the nose windows with Proseal 890B2 (C153).
14. In accordance with weight and balance changes, update the Chart A (see Rotorcraft Flight Manual, Part II, section 6).
15. Return helicopter to flight configuration and record for compliance with part II of this Bollettino Tecnico on the helicopter logbook.

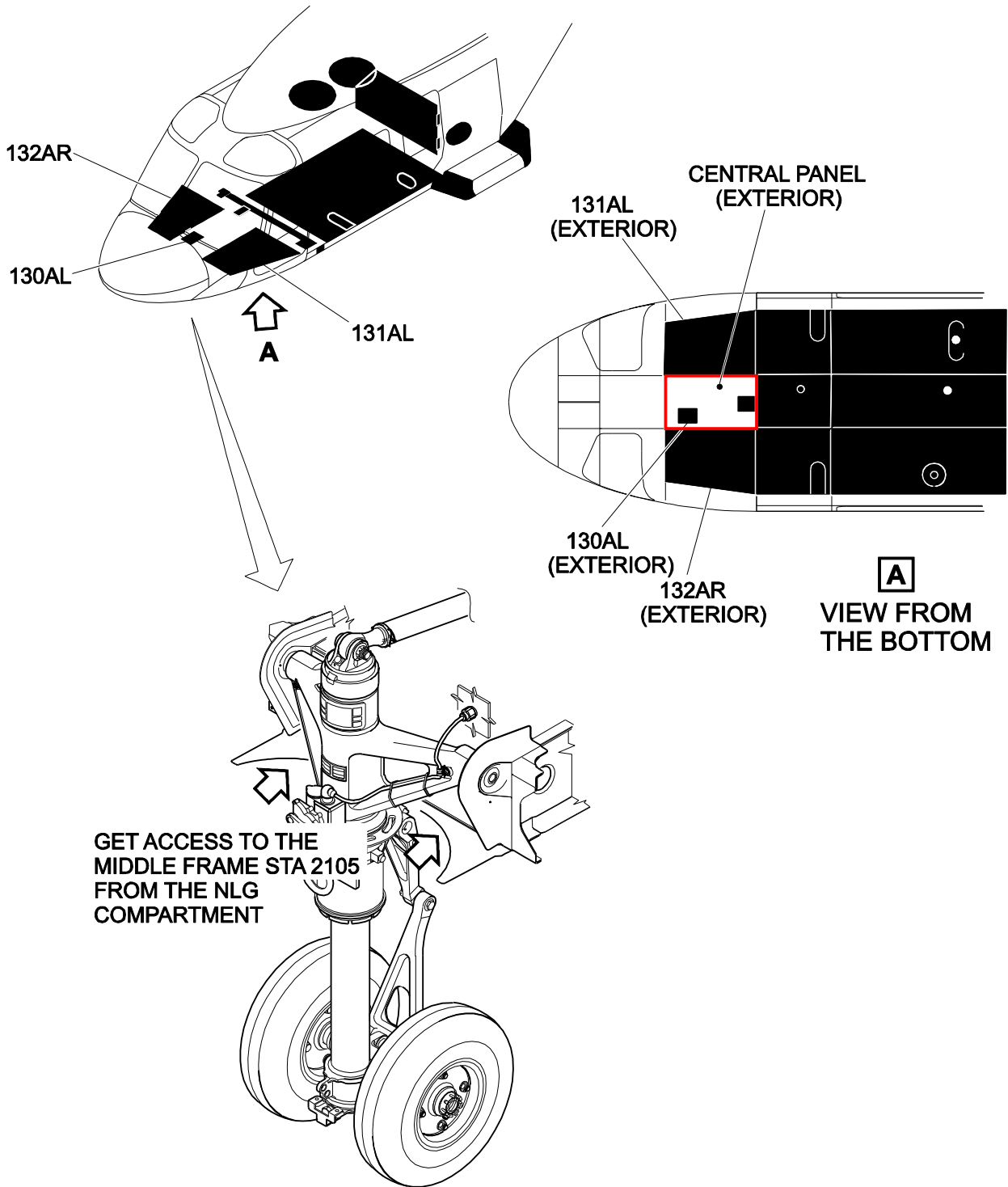


Figure 1

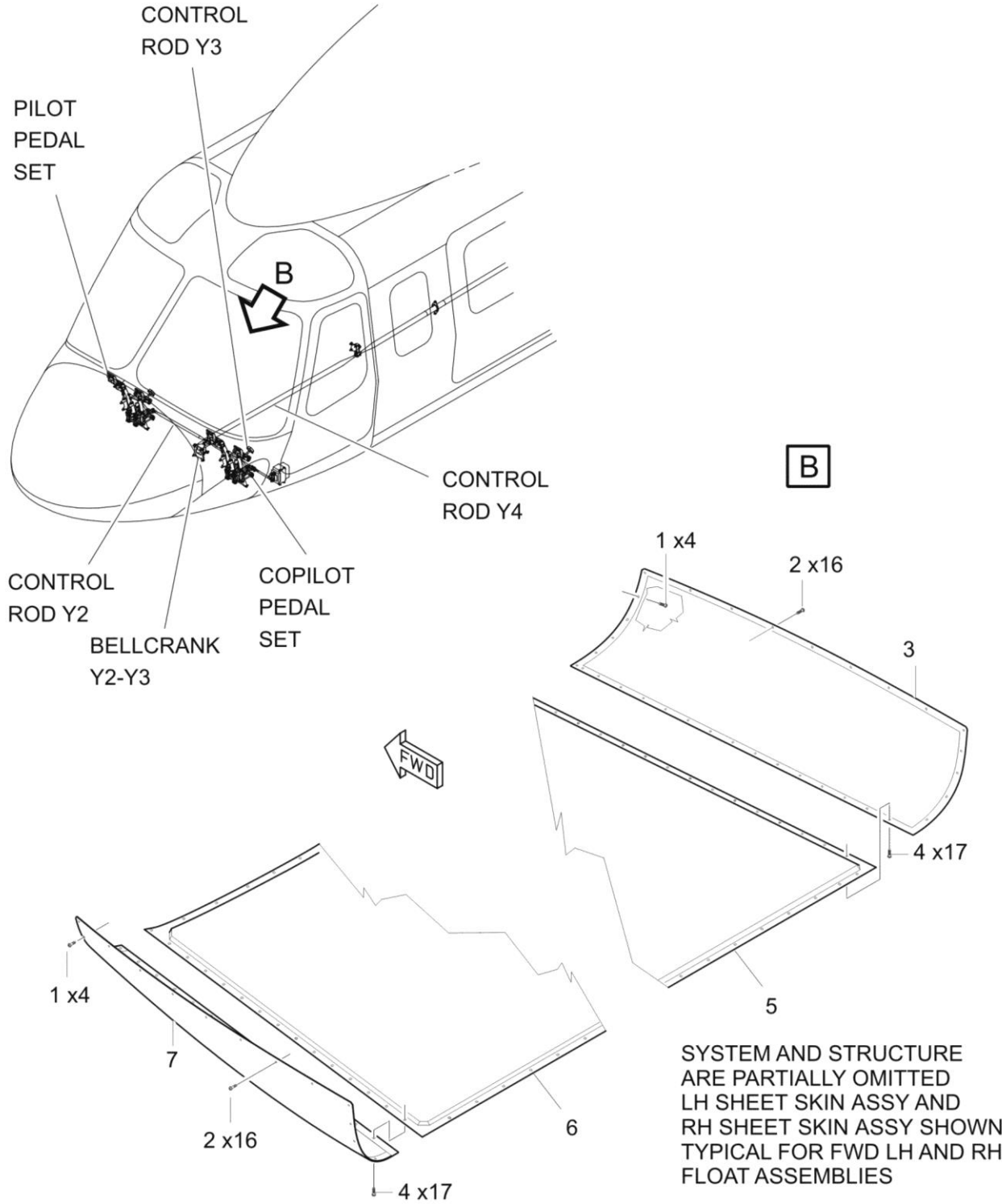


Figure 2

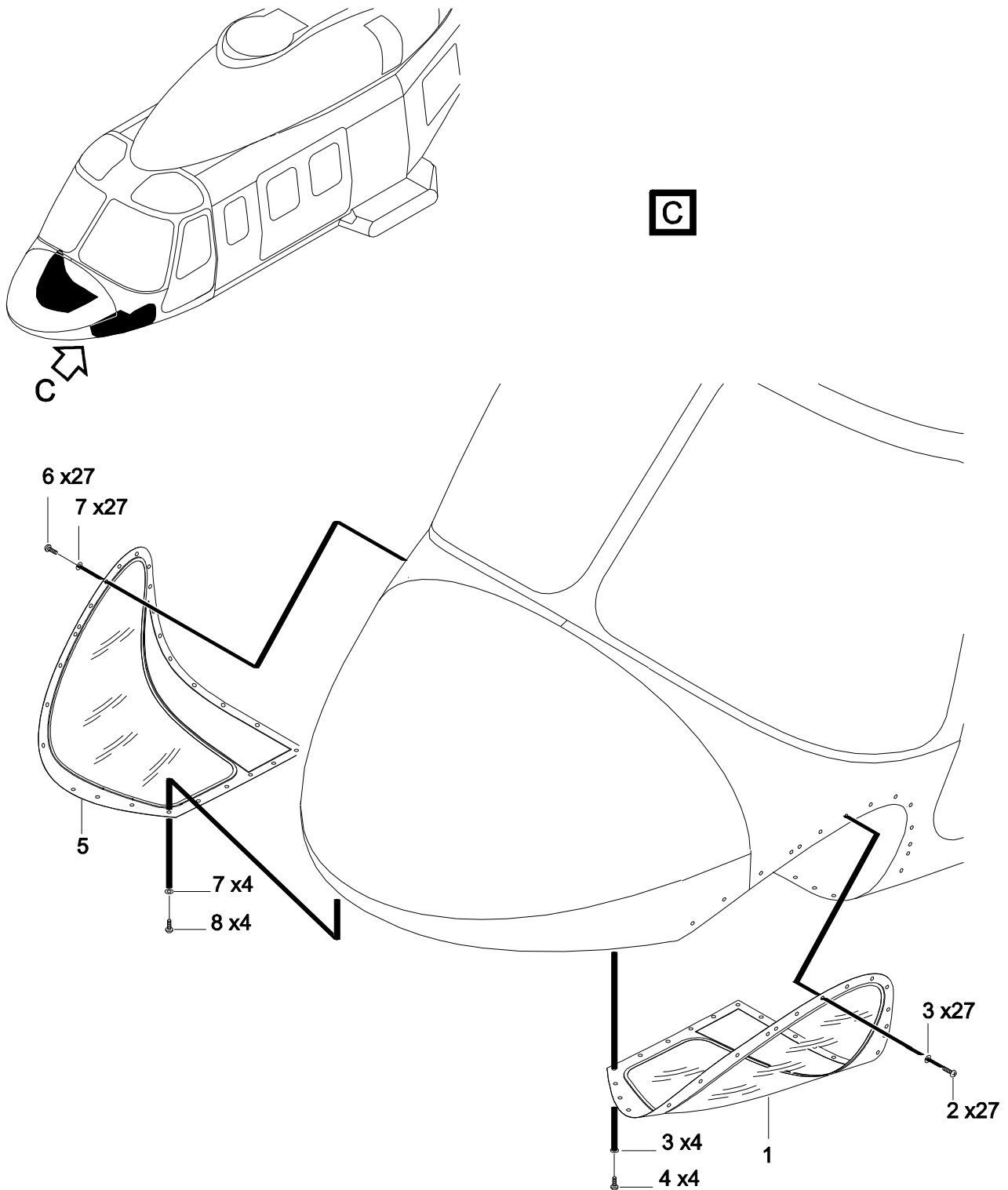
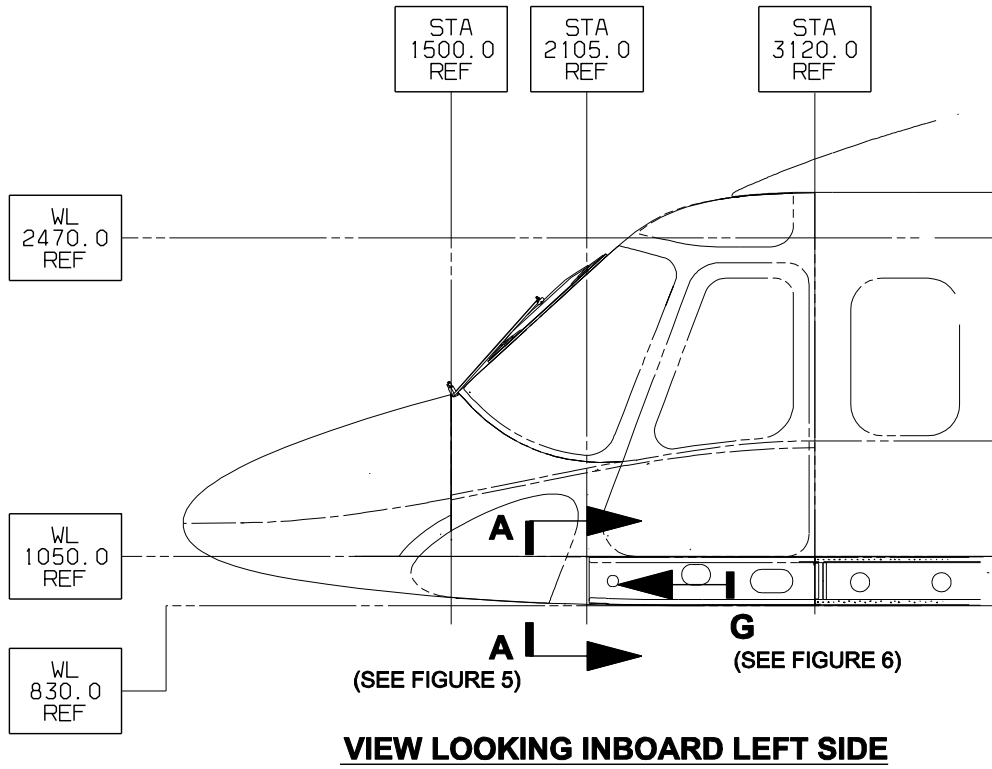


Figure 3



SECTION F-F
SCHEMATIC ONLY
(REFER TO FIGURE 6)

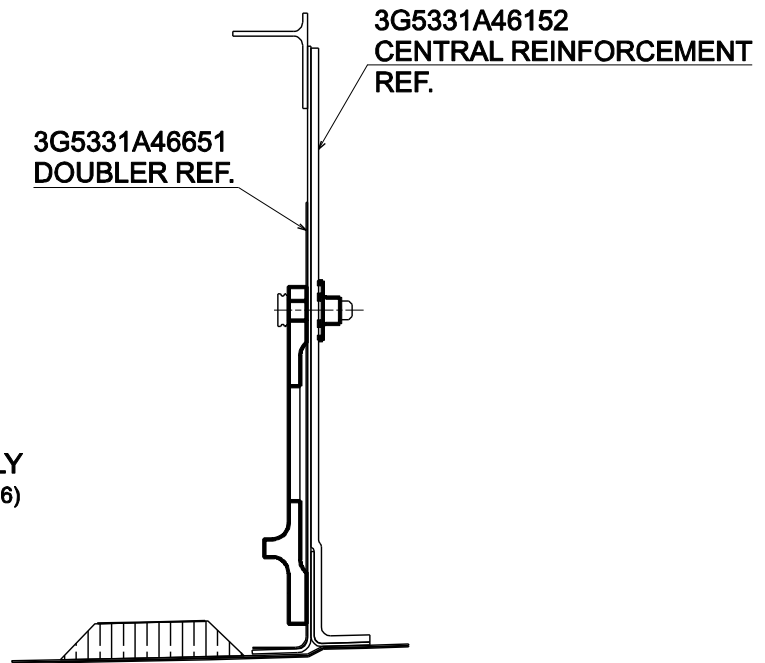


Figure 4

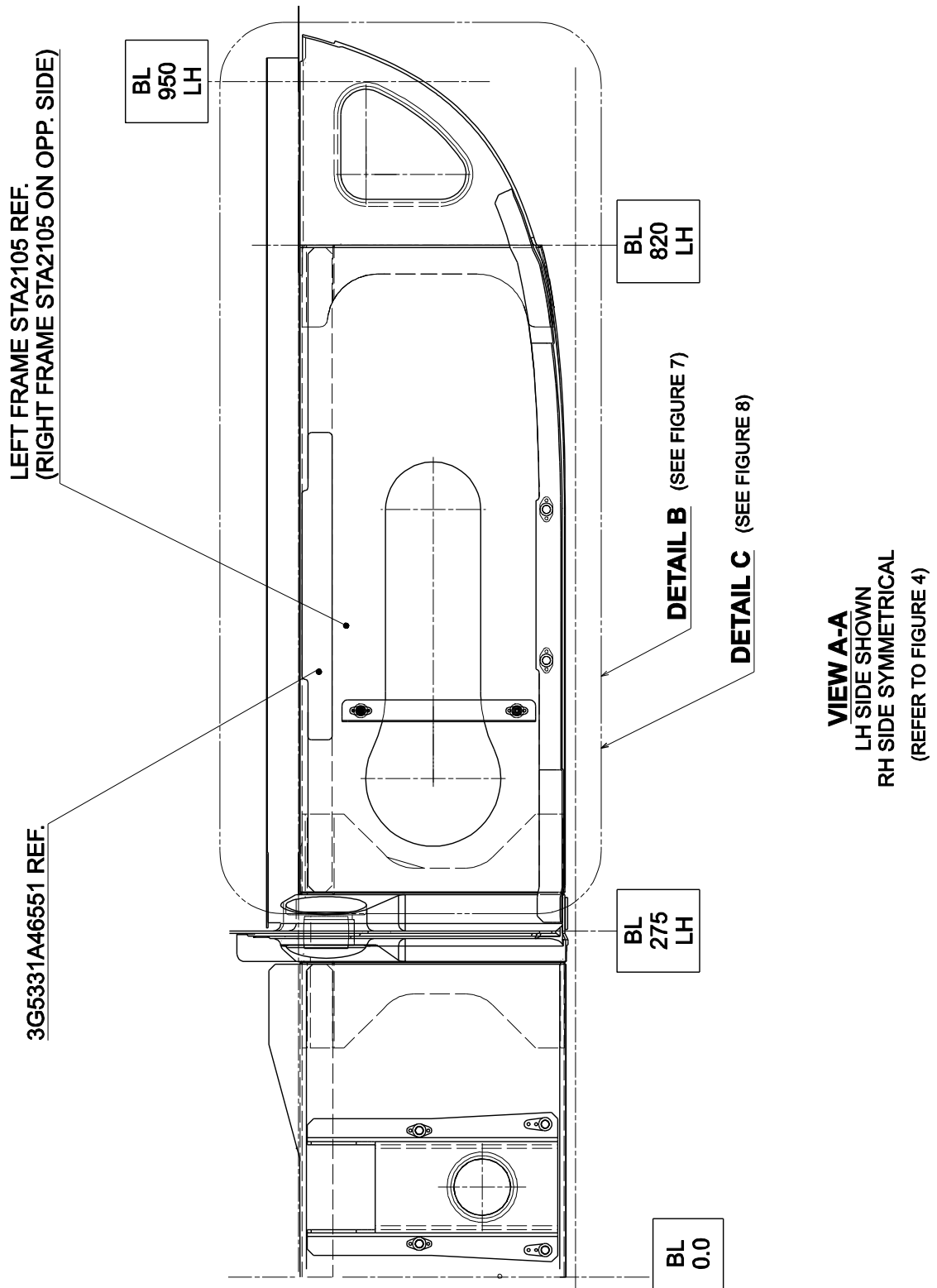
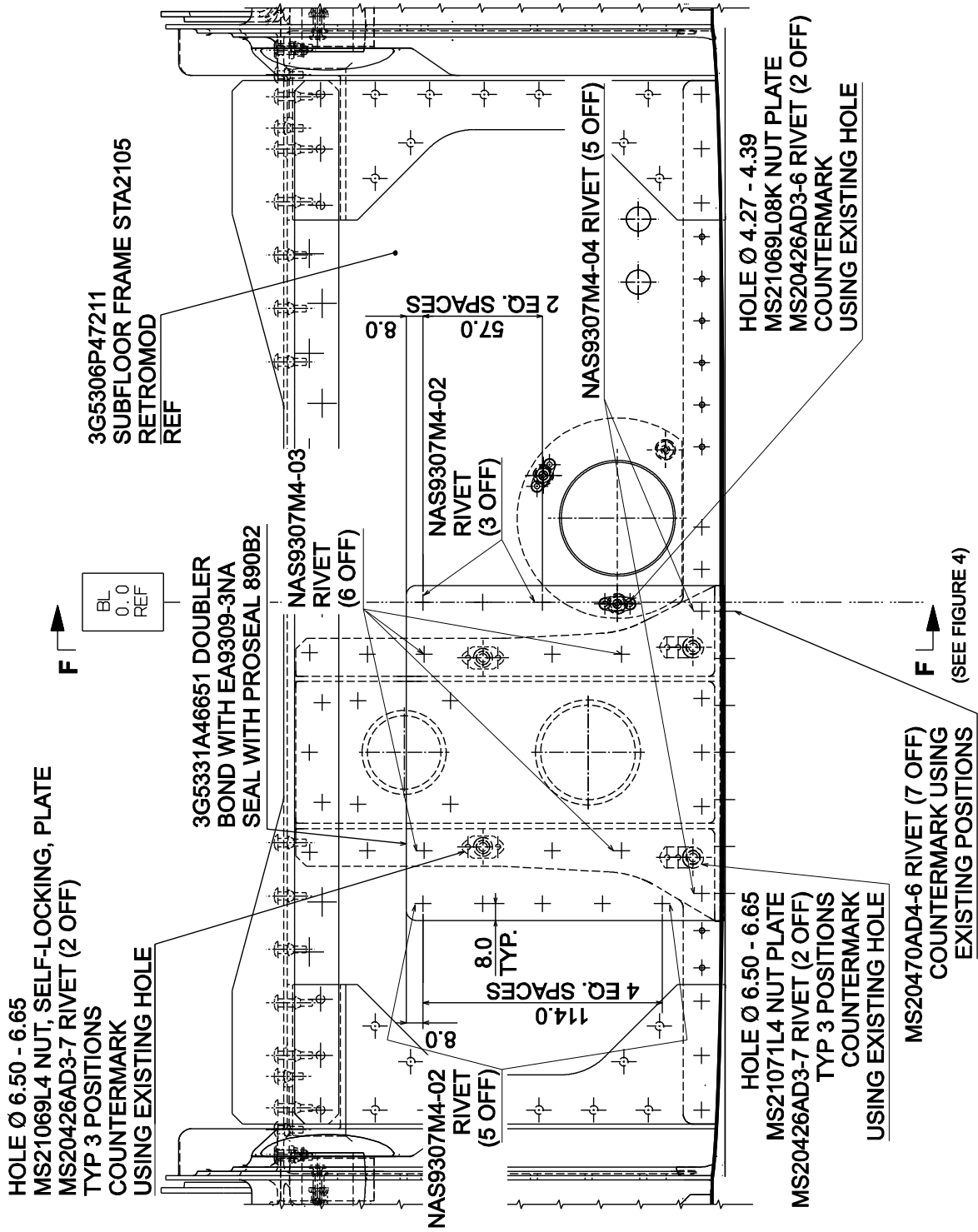


Figure 5



VIEW G
(REFER TO FIGURE 4)

Figure 6

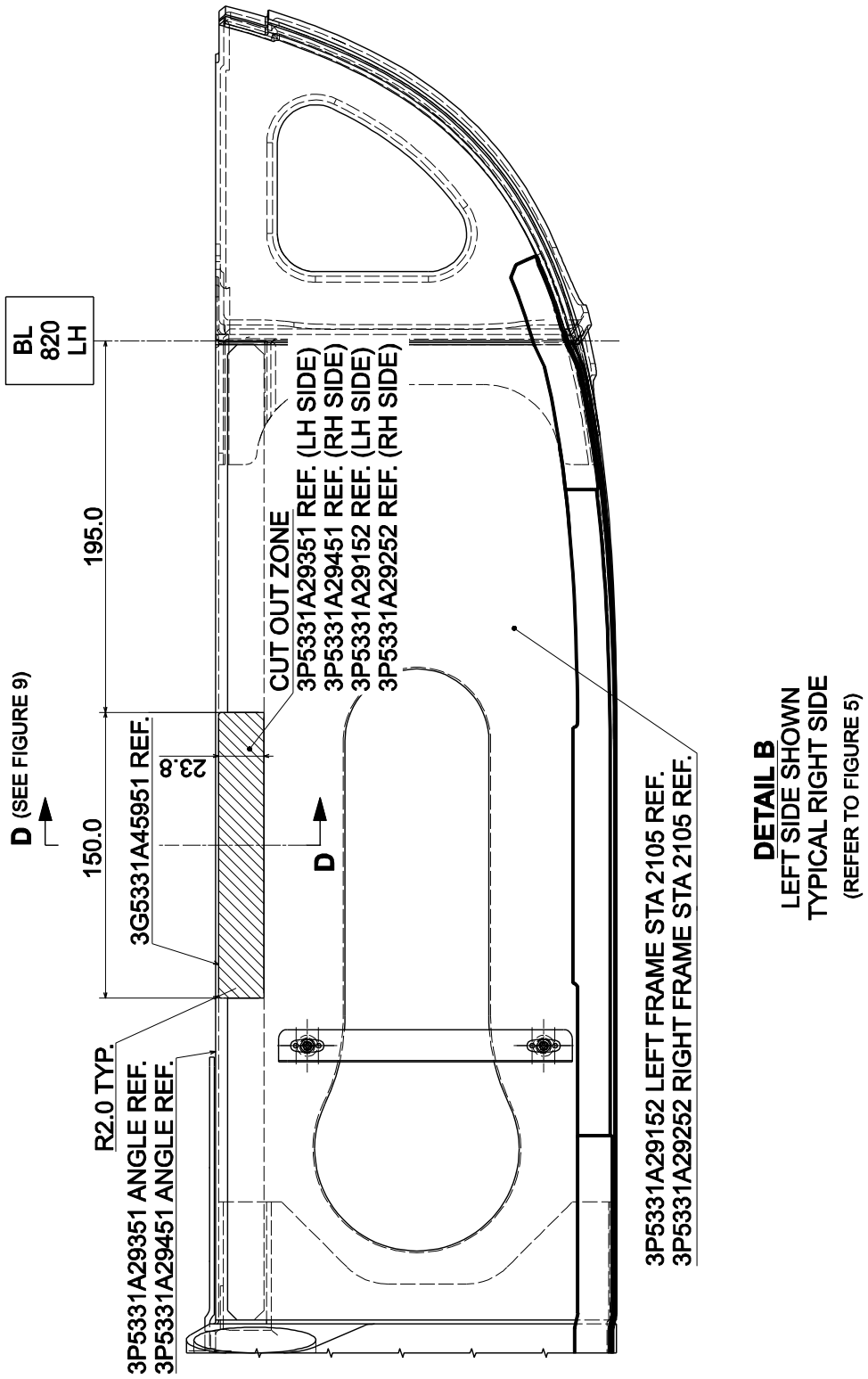


Figure 7

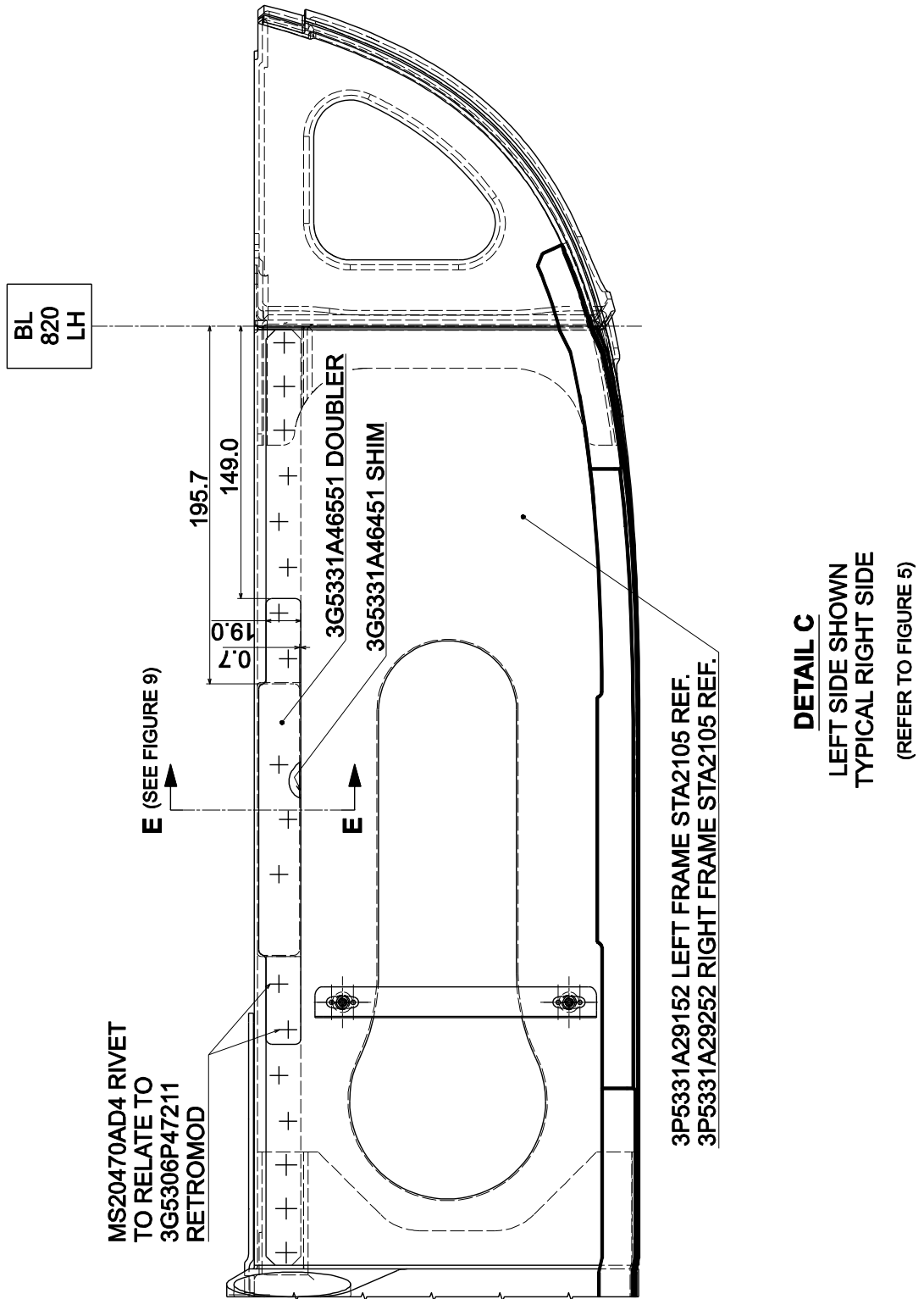


Figure 8

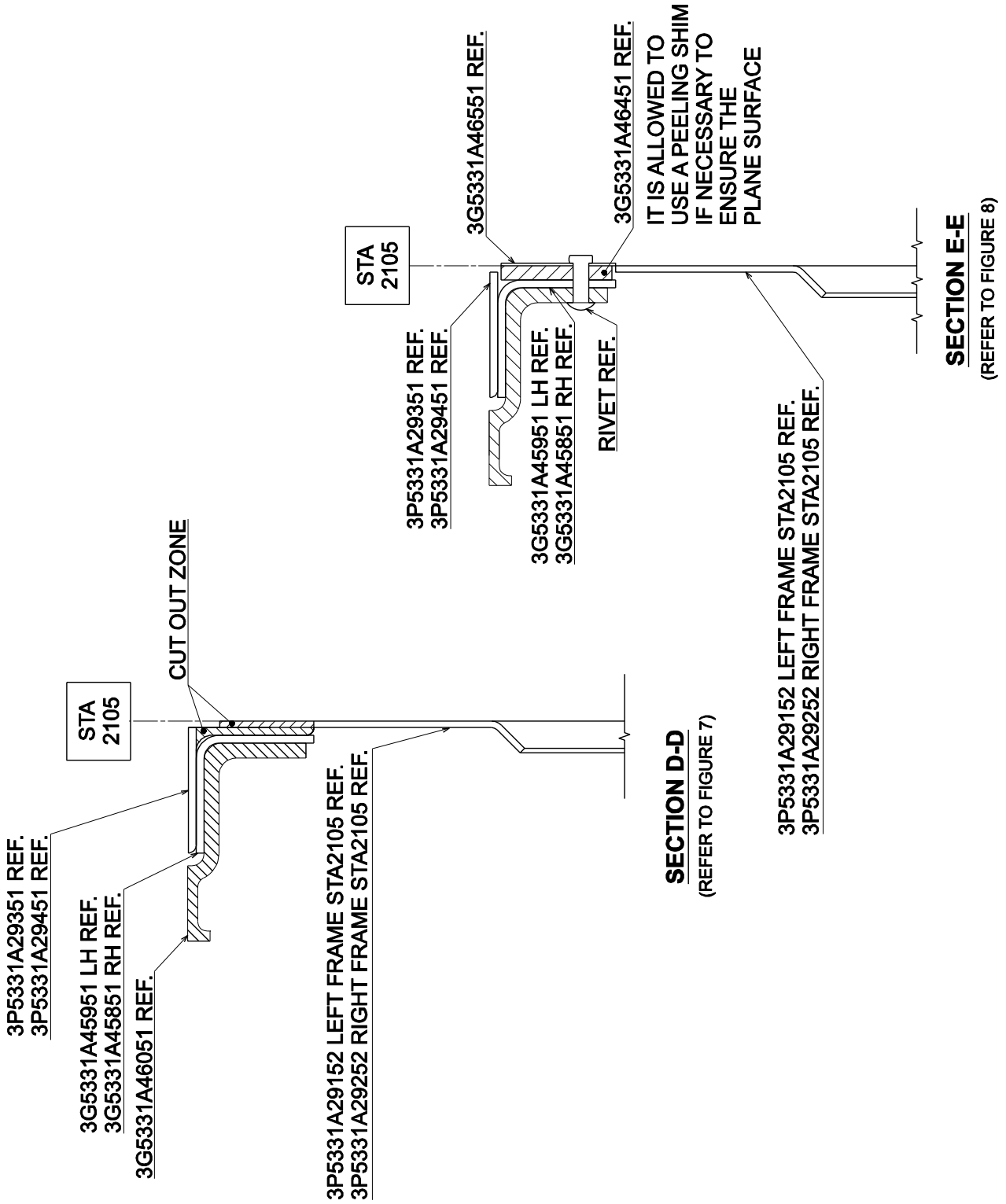
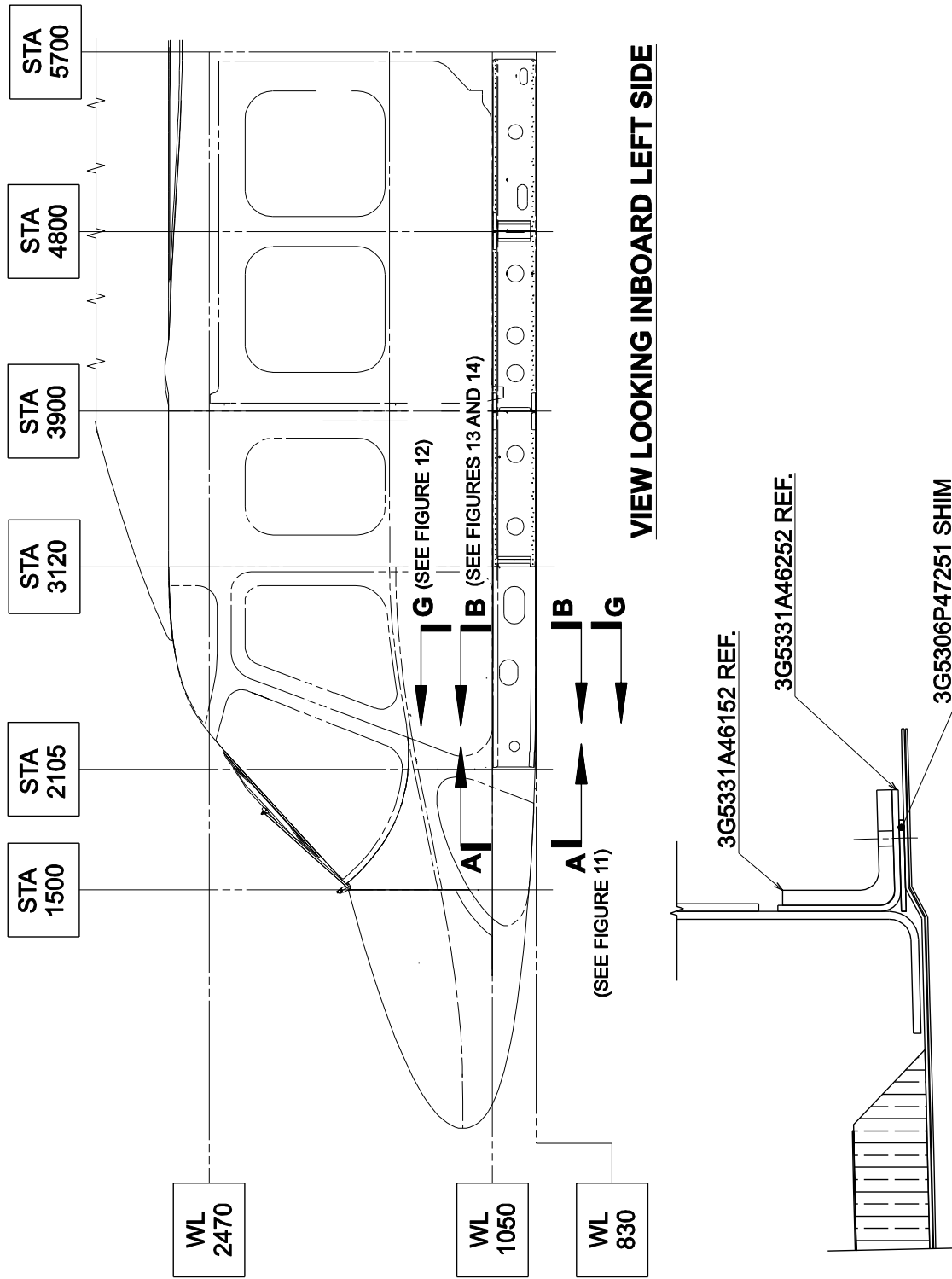


Figure 9



VIEW LOOKING INBOARD LEFT SIDE

SECTION F-F
(REFER TO FIGURE 19)

Figure 10

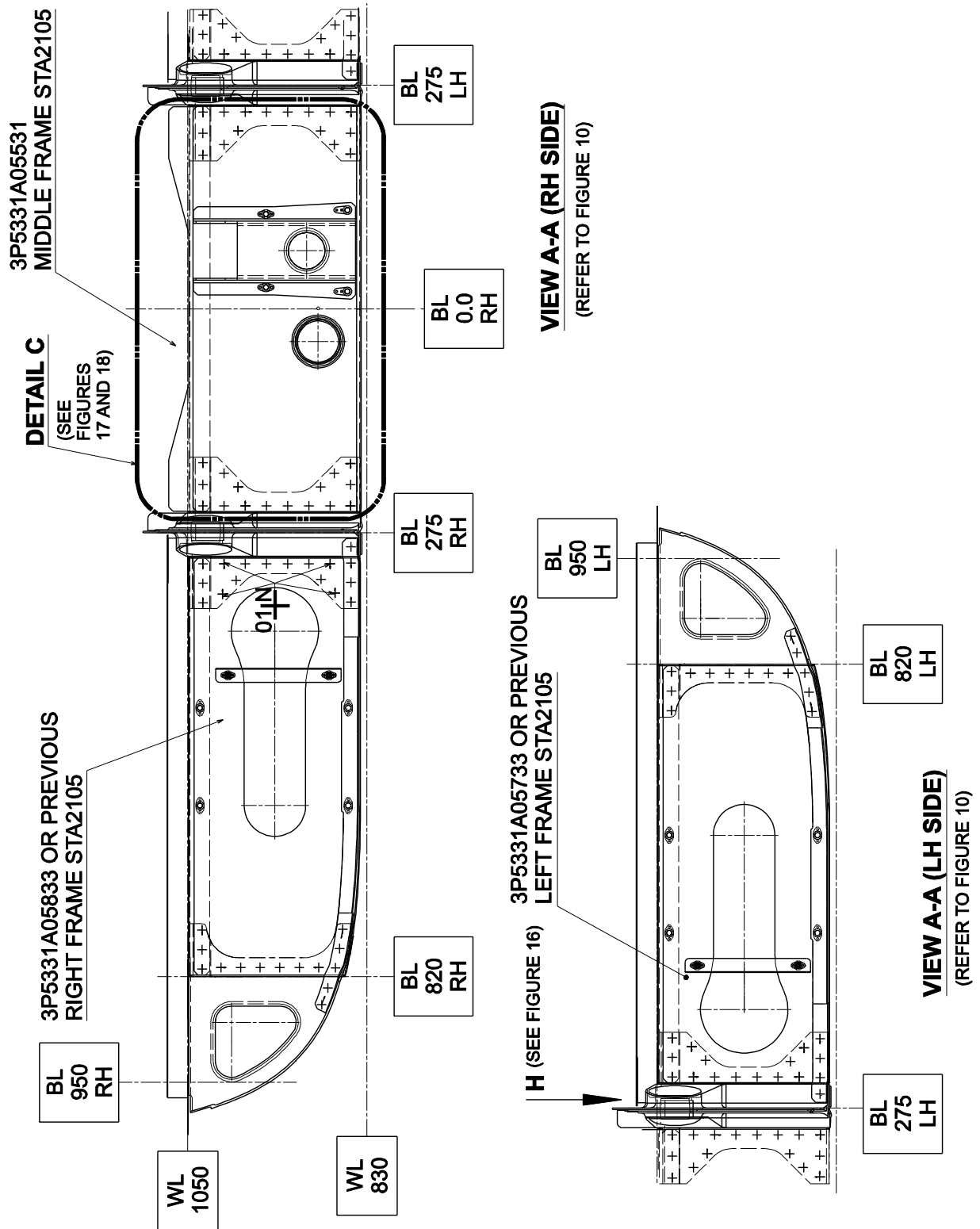
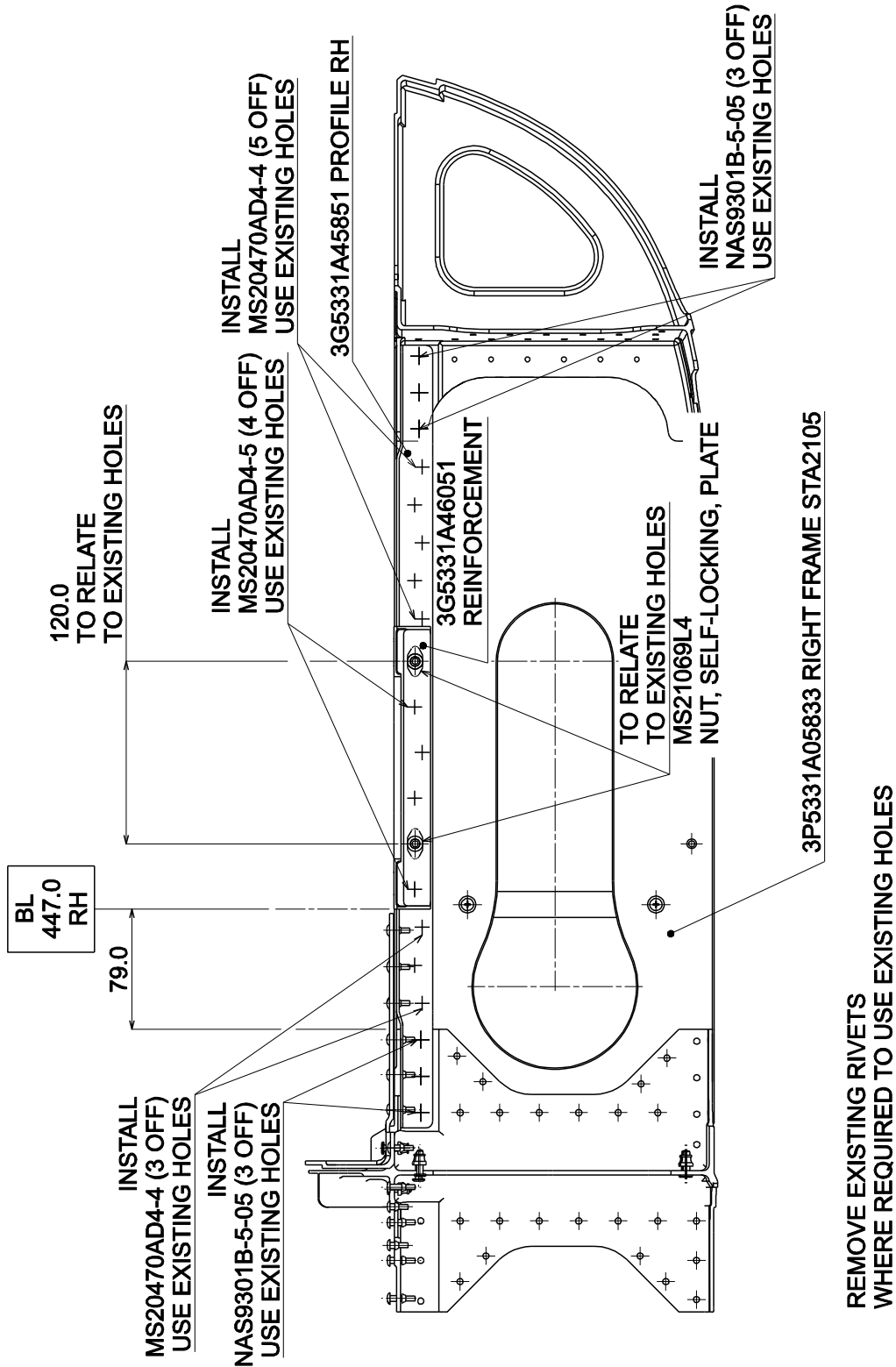
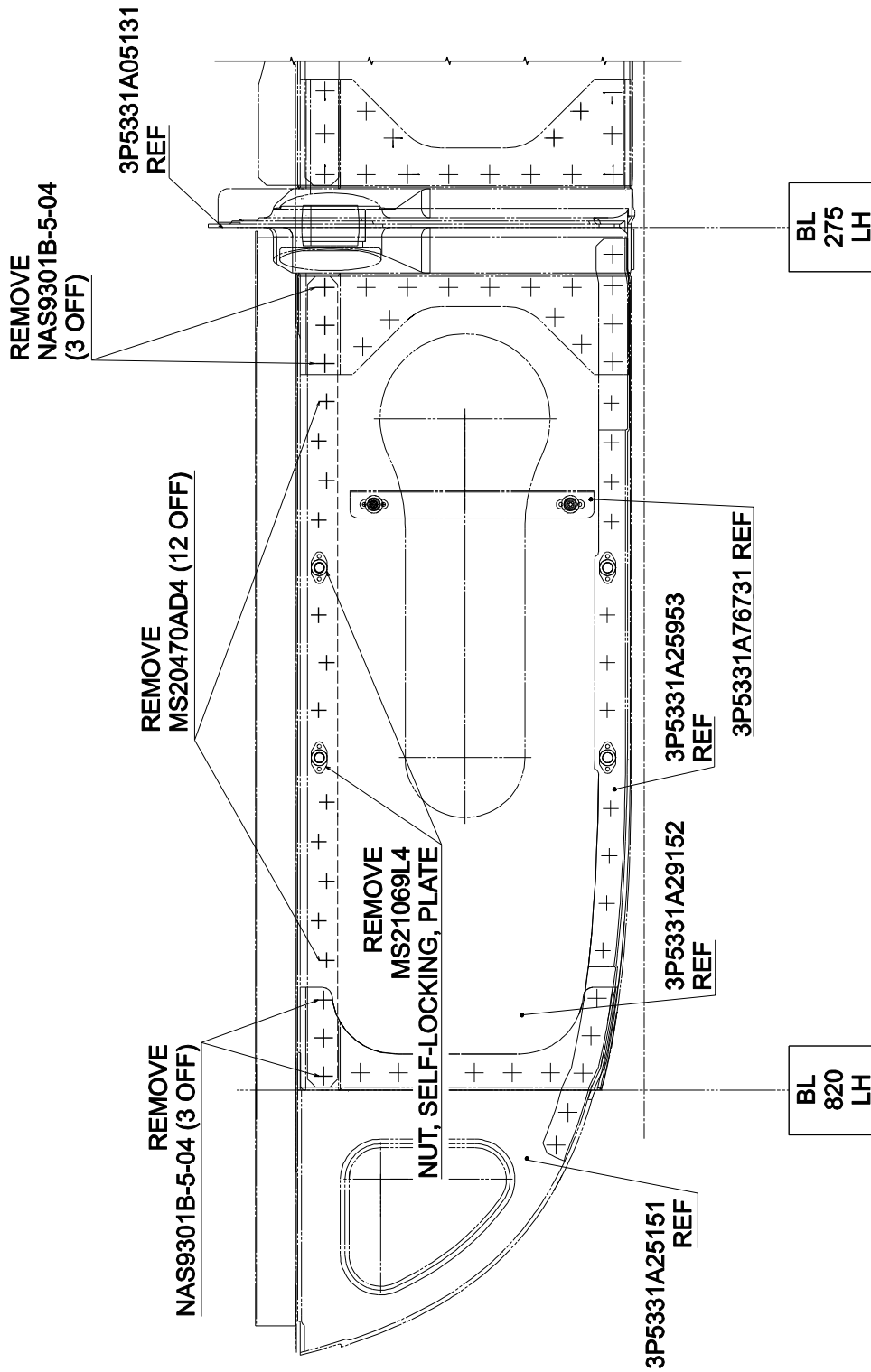


Figure 11



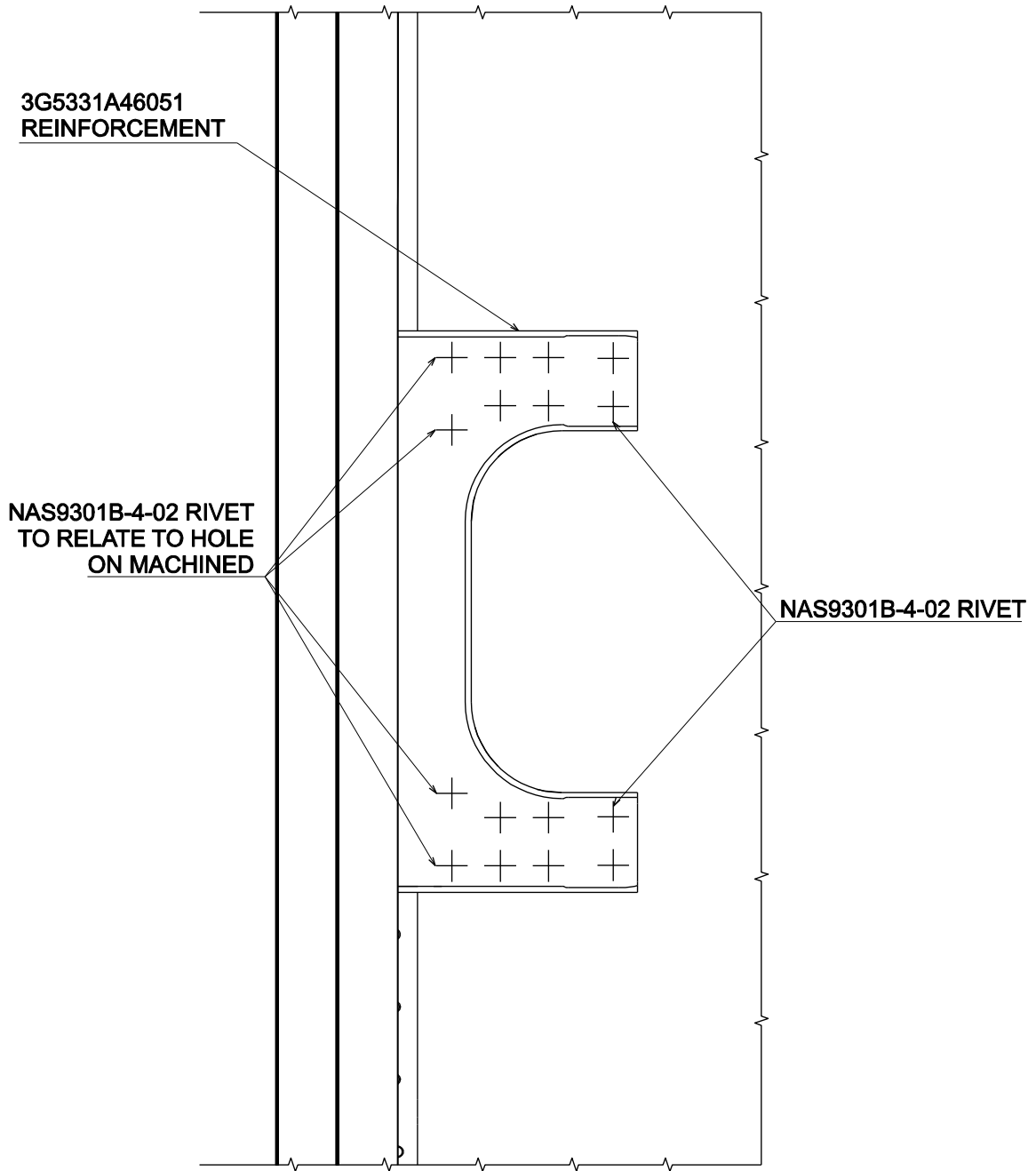
VIEW G-G
 VALID ONLY FOR RH SIDE
 (STRUCTURE PARTIALLY OMITTED FOR CLARITY)
 (REFER TO FIGURE 10)

Figure 12



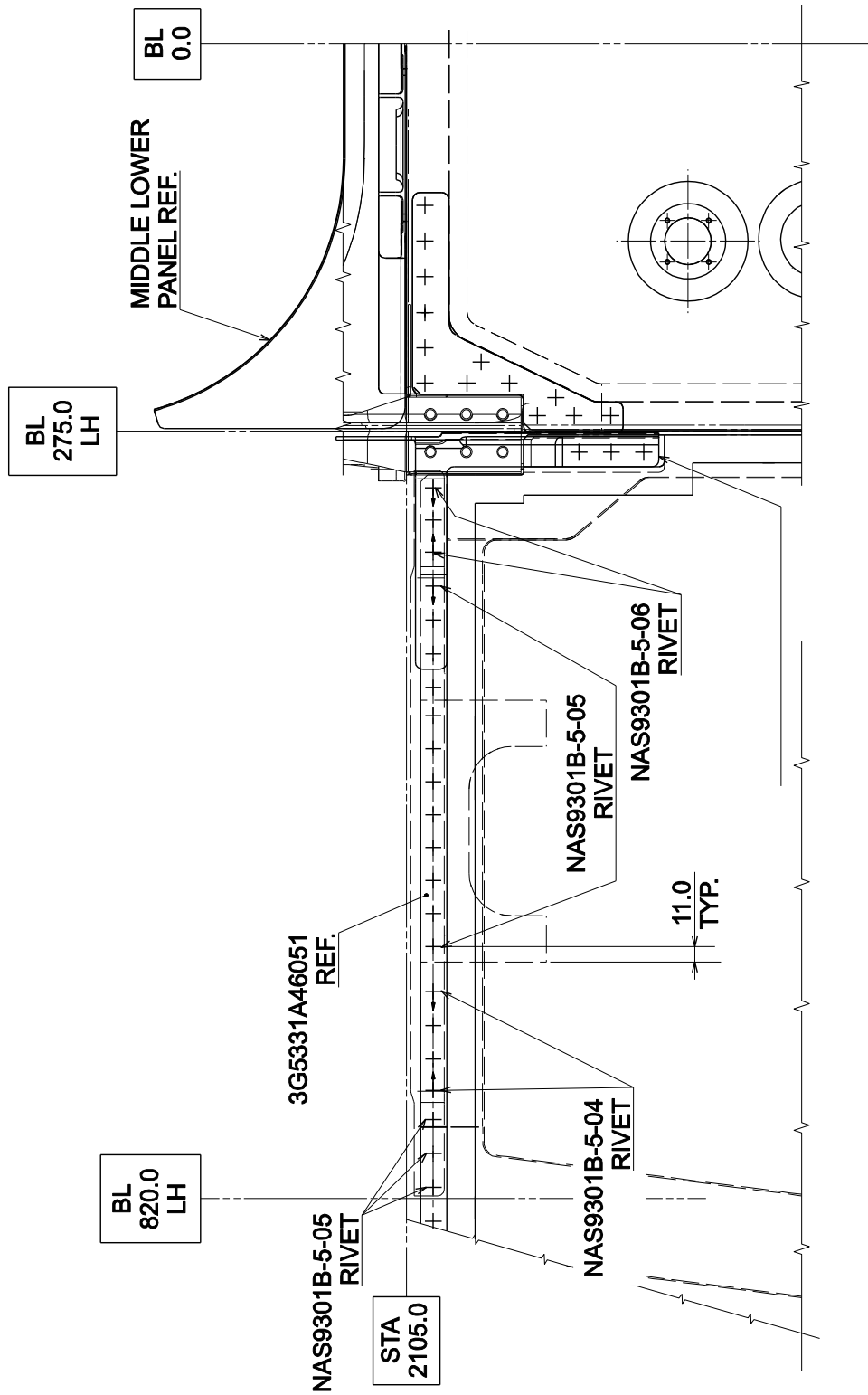
VIEW B-B
 VALID ONLY FOR LH SIDE
 (STRUCTURE PARTIALLY OMITTED FOR CLARITY)
BEFORE REWORK
 (REFER TO FIGURE 10)

Figure 13



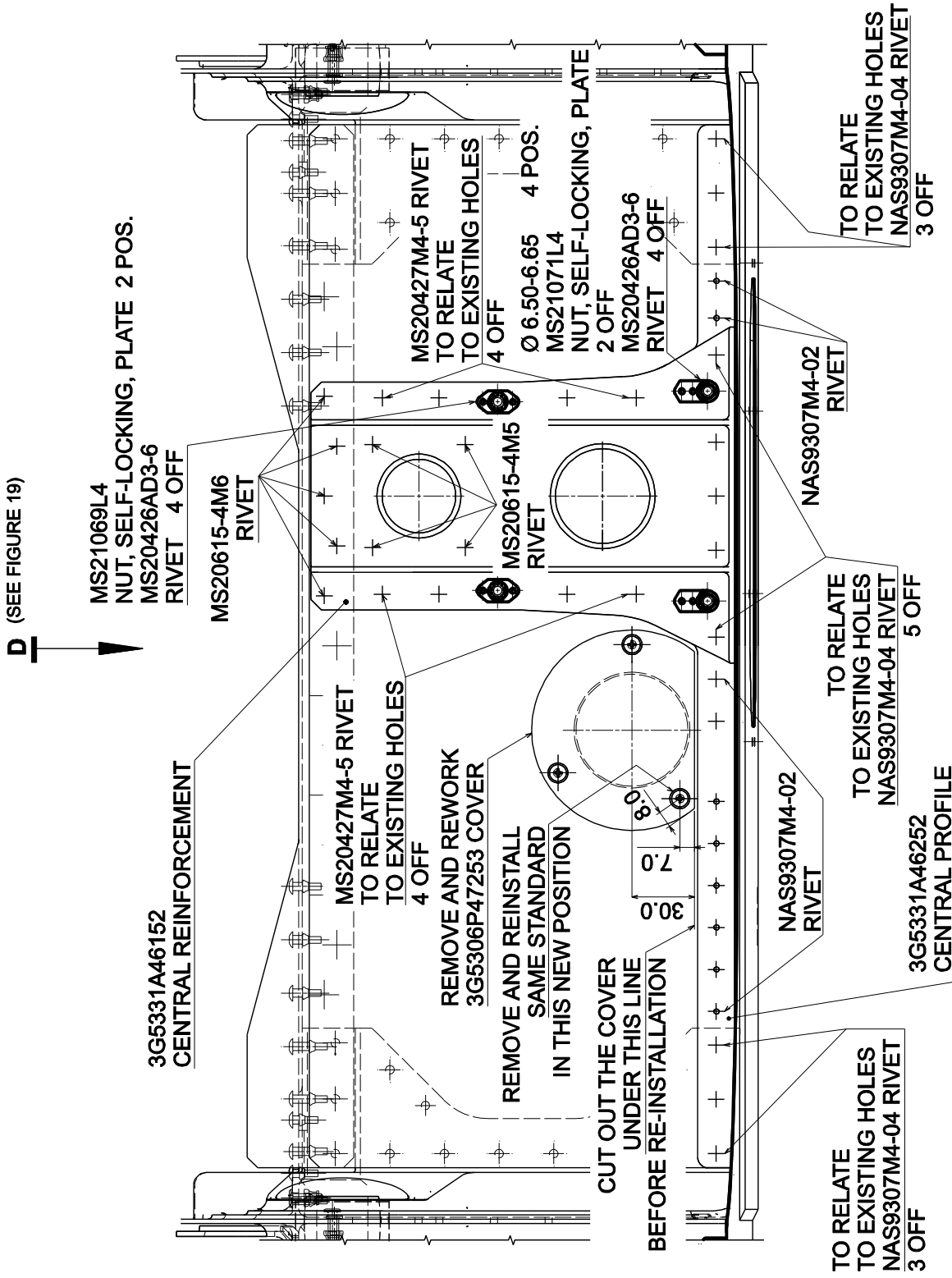
VIEW E
TYP OPP SIDE (ROTATED 90°)
(REFER TO FIGURE 14)

Figure 15



DETAIL H
LH SIDE SHOWN
TYP. FOR RH SIDE
(REFER TO FIGURE 11)

Figure 16



DETAIL C
AFTER REWORK
(REFER TO FIGURE 11)

Figure 18

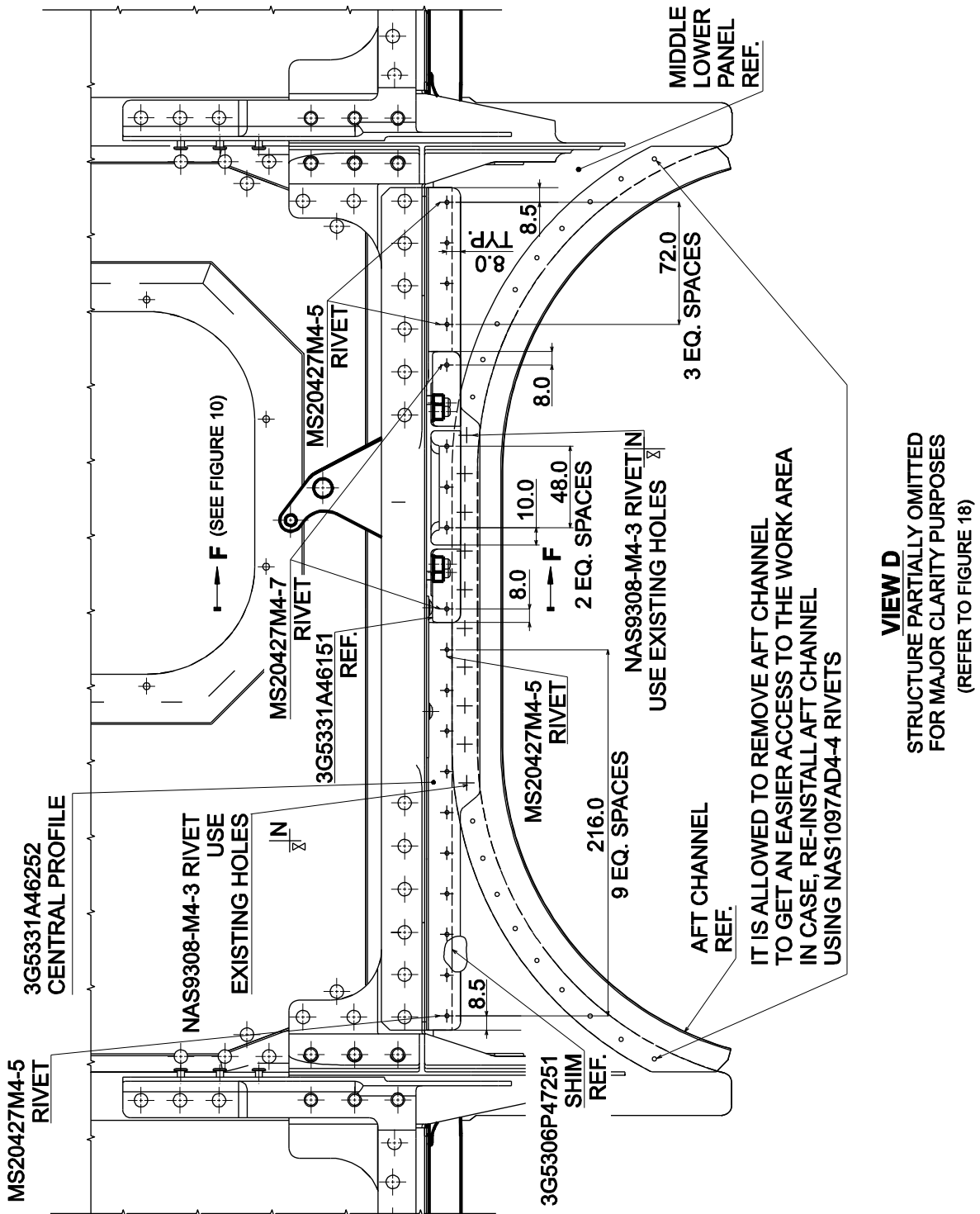


Figure 19

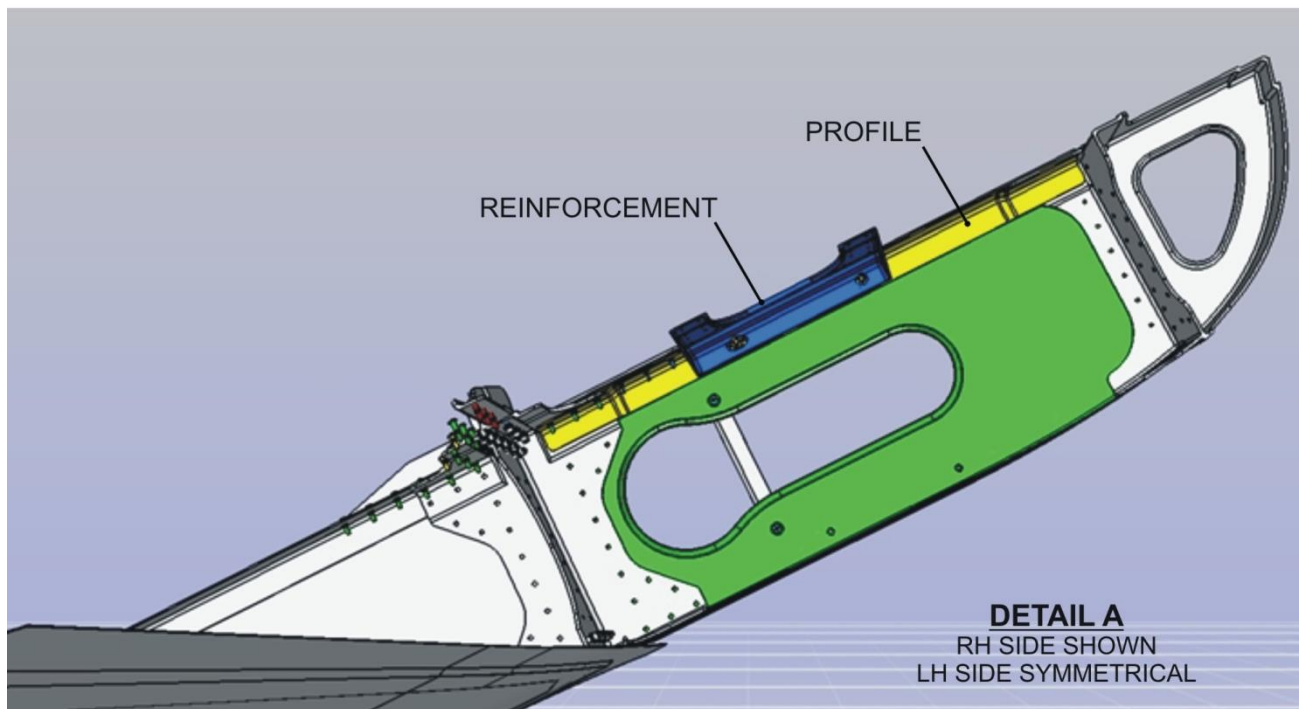
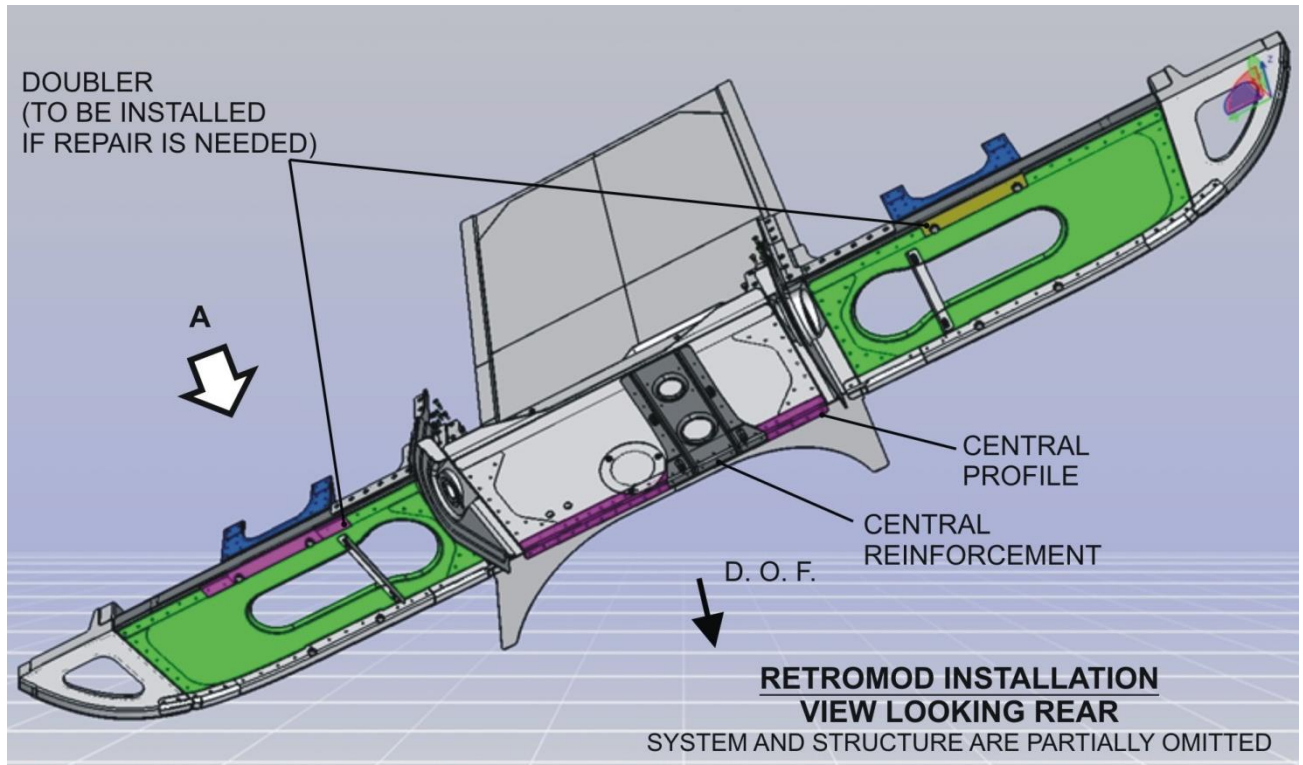


Figure 20

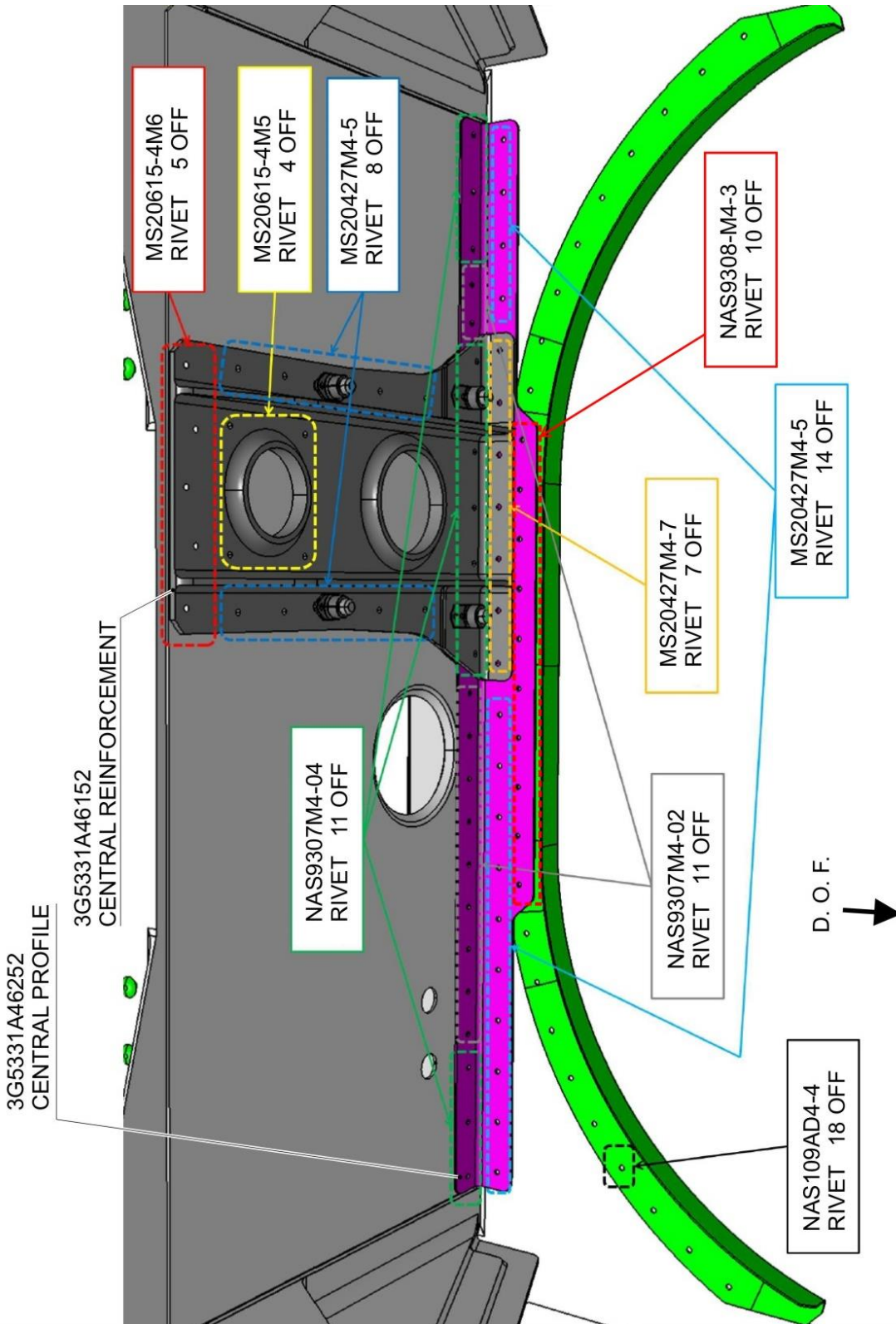


Figure 21

RETROMOD INSTALLATION
3D VIEW LOOKING REAR AND DOWN
 FOR REFERENCE ONLY
 SYSTEM AND STRUCTURE ARE PARTIALLY OMITTED

HARDWARE REQUIRED FOR
CENTRAL PANEL INSTALLATION

1	NAS9301B-4-02
2	NAS9301B-4-03
3	NAS9301B-4-04
4	MS20426AD4

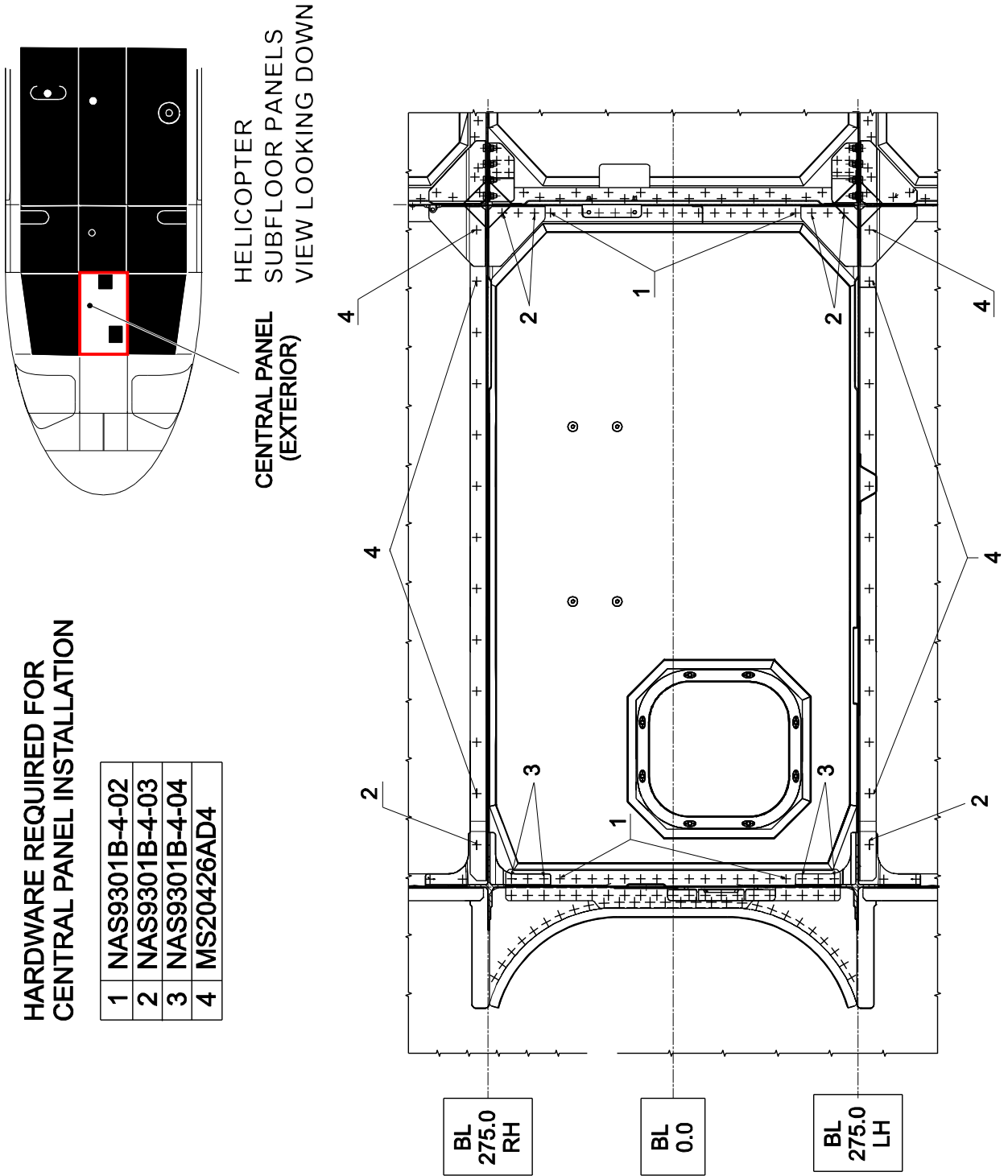


Figure 22

Prego spedire a questo indirizzo: <i>Please send to the following address:</i>		MODULO APPLICAZIONE BOLLETTINO TECNICO <i>TECHNICAL BULLETIN COMPLIANCE FORM</i>		Data: <i>Date:</i>
AGUSTAWESTLAND s.p.A. CUSTOMER SUPPORT & SERVICES - ITALY PRODUCT SUPPORT ENGINEERING DPT. Via del Gregge, 100 21015 Lonate Pozzolo (VA) - ITALY Tel.: +39 0331 664905 Fax: +39 0331 664684		Numero: <i>Number:</i>		
		Revisione: <i>Revision:</i>		
Denominazione Cliente ed Indirizzo: <i>Customer Name and Address:</i>			Telefono: <i>Telephone:</i>	
			Fax:	
			Data Applicazione B.T.: <i>B.T. Compliance Date:</i>	
Modello Elicottero <i>Helicopter Model</i>	S/N	Matricola <i>Tail Number</i>	Ore Totali <i>Total Hours</i>	Ore D.U.R. <i>T.S.O.</i>
Note: <i>Remarks:</i>				
Informazioni: <i>Information:</i>				
Al fine di gestire le varianti alla configurazione base, in relazione all'emissione del Bollettino Tecnico, preghiamo di voler compilare il presente modulo in tutte le sue parti e spedirlo all'indirizzo sopra indicato. Si ringrazia per la gentile collaborazione data.				
We request your cooperation in filling this form, in order to keep out statistical data relevant to aircraft configuration up-to-date. The form should be filled in all its parts and sent to the above address. We thank you beforehand for the information given.				